EASA	AIRWORTHINESS DIRECTIVE
	AD No.: 2011-0231 **Date: 09 December 2011** Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.

This AD is issued in accordance with EC 1702/2003, Part 21A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].

Type Approval Holder's Name :	**Type/Model designation(s) :**
AIRBUS	A319, A320 and A321 aeroplanes

TCDS Number :	EASA.A.064
Foreign AD :	Not applicable
Supersedure :	None

ATA 53	**Fuselage – Windshield Central Lower Node Continuity Fittings – Inspection / Repair**
Manufacturer(s):	Airbus (formerly Airbus Industrie)
Applicability:	Airbus A319-112, A319-113, A319-132, A320-211, A320-212, A320-214, A320-231, A320-232, A321-111 and A321-131 aeroplanes models, manufacturer serial numbers 0259, 0260, 0264, 0266 to 0270 inclusive, 0275, 0276, 0278, 0287, 0296, 0300, 0303, 0312, 0320, 0321, 0323, 0325, 0328, 0332, 0334, 0335, 0337, 0346, 0352, 0353, 0356, 0365, 0369, 0375, 0377, 0382, 0383, 0396, 0398, 0401, 0412, 0413, 0416, 0419, 0421, 0431, 0432, 0438, 0440, 0441, 0445, 0453, 0458, 0459, 0466, 0468, 0473, 0474, 0482, 0484, 0491, 0493, 0497, 0498, 0501, 0502, 0505, 0507, 0509, 0518, 0520, 0521, 0529, 0531, 0534, 0537, 0538, 0544, 0549, 0554, 0555, 0560, 0563, 0577, 0578, 0585, 0598, 0600, 0608, 0612, 0618, 0621, 0625, 0637, 0660, 0685, 0976, 1010, 1092, 1096, 1103, 1139, 1143, 1158, 1251, 1356 and 1511.
Reason:	One operator reported finding two fatigue cracks on continuity fittings at left-hand (LH) and right-hand (RH) sides at the front windshield lower framing on an A319 aeroplane on which Airbus modification (mod.) 22058 had been embodied in production. Airbus mod. 22058 (which is included in Airbus mod. 21999) was introduced to improve the fatigue strength of the windshield front framing by increasing the thickness of framing flanges adjacent to the concerned fittings. Further analyses have demonstrated that the damage tolerance and fatigue requirements of JAR 25.571 (b) are not met on aeroplanes in post-mod. 22058 configuration. This condition, if not detected and corrected, could reduce the structural integrity of the affected aeroplanes.

	For the reasons describe above, this AD requires, as a first step, for aeroplanes that are above or close to the threshold, a one-time High Frequency Eddy Current (HFEC) inspection of the windshield central lower node continuity fittings to detect cracks, the reporting of the inspection results and, depending on findings, the accomplishment of associated corrective actions. This inspection will be extended to all post mod. 22058 and post mod. 21999 Airbus A318/A319/ A320/A321 aeroplanes with repetitive inspection requirements in the next Airbus A318/A319/ A320/A321 Airworthiness Limitations Section (ALS) part 2 revision.
Effective Date:	23 December 2011
Required Action(s) and Compliance Time(s):	Required as indicated, unless accomplished previously: (1) Before the accumulation of 34 000 flight cycles (FC) since aeroplane first flight, or within 4 500 FC after the effective date of this AD, whichever occurs later, accomplish an HFEC inspection of the windshield central lower node continuity fittings at LH and RH sides in accordance with the instructions of Airbus Service Bulletin (SB) A320-53-1245 Revision 01. (2) If, during the HFEC inspection as required by paragraph (1) of this AD, cracks are detected, before next flight, contact Airbus for approved repair instructions and accomplish those instructions accordingly. (3) Within 30 days after the accomplishment of the HFEC inspection as required by paragraph (1) of this AD, report the results, including no findings, to Airbus. (4) Aeroplanes which have passed the HFEC inspection and applied the associated corrective actions before the effective date of this AD, in accordance with the accomplishment instructions of Airbus SB A320-53-1245 at original issue, are compliant with the requirements of paragraphs (1) and (2) of this AD. Within 30 days after the effective date of this AD, report the results of the inspection, including no findings, to Airbus.
Ref. Publications:	Airbus Service Bulletin A320-53-1245 Revision 01 dated 17 May 2011. The use of later approved revisions of this document is acceptable for compliance with the requirements of this AD.
Remarks :	1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD. 2. This AD was posted on 03 November 2011 as PAD 11-112 for consultation until 01 December 2011. The Comment Response Document can be found at http://ad.easa.europa.eu/. 3. Enquiries regarding this AD should be referred to the Safety Information Section, Executive Directorate, EASA. E-mail ADs@easa.europa.eu. 4. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – Airworthiness Office – EIAS, Fax +33 5 61 93 44 51; E-mail: account.airworth-eas@airbus.com.