

CESSNA AIRCRAFT COMPANY
MODEL 200 SERIES (1960 - 1965)
SERVICE MANUAL

SUPPLEMENTAL INSPECTION NUMBER: 32-10-01

1. **TITLE:**
Main Landing Gear Retraction System Inspection

2. **EFFECTIVITY**
21057001 thru 21058715

INSPECTION COMPLIANCE

ALL USAGE:	INITIAL	6,000 hours	or	10 Years (NOTE)
	REPEAT	1,000 hours	or	3 Years (NOTE)

NOTE: Refer to Note 1, Section 2A-14-00

3. **PURPOSE**
To ensure integrity of main landing gear retraction system.

4. **INSPECTION INSTRUCTIONS**

- A. Check airplane records to verify that Service Bulletin SEB01-2 has been accomplished. If not, complete SEB01-2 with this inspection.
- B. Remove seats and carpeting as required to gain access to the plate above the actuator.
- C. Inspect the actuator, actuator support assembly, and shaft for cracks and corrosion. Refer to Figure 1.
 - (1) Clean area before inspecting if grime or debris are present.
- D. Inspect the actuator pins, piston rack teeth, and sectors for cracks and wear.
- E. Install parts removed for access.

5. **ACCESS AND DETECTABLE CRACK SIZE**

ACCESS/LOCATION/ZONE

Main Gear Section

DETECTABLE CRACK SIZE

Not Allowed

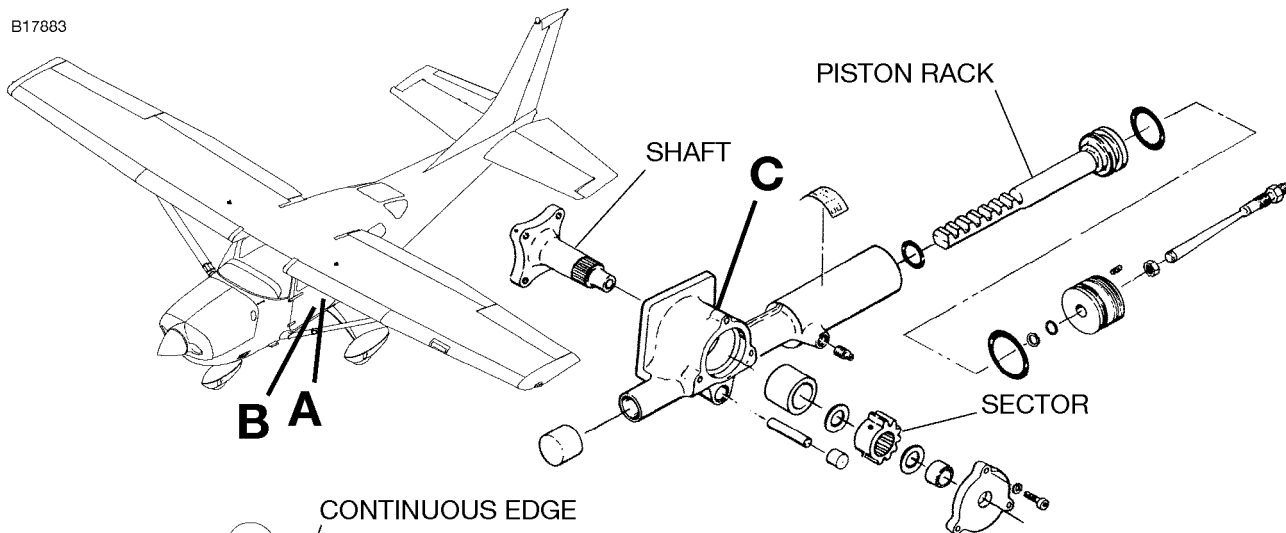
6. **INSPECTION METHOD**
Visual

7. **REPAIR/MODIFICATION**
A. Replace damaged parts.

8. **COMMENTS**

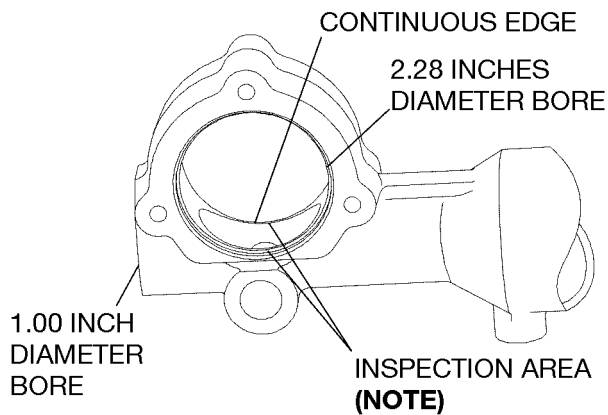
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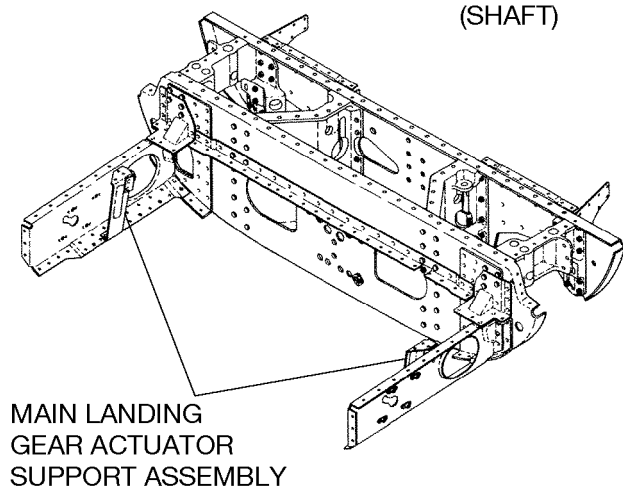


DETAIL A

MAIN GEAR ROTARY ACTUATOR
 -21058461 THRU -21058715



DETAIL C



SPINDLE FLANGE
 (SHAFT)

SPINDLE GEAR
 (SECTOR)

PISTON RACK

DETAIL A

MAIN GEAR ROTARY ACTUATOR
 -21057001 THRU -21058460

DETAIL B

NOTE: INSPECT CONTINUOUS EDGE OF SADDLE-SHAPED HOLE AND CIRCUMFERENCE OF 2.28 INCHES DIAMETER BORE AND STEP.

MAIN LANDING GEAR RETRACTION SYSTEM INSPECTION
 Figure 1 (Sheet 1)

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 C0541T1005