

**State of Israel
Ministry of Transport
Civil Aviation Authority**

AD 85-001R3

Date: February 18, 2008

AIRWORTHINESS DIRECTIVE

This Airworthiness Directive, issued by the Civil Aviation Authority in accordance with the Israeli Aviation Regulations, applies to an aircraft for which our records list you as the registered owner. It affects aviation safety and requires your immediate attention. No person may operate an aircraft to which an Airworthiness Directive applies, except in accordance with that Directive.

APPLIES TO: Israel Aircraft Industries Ltd. (IAI), models 1121/1121A/1121B/1123/1124/1124A, all serial numbers.

SUBJECT: Inspection for cracks in the Outer Hinge Lugs of Horizontal Stabilizer Aft Spar Splice Fitting,(Hinge Assembly), part number 453005-501.
Revision 2 changes the inspection requirements.
Revision 3 adds terminating action by replacement the affected part made of aluminum to a titanium one .

REASON: Cracks have been found in the Splice Fitting Hinge Lugs on several aircraft.

REASON FOR REVISION 2: Service experience and additional analysis have shown that the dye penetrant inspections every 600 flight hours, for airplanes without scissors, and every 2400 flight hours for airplanes with scissors installed, as indicated in revision 1 of this AD, must be replaced by a visual inspection every 300 flight hours.

REASON FOR REVISION 3: Availability of strengthened fitting made of titanium. Required inspection intervals are considerably longer when the titanium fitting is installed.

ACTION: 1. Within the next 75 flight hours time in service after the effective date of this AD, unless previously inspected within the last 225 flight hours time in service, inspect the horizontal stabilizer Aft Spar Splice Fitting P/N 453005-501, in accordance with the following IAI Service Bulletins:-

<u>Model</u>	<u>Service Bulletin</u>	
1121, 1121A,1121B	1121-55-003-Rev.1	8/8/88
1123	1123-55-006 Rev.1	8/8/88
1124, 1124A	1124-55-020-Rev.2	8/8/88

or later approved revision.

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2. If no cracks are found, repeat this inspection at intervals not to exceed 300 flight hours.
3. If cracks are found, the Splice Fitting must be replaced before further flight. Replacement should be accomplished in accordance with the following IAI Service Bulletins:-

<u>Model</u>	<u>Service Bulletin</u>
1121, 1121A, 1121B	1121-55-004-Rev.2 10/21/88
1123	1123-55-007 Rev.2 10/21/88
1124, 1124A	1124-55-021-Rev.3 10/21/88

NOTE: A Titanium Splice Fitting, P/N WW4453005-509 may be installed, mod no. AFC2081, per Aircraft Maintenance Manual (AMM) instructions. Inspection interval for the modified aircraft are included in Chapter 5 of the AMM. Incorporation of this mod constitutes terminating action for the inspection requirements of this AD.

EFFECTIVE DATE: February 18, 2008 for the optional Rev. 3 modification.
October 25, 1988 for the inspection requirements per Rev. 2.