

	<b>AIRWORTHINESS DIRECTIVE</b> <b>No F-1992-106-132 R7</b>	Distribution: <b>B</b>	Issue date: <b>January 05, 2005</b>	Page : <b>1/3</b>
Direction générale de l'aviation civile France  GSAC publication	This Airworthiness Directive is published by the DGAC on behalf of EASA, Airworthiness Authority of the State of Design for the affected product, part or appliance.		<i>Translation of « Consigne de Navigabilité » of same number. In case of difficulty, reference should be made to the French issue.</i>	
	<b>No person may operate an aircraft to which an Airworthiness Directive applies, except in accordance with the requirements of that Airworthiness Directive, unless otherwise agreed with the Authority of the State of Registry.</b>			
Corresponding foreign Airworthiness Directive(s): <b>Not applicable</b>		Airworthiness Directive(s) replaced: <b>1992-106-132 R6</b>		
Person in charge of airworthiness: <b>AIRBUS SAS</b>		Type(s): <b>A310 aircraft</b>		
Type certificate(s) No. <b>72</b> TCDS No <b>145</b>				
ATA chapter: <b>53, 55, 57</b>	Subject: <b>Structure fatigue</b>			

**1. EFFECTIVITY:**

AIRBUS A310 aircraft, all certified models and all serial numbers.

**2. REASONS:**

The goal of this Airworthiness Directive (AD) is to detect or prevent possible damage associated with a phenomenon of structural fatigue.

Revision 6 of this AD cancels part 1.14 of paragraph COMPLIANCE which is replaced by AD 2003-242.

Revision 7 of this AD cancels part 1.14 of paragraph 3 which is replaced by AD 2005-001.

**3. MANDATORY ACTIONS AND COMPLIANCE TIMES:**

1. Perform the inspections and/or modifications defined in the following AIRBUS Service Bulletins, in accordance with the thresholds indicated in each of these Service Bulletins or within 1000 flights following the effective date of this Airworthiness Directive Revision 3 (June 17, 1995), whichever occurs later.

- 1.1. S.B. A310-53-2014 Revision 5 and Change Notice 5B:  
Fuselage - Center Section - Reinforce bottom joint angle fitting at FR40.
- 1.2. S.B. A310-53-2016 Revision 5:  
Fuselage - Center Section - FR47 and FR54 area - Provide local reinforcement.
- 1.3. S.B. A310-53-2054 Revision 2:  
Fuselage - Center Section - Inspect frame reinforcement angle runout of FR46 between STGR21 and 22 LH/RH.



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- 1.4. S.B. A310-53-2057 Revision 1:  
Fuselage - Center section - Inspection of the T-section connecting FR50A to the beam between FR50 and FR51 LH/RH.
- 1.5. S.B. A310-53-2059 Revision 1:  
Fuselage - Inspect the lower side panel at the lap joint with the upper side panel at FR47 and STGR22, LH/RH.
- 1.6. S.B. A310-55-2002 Revision 4:  
Stabilizers - Horizontal stabilizer structure - Reinforcement of junction parts.
- 1.7. S.B. A310-55-2004 Revision 3:  
Stabilizers - Horizontal stabilizer structure - Inspection program after reinforcement as per Service Bulletin n° A310-55-2002 embodiment.
- 1.8. S.B. A310-57-2002 Revision 2:  
Wings - Leading edge access panels landing - Lower skin - Inspection for cracks at bolt holes.
- 1.9. S.B. A310-57-2006 Revision 3:  
Wings - Holes around overwing refueling aperture at ribs 3 and 4, inspection.
- 1.10. S.B. A310-57-2032 Revision 3:  
Wings - Upper skin forward of front spar - Inspection for cracks.
- 1.11. Cancelled [see AD 2003-242(B)].
- 1.12. S.B. A310-57-2039:  
Wings - Center wing - Inspection of front spar web at rib posts.
- 1.13. S.B. A310-57-2046 Revision 1:  
Wings - Rear spar - Select bolt locations for attachment of MLG forward pick up fitting.
- 1.14. Cancelled [see AD F-2005-001].
- 1.15. S.B. A310-57-2050 and Change Notice OB:  
Wings - Center wing box - Inspect treillis boom drain holes.
- 1.16. S.B. A310-57-2004 Revision 2:  
Fuselage - Tail cone - Inspect the lower horizontal-stabilizer cutout longeron, the corner fitting, the skin star and the skin between FR87 and FR89 and between STGR24 and STGR27, LH and RH.
- 1.17. S.B. A310-57-2064:  
Fuselage - FR40 - Inspect upper corner fitting.
- 1.18. S.B. A310-57-2038 Revision 3:  
Wings - Stringer flanges at rib 14 wing bottom skin - Inspect for cracks.

**Note 1:** For paragraphs 1.1, 1.10, [...], 1.17, 1.18 comply with the instructions of above-mentioned Service Bulletins at the defined thresholds or within 1,000 flight following the effective date of this Airworthiness Directive Revision 4, whichever occurs later.

2. Repeat the inspections defined in the Service Bulletins mentioned in paragraph 1 above, in accordance with the intervals indicated in each of the Service Bulletins when applicable.

**Note 2:** The thresholds and intervals which are established and valid for aircraft operated for an individual average airborne flight time of 96 minutes must be adjusted in accordance with the instructions given in the Service Bulletins quoted above.



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**4. REFERENCE PUBLICATIONS:**

AIRBUS Service Bulletins:

A310-53-2014, -2016, -2054, -2057, -2059, -2074,

A310-55-2002, -2004

A310-57-2002, -2006, -2032, [...], -2038, -2039, -2046, [...], -2050, -2064.

Any later approved revision of these SB's is acceptable.

**5. EFFECTIVE DATES :**

**Original issue** : May 09, 1992

**Revision 1** : May 09, 1992

**Revision 2** : January 28, 1995

**Revision 3** : June 17, 1995

**Revision 4** : June 15, 1996

**Revision 5** : November 28, 1998

**Revision 6** : July 05, 2003

**Revision 7** : January 15, 2005.

**6. REMARK:**

For questions concerning the technical content of this AD's requirements, contact:

AIRBUS SAS - Didier AURICHE - Fax: 33 5 61 21 22 00.

**7. APPROVAL:**

This AD Revision is approved under EASA reference No 2004-12551 dated December 27, 2004.

**SUPERSEDED**