

**U.S. DEPARTMENT OF TRANSPORTATION
FEDERAL AVIATION ADMINISTRATION**

Airworthiness Directive

93-25-07 BEECH AIRCRAFT CORPORATION: Amendment 39-8773. Docket No. 93-CE-38-AD.

Applicability: The model and serial number airplanes presented below, certificated in any category, that do not have all the fuselage stringers Nos. 5 through 11 modified on both the right and left hand sides in accordance with either (1) Beech Service Bulletin (SB) 2472, Revision 1, dated September 1993; (2) Chapter 51-10 or 53-10, as applicable, of the maintenance manual; or (3) the instructions to Priester Aviation Service Supplemental Type Certificate (STC) SA63CH:

Models	Serial Numbers
200, A200, B200, and A100-1	BB-2 through BB-1462, BC-1 through BC-75, and BD-1 through BD-30
200C, A200C, and B200C	BL-1 through BL-138, BJ-1 through BJ-66, BU-1 through BU-12, and BV-1 through BV-12
200CT, A200CT, B200CT	BN-1 through BN-4, BP-1 through BP-71, FC-1, FC-2, FC-3, FE-1 through FE-31, FG-1, FG-2, and GR-1 through GR-19
200T and B200T	BT-1 through BT-38
300	FA-1 through FA-228
300 (FAA)	FF-1 through FF-19
B300	FL-1 through FL-103
B300C	FM-1 through FM-8 and FN-1

Compliance: Required upon the accumulation of 3,000 hours time-in-service (TIS) or within the next 100 hours TIS after the effective date of this AD, whichever occurs later, unless already accomplished, and thereafter as indicated.

To prevent structural damage to the fuselage caused by cracked stringers in the rear pressure bulkhead area, accomplish the following:

(a) Inspect fuselage stringers Numbers (Nos.) 5 through 11 on both the left and right hand sides for cracks in accordance with the ACCOMPLISHMENT INSTRUCTIONS section of Beech SB No. 2472, Revision 1, dated September 1993.

(1) If no cracks are found, reinspect at intervals that correspond with the following:

**Stringers Modified in Accordance With
One of the Three Modifications Referenced
in the Applicability Section of this AD**

Inspection Interval

No Stringers Modified	600 hours TIS on all stringers
Nos. 8, 9, and 10 (one side) with Internal Modification	1,200 hours TIS on unmodified Stringers
Nos. 8, 9, and 10 (one side) with External Modification	600 hours TIS on unmodified Stringers
Nos. 5 through 11	AD no longer applies

(2) If cracks are found, modify all cracked fuselage stringers at any time up to the time specified in the chart presented in paragraph (a)(3) of this AD. Accomplish this modification in accordance with the instructions in one of the three modifications specified in the Applicability section of this AD, and reinspect at intervals presented in the chart in paragraph (a)(1) of this AD.

(3) The following chart specifies the required compliance time where cracked stringers must be modified:

**No. of Stringers Cracked on Any
One Side of Fuselage**

**When Modification Must
be Accomplished (Hours TIS)**

1 to 3	150
3	30
5 or more	Prior to further flight

(b) The modifications specified in the Applicability section of this AD may be accomplished at any time as terminating action for the inspection requirement of this AD provided that all fuselage stringers Nos. 5 through 11 are modified.

(c) Special flight permits may be issued in accordance with FAR 21.197 and 21.199 to operate the airplane to a location where the requirements of this AD can be accomplished.

(d) An alternative method of compliance or adjustment of the compliance time that provides an equivalent level of safety may be approved by the Manager, Wichita Aircraft Certification Office (ACO), FAA, 1801 Airport Road, Room 100, Wichita, Kansas 67209. The request should be forwarded through an appropriate FAA Maintenance Inspector, who may add comments and send it to the Manager, Wichita ACO.

NOTE: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Wichita ACO.

(e) The inspections and modification required by this AD shall be done in accordance with Beech Service Bulletin 2472, Revision 1, dated September 1993. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR Part 51. Copies may be obtained from the Beech Aircraft Corporation, P.O. Box 85, Wichita, Kansas 67201-0085. Copies may be inspected at the FAA, Central Region, Office of the Assistant Chief Counsel, Room 1558, 601 E. 12th Street, Kansas City, Missouri, or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(f) This amendment (39-8773) becomes effective on February 15, 1994.