


<b>EASA</b>	<b>AIRWORTHINESS DIRECTIVE</b>
	<p><b>AD No.: 2012-0048</b></p> <p><b>Date: 03 April 2012</b></p> <p>Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.</p>
<p>This AD is issued in accordance with EC 1702/2003, Part 21A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].</p>	
<p><b>Type Approval Holder's Name :</b></p> <p>AIRBUS</p>	<p><b>Type/Model designation(s) :</b></p> <p>A380 aeroplanes</p>
TCDS Number :	EASA.A.110
Foreign AD :	Not applicable
Supersedure :	None
<b>ATA 53</b>	<b>Fuselage – Rivets at Junction of Stringer 21 and Frame 0 – Replacement</b>
Manufacturer(s):	Airbus
Applicability:	Airbus A380-841, A380-842, and A380-861 aeroplanes, all manufacturer serial numbers, except aeroplanes on which Airbus modification 72043 has been embodied in production.
Reason:	<p>During an engineering review, it has been identified that aluminium rivets which have been used at the junction of fuselage stringer (STGR) 21 Left Hand (LH) and Right Hand (RH) and frame (FR) 0 are not in compliance with the certification requirements.</p> <p>This condition, if not corrected, could lead, in case of rapid decompression, to in-flight loss of the radome which could affect the structural integrity of the aeroplane.</p> <p>To address this unsafe condition, Airbus developed the modification 72043, embodied on aeroplanes in production, and published the Service Bulletin (SB) A380-53A8044 with instructions to retrofit aeroplanes in-service.</p> <p>For the reasons described above, this AD requires the replacement of the affected 12 aluminium rivets with 12 hi-lite titanium fasteners (6 LH and 6 RH).</p>
Effective Date:	17 April 2012

Required Action(s) and Compliance Time(s):	<p>Required as indicated, unless accomplished previously:</p> <p>Within 7 months after the effective date of this AD, replace the aluminium rivets with hi-lite titanium fasteners at the junctions of the STGR21 LH and STGR21 RH and the FR0 in accordance with the instructions of Airbus SB A380-53A8044.</p>
Ref. Publications:	<p>Airbus SB A380-53A8044 at Original Issue dated 17 October 2011.</p> <p>The use of later approved revisions of this document is acceptable for compliance with the requirements of this AD.</p>
Remarks :	<ol style="list-style-type: none"> <li>1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.</li> <li>2. This AD was posted on 15 February 2012 as PAD 12-014 for consultation until 14 March 2012. The Comment Response Document can be found at <a href="http://ad.easa.europa.eu/">http://ad.easa.europa.eu/</a>.</li> <li>3. Enquiries regarding this AD should be referred to the Safety Information Section, Executive Directorate, EASA. E-mail: <a href="mailto:ADs@easa.europa.eu">ADs@easa.europa.eu</a>.</li> <li>4. For any question concerning the technical content of the requirements in this AD, please contact:  AIRBUS SAS - EIANA (Airworthiness Office),  Telephone: +33 562110253 ; Fax:+33 562 110 307.  E-mail: <a href="mailto:account.airworth-A380@airbus.com">account.airworth-A380@airbus.com</a> and/or  <a href="mailto:Nicolas.Cordeau@airbus.com">Nicolas.Cordeau@airbus.com</a> and/or <a href="mailto:Sandra.Cuiec@airbus.com">Sandra.Cuiec@airbus.com</a>.</li> </ol>