


EASA	AIRWORTHINESS DIRECTIVE	
	<p>AD No.: 2012-0052</p> <p>Date: 29 March 2012</p> <p>Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation</p>	
<p>This AD is issued in accordance with EC 1702/2003, Part 21A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].</p>		
<p>Type Approval Holder's Name :</p> <p>AIRBUS</p>		<p>Type/Model designation(s) :</p> <p>A380 aeroplanes</p>
TCDS Number:	EASA.A.110	
Foreign AD:	Not applicable	
Supersedure:	None	
ATA 57 Wings – Leading Edge Shear Cleats – Inspection / Replacement		
Manufacturer(s):	Airbus	
Applicability:	Airbus A380-841, A380-842 and A380-861 aeroplanes, all manufacturer serial numbers.	
Reason:	<p>During full scale fatigue tests, cracks were found on two intercostal shear cleats on the inboard outer fixed leading edge (IOFLE) of the wing. The affected shear cleats of each wing are located at the lower forward (FWD) intercostal to closing rib interface, and at the lower aft (AFT) intercostal to drive rib 3 inboard interface.</p> <p>This condition, if not detected and corrected, could reduce the structural integrity of the wings.</p> <p>For the reason described above, this AD requires a one-time detailed visual inspection (DVI) of the affected shear cleats and, depending on the findings, the replacement of the parts.</p> <p>This AD is considered to be an interim measure, pending the approval of a modification to resolve the issue, and further AD action may follow.</p>	
Effective Date:	12 April 2012	

<p>Required Action(s) and Compliance Time(s):</p>	<p>Required as indicated, unless accomplished previously:</p> <ol style="list-style-type: none"> (1) Within 2 500 flight cycles since the first flight of the aeroplane, accomplish a DVI of the shear cleats of the lower FWD and AFT intercostals of each wing in accordance with the instructions of Airbus Service bulletin (SB) A380-57-8035. (2) If, during the inspection as required by paragraph (1) of this AD, any crack is found, before next flight, replace the affected shear cleat(s) in accordance with the instructions of Airbus SB A380-57-8035, and contact Airbus to report the inspection results to obtain approved instructions for inspection of the surrounding area where any shear cleat crack has been identified and, depending on findings, accomplish those instructions accordingly.
<p>Ref. Publications:</p>	<p>Airbus SB A380-57-8035 original issue, dated 16 September 2011.</p> <p>The use of later approved revisions of this document is acceptable for compliance with the requirements of this AD.</p>
<p>Remarks:</p>	<ol style="list-style-type: none"> 1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD. 2. The required actions and the risk allowance have granted the issuance of a Final AD with Request for Comments, postponing the public consultation process after publication. 3. Enquiries regarding this AD should be referred to the Safety Information Section, Executive Directorate, EASA. E-mail: ADs@easa.europa.eu. 4. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS SAS - EIANA (Airworthiness Office), Telephone: +33 562110253 ; Fax: +33 562 110 307. E-mail: account.airworth-A380@airbus.com or Nicolas.Cordeau@airbus.com.