

CESSNA AIRCRAFT COMPANY
MODEL R182 SERIES
SERVICE MANUAL

SUPPLEMENTAL INSPECTION NUMBER: 32-10-01

1. **TITLE:**
Main Landing Gear Retraction System Inspection

2. **EFFECTIVITY**
R18200001 thru R18202032, FR18200001 thru FR18200070.

INSPECTION COMPLIANCE

ALL USAGE:	INITIAL	3,000 Hours	or	10 Years (NOTE)
	REPEAT	500 Hours	or	5 Years (NOTE)

NOTE: Refer to Note 1, Section 2A-14-00.

3. **PURPOSE**
To ensure structural integrity of the main landing gear retraction system.

4. **INSPECTION INSTRUCTIONS**

- A. Check airplane records to verify that the latest revision of SEB90-1 has been accomplished. If not, complete the latest revision of SEB90-1 with this inspection.
- B. Check airplane records to verify that SEB01-2 has been accomplished. If not, complete SEB01-2 with this inspection.
- C. Remove seats and carpeting as required to gain access to the plate above the actuator.
- D. Inspect the actuator, the actuator support assembly and the rod end bearings for cracks and corrosion.
 - (1) Clean area before inspecting if grime or debris is present.
 - (2) Conduct a surface eddy current inspection for cracks in the location indicated by Figure 1, Detail E on both sides of the main landing gear actuator body. Refer to Section 2A-13-01, Nondestructive Inspection Methods and Requirements, Eddy Current Inspection - (Surface Inspection), for additional instructions.
- E. Inspect the actuator pins, gears and sectors for cracks and wear.
- F. Refer to Figure 1, Details B, C, and D. Inspect the main landing gear pivot fittings for cracks and corrosion.
 - (1) Clean area before inspecting if grime or debris is present.
- G. Inspect the downlock support assembly for cracks or corrosion.
 - (1) Clean area before inspecting if grime or debris is present.
- H. Install parts removed for access.

5. **ACCESS AND DETECTABLE CRACK SIZE**

ACCESS/LOCATION

Main Landing Gear

DETECTABLE CRACK SIZE

Not Allowed

6. **INSPECTION METHOD**
Visual and Eddy Current

7. **REPAIR/MODIFICATION**

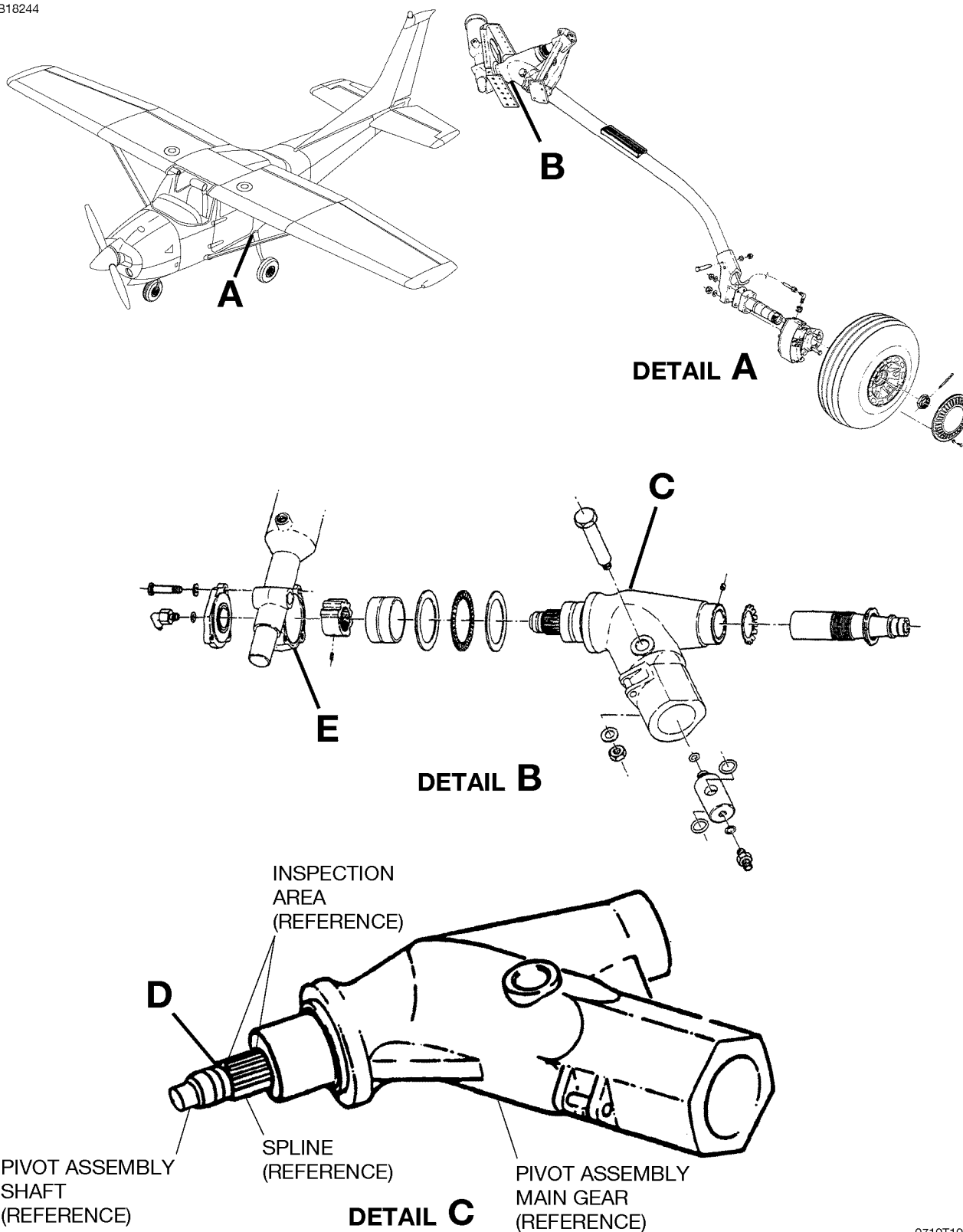
- A. If cracks are detected, the actuator or actuator body must be replaced.
- B. If no cracks are detected, rework in accordance with SEB01-2 Rev. 2 or latest revision.

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8. COMMENTS

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B18244



0710T1001

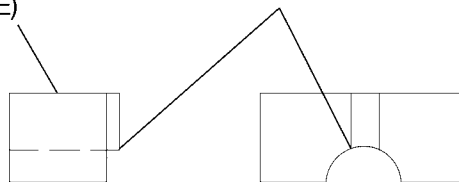
MAIN LANDING GEAR RETRACTION SYSTEM INSPECTION
 Figure 1 (Sheet 1)

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B18245

1820085-1
DEBURR TOOL
(REFERENCE)

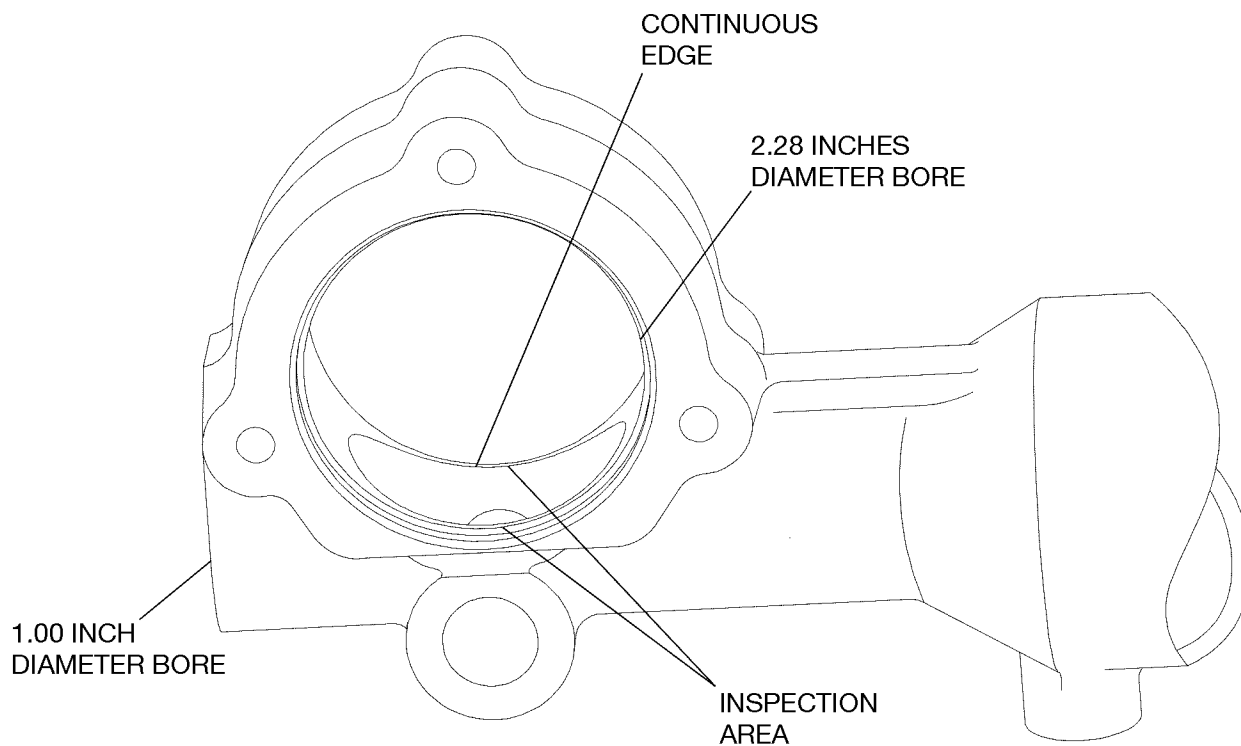
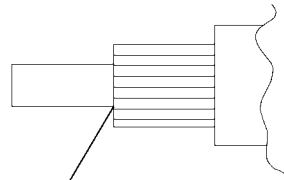
CUTTING
EDGE



DEBURR THIS RADIUS
BEFORE INSPECTION
IF REQUIRED

DETAIL D

(PIVOT ASSEMBLY SHAFT AND
SPLINE WITH BUSHING REMOVED)



DETAIL E

NOTE: INSPECT CONTINUOUS EDGE OF SADDLE-SHAPED HOLE AND CIRCUMFERENCE
OF 2.28 INCHES DIAMETER BORE AND STEP.

D0541T1006
E0541T1005

MAIN LANDING GEAR RETRACTION SYSTEM INSPECTION
Figure 1 (Sheet 2)