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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 98-ANE-39-AD; Amendment 39-12668; AD 2002-04-11]

RIN 2120-AA64

Airworthiness Directives; General Electric Company GE90 Series Turbofan Engines; Correction

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; correction.

SUMMARY: This document makes a correction to Airworthiness Directive (AD) 2002-04-11 applicable to General Electric Company GE90 series turbofan engines that was published in the Federal Register on March 4, 2002 (67 FR 9582). The Table in the regulatory text section is incorrect. This document corrects that Table. In all other respects, the original document remains the same.

EFFECTIVE DATE: April 8, 2002.

FOR FURTHER INFORMATION CONTACT: Ian Dargin, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803-5299; telephone (781) 238-7178, fax (781) 238-7199.

SUPPLEMENTARY INFORMATION: A final rule airworthiness directive (FR Doc. 02-5003) applicable to General Electric Company GE90 series turbofan engines, was published in the Federal Register on March 4, 2002 (67 FR 9582). The following corrections are needed:

Sec. 39.13 [Corrected]

1. On page 9583, in the third column entitled, Inspect per engine manual chapter, in the third entry, (HPCR, Disk, Stage 7) "72-31-07-200-001-001 Fluorescent Penetrant Inspection (subtask 72-31-07-230-051), and 72-31-07-200-001-001 Eddy Current Inspection (subtask 72-31-07-250-051 or 72-31-07-230-052 or 72-31-07-230-053" is corrected to read "72-31-07-200-001-001 Fluorescent Penetrant Inspection (subtask 72-31-07-230-051), and 72-31-07-200-001-001 Eddy Current Inspection of the Rim Boltholes (subtask 72-31-07-250-051 or 72-31-07-250-052 or 72-31-07-250-053".

2. On the same page, in the same column entitled, Inspect per engine manual chapter, in the ninth entry, (HPTR Disk, Stage 1) "72-53-02-200-001-002 Fluorescent Penetrant Inspection (subtask 72-53-02-160-051), and 72-53-02-200-001-002 Eddy Current Inspection of the Bore " is corrected to read "72-53-02-200-001-002 Fluorescent Penetrant Inspection (subtask 72-53-02-230-052), and 72-53-02-200-001-002 Eddy Current Inspection of the Bore".

Issued in Burlington, MA, on April 18, 2002.

Francis A. Favara,

Acting Manager, Engine and Propeller Directorate, Aircraft Certification Service.

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AIRWORTHINESS DIRECTIVE



Aircraft Certification Service
Washington, DC

U.S. Department
of Transportation
**Federal Aviation
Administration**

We post ADs on the internet at "www.airweb.faa.gov/rgl"

The following Airworthiness Directive issued by the Federal Aviation Administration in accordance with the provisions of Title 14 of the Code of Federal Regulations (14 CFR) part 39, applies to an aircraft model of which our records indicate you may be the registered owner. Airworthiness Directives affect aviation safety and are regulations which require immediate attention. You are cautioned that no person may operate an aircraft to which an Airworthiness Directive applies, except in accordance with the requirements of the Airworthiness Directive (reference 14 CFR part 39, subpart 39.3).

CORRECTION: *This document makes a correction to Airworthiness Directive (AD) 2002-04-11 applicable to General Electric Company GE90 series turbofan engines that was published in the Federal Register on March 4, 2002 (67 FR 9582). The Table in the regulatory text section is incorrect. This document corrects that Table. In all other respects, the original document remains the same. [Federal Register: April 29, 2002 (Volume 67, Number 82)]*

AD 2002-04-11 General Electric Company: Docket No. 98-ANE-39-AD. Supersedes AD 2000-08-10, Amendment 39-11696.

Applicability

This airworthiness directive (AD) is applicable to General Electric Company (GE) GE90-76B/-77B/-85B/-90B/-94B series turbofan engines. These engines are installed on but not limited to Boeing 777 series airplanes.

Note 1: This AD applies to each engine identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For engines that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (c) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance

Compliance with this AD is required as indicated, unless already done.

To prevent critical life-limited rotating engine part failure, which could result in an uncontained engine failure and damage to the airplane, do the following:

Inspections

(a) Within 30 days after the effective date of this AD, revise the manufacturer's Life Limits Section of the Instructions for Continued Airworthiness (ICA), and for air carrier operations revise the approved continuous airworthiness maintenance program, by adding the following:

"MANDATORY INSPECTIONS

(1) Perform inspections of the following parts at each piece-part opportunity in accordance with the instructions provided in the applicable manual provisions:

Part nomenclature	Part no. (P/N)	Inspect per engine manual chapter
For GE90 Engines:		
HPCR Disk, Stage 1	All	72-31-05-200-001-001 Fluorescent Penetrant Inspection (subtask 72-31-05-230-051), and 72-31-05-200-001-001 Eddy Current Inspection of the Bore, and 72-31-05-200-001-001 Eddy Current Inspection of the Dovetail Slots.
HPCR Spool, Stage 2-6	All	72-31-06-200-001-001 Fluorescent Penetrant Inspection (subtask 72-31-06-230-051), and 72-31-06-200-001-001 Eddy Current Inspection of the S2 Dovetail Slots.
HPCR, Disk, Stage 7	All	72-31-07-200-001-001 Fluorescent Penetrant Inspection (subtask 72-31-07-230-051), and 72-31-07-200-001-001 Eddy Current Inspection of the Rim Boltholes (subtask 72-31-07-250-051 or 72-31-07-230-052 or 72-31-07-230-053).
HPCR Spool, Stage 8-10	All	72-31-08-200-001-001 Fluorescent Penetrant Inspection and 72-31-08-800-001 Eddy Current Inspection of the stage 8-9 inertia weld.
HPCR Seal, Compressor Discharge Pressure	All	72-31-09-200-001-001 Fluorescent Penetrant Inspection (subtask 72-31-09-230-051), and 72-31-09-200-001-001 Eddy Current Inspection of the Boltholes.
HPCR Ring, Tube Supporter	All	72-31-10-200-001-001 Fluorescent Penetrant Inspection.
HPTR, Interstage Seal	All	72-53-03-200-001-001 Fluorescent Penetrant Inspection (subtask 72-53-03-230-053), and 72-53-03-200-001-001 Eddy Current Inspection of the Bore.
Fan Disk, Stage 1	All	72-21-03-200-001-001 Fluorescent Penetrant Inspection (subtask 72-21-03-230-051), and 72-21-03-200-001-001 Eddy Current of the bore, and 72-21-03-200-001-001 Ultrasonic Inspection of Dovetail Slots.
HPTR Disk, Stage 1	All	72-53-02-200-001-002 Fluorescent Penetrant Inspection (subtask 72-53-02-230-052), and 72-53-02-200-001-002 Eddy Current Inspection of the Bore.
HPTR Disk, Stage 2	All	72-53-04-200-001-004 Fluorescent Penetrant Inspection (subtask 72-53-04-230-052), and 72-53-04-200-001-004 Eddy Current Inspection of the Bore.
LPTR Cone Shaft	All	72-56-07-200-001-001 Fluorescent Penetrant Inspection.
LPTR Fan Mid Shaft	All	72-58-01-200-001-001 Magnetic Particle Inspection.
LPTR Disk, Stage 1	All	72-56-02-200-001-001 Fluorescent Penetrant Inspection.
LPTR Disk, Stage 2	All	72-56-02-200-001-001 Fluorescent Penetrant Inspection.
LPTR Disk, Stage 3	All	72-56-02-200-001-001 Fluorescent Penetrant Inspection.
LPTR Disk, Stage 4	All	72-56-02-200-001-001 Fluorescent Penetrant Inspection.
LPTR Disk, Stage 5	All	72-56-02-200-001-001 Fluorescent Penetrant Inspection.
LPTR Disk, Stage 6	All	72-56-02-200-001-001 Fluorescent Penetrant Inspection.
Fan Shaft, Forward	All	72-22-01-200-001-001 Fluorescent Penetrant Inspection.

(2) For the purposes of these mandatory inspections, piece-part opportunity means:

(i) The part is considered completely disassembled when accomplished in accordance with the disassembly instructions in the manufacturer's engine manual; and

(ii) The part has accumulated more than 100 cycles in service since the last piece-part opportunity inspection, provided that the part was not damaged or related to the cause for its removal from the engine."

(b) Except as provided in paragraph (c) of this AD, and notwithstanding contrary provisions in section 43.16 of the Federal Aviation Regulations (14 CFR 43.16), these mandatory inspections must be performed only in accordance with the Life Limits Section of the manufacturer's ICA.

Alternative Methods of Compliance

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Engine Certification Office (ECO). Operators must submit their requests through an appropriate FAA Principal Maintenance Inspector (PMI), who may add comments and then send it to the ECO.

Note 2: Information concerning the existence of approved alternative methods of compliance with this airworthiness directive, if any, may be obtained from the ECO.

Special Flight Permits

(d) Special flight permits may be issued in accordance with Secs. 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be done.

Continuous Airworthiness Maintenance Program

(e) FAA-certificated air carriers that have an approved continuous airworthiness maintenance program in accordance with the record keeping requirement of Sec. 121.369 (c) of the Federal Aviation Regulations (14 CFR 121.369 (c)) of this chapter must maintain records of the mandatory inspections that result from revising the Life Limits Section of the ICA and the air carrier's continuous airworthiness program. Alternatively, certificated air carriers may establish an approved system of record retention that provides a method for preservation and retrieval of the maintenance records that include the inspections resulting from this AD, and include the policy and procedures for implementing this alternate method in the air carrier's maintenance manual required by Sec. 121.369 (c) of the Federal Aviation Regulations (14 CFR 121.369 (c)); however, the alternate system must be accepted by the appropriate PMI and require the maintenance records be maintained either indefinitely or until the work is repeated. Records of the piece-part inspections are not required under Sec. 121.380 (a) (2) (vi) of the Federal Aviation Regulations (14 CFR 121.380 (a) (2) (vi)). All other Operators must maintain the records of mandatory inspections required by the applicable regulations governing their operations.

Note 3: The requirements of this AD have been met when the engine manual changes are made and air carriers have modified their continuous airworthiness maintenance plans to reflect the requirements in the engine manuals.

Effective Date

(f) This amendment becomes effective on April 8, 2002.

Issued in Burlington, Massachusetts, on February 21, 2002.

Jay J. Pardee,

Manager, Engine and Propeller Directorate, Aircraft Certification Service.

[FR Doc. 02-5003 Filed 3-1-02; 8:45 am]

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