


EASA	AIRWORTHINESS DIRECTIVE	
	<p>AD No.: 2013-0310</p> <p>Date: 20 December 2013</p> <p>Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.</p>	
<p>This AD is issued in accordance with EU 748/2012, Part 21.A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].</p>		
<p>Design Approval Holder's Name: AIRBUS</p>	<p>Type/Model designation(s): A319, A320 and A321 aeroplanes</p>	
<p>TCDS Number:</p>	<p>EASA.A.064</p>	
<p>Foreign AD:</p>	<p>Not applicable</p>	
<p>Supersedure:</p>	<p>None</p>	
<p>ATA 53</p>	<p>Fuselage – Tack and Rivet Holes in Fitting Brackets at Frame 50 to Frame 63 – Inspection / Repair</p>	
<p>Manufacturer(s):</p>	<p>Airbus (formerly Airbus Industrie)</p>	
<p>Applicability:</p>	<p>Airbus A319-111, A319-112, A319-113, A319-114, A319-115, A319-131, A319-132, A319-133, A320-211, A320-212, A320-214, A320-215, A320-216, A320-231, A320-232, A320-233, A321-111, A321-112, A321-131, A321-211, A321-212, A321-213, A321-231 and A321-232 aeroplanes, all manufacturer serial numbers.</p>	
<p>Reason:</p>	<p>During a full scale fatigue test, several broken frames in the cargo compartment area between Frame (FR) 50 and FR 63, have been found, especially on the cargo floor support fittings and open tack holes on left hand side.</p> <p>This condition, if not detected and corrected, could affect the structural integrity of the aeroplane.</p> <p>For the reason described above, this AD requires repetitive inspections of the frames in the cargo compartment area and of the cargo floor support fittings and open tack holes on the left hand (LH) side and, depending on findings, the accomplishment of applicable corrective action(s). This AD also requires a modification, which constitutes terminating action for the repetitive inspections required by this AD.</p>	
<p>Effective Date:</p>	<p>03 January 2014</p>	

<p>Required Action(s) and Compliance Time(s):</p>	<p>Required as indicated, unless accomplished previously:</p> <p>(1) Within the compliance time(s) defined in Table 1 of this AD, as applicable, and, thereafter, at intervals not to exceed 5 000 flight cycles (FC) or 10 000 flight hours (FH), whichever occurs first, accomplish a rototest inspection of open tack holes and rivet holes at the cargo floor support fittings between frame (FR) 50 and FR 63 (LH side only) for A320/A321 aeroplanes, and between FR 53 and FR 63 (LH side only) for A319 aeroplanes, as applicable, in accordance with the instructions of Airbus SB A320-53-1257.</p> <p style="text-align: center;">Table 1 - Initial inspection</p> <table border="1" data-bbox="544 495 1461 965"> <thead> <tr> <th data-bbox="544 495 1023 600">Time accumulated on the effective date of this AD (whichever occurs first since aeroplane first flight)</th> <th data-bbox="1023 495 1461 600">Compliance Time (whichever occurs first)</th> </tr> </thead> <tbody> <tr> <td data-bbox="544 600 1023 680">Equal to or more than 45 000 FC or 90 000 FH</td> <td data-bbox="1023 600 1461 680">Within 1 000 FC or 2 000 FH after the effective date of this AD</td> </tr> <tr> <td data-bbox="544 680 1023 853">Equal to or more than 36 200 FC or 72 400 FH, but less than 45 000 FC or 90 000 FH</td> <td data-bbox="1023 680 1461 853">Within 2 000 FC or 4 000 FH after the effective date of this AD, but not exceeding 46 000 FC or 92 000 FH since aeroplane first flight</td> </tr> <tr> <td data-bbox="544 853 1023 965">Less than 36 200 FC or 72 400 FH</td> <td data-bbox="1023 853 1461 965">Before exceeding 38 200 FC or 76 400 FH since aeroplane first flight</td> </tr> </tbody> </table> <p>(2) If, during any inspection as required by paragraph (1) of this AD, a crack is detected, before next flight, contact Airbus for approved repair instructions and accomplish those instructions accordingly.</p> <p>(3) Before exceeding 48 000 FC or 96 000 FH, whichever occurs first since first flight, modify the aeroplane in accordance with the instructions of Airbus SB A320-53-1261.</p> <p>(4) Modification of an aeroplane as required by paragraph (3) of this AD constitutes terminating action for the repetitive inspections required by paragraph (1) of this AD for that aeroplane.</p>	Time accumulated on the effective date of this AD (whichever occurs first since aeroplane first flight)	Compliance Time (whichever occurs first)	Equal to or more than 45 000 FC or 90 000 FH	Within 1 000 FC or 2 000 FH after the effective date of this AD	Equal to or more than 36 200 FC or 72 400 FH, but less than 45 000 FC or 90 000 FH	Within 2 000 FC or 4 000 FH after the effective date of this AD, but not exceeding 46 000 FC or 92 000 FH since aeroplane first flight	Less than 36 200 FC or 72 400 FH	Before exceeding 38 200 FC or 76 400 FH since aeroplane first flight
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<p>Ref. Publications:</p>	<p>Airbus SB A320-53-1257 original Issue dated 21 December 2012.</p> <p>Airbus SB A320-53-1261 original Issue dated 21 December 2012, or Revision 01 [to be issued].</p> <p>The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.</p>								
<p>Remarks:</p>	<ol style="list-style-type: none"> <li data-bbox="496 1541 1449 1603">If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD. <li data-bbox="496 1619 1471 1742">This AD was posted on 06 February 2013 as PASD 13-030 for consultation until 06 March 2013, and re-published on 07 August 2013 as PAD 13-030R1 for additional consultation until 04 September 2013. The Comment Response Document can be found at http://ad.easa.europa.eu. <li data-bbox="496 1758 1398 1821">Enquiries regarding this AD should be referred to the Safety Information Section, Executive Directorate, EASA. E-mail: ADs@easa.europa.eu. <li data-bbox="496 1836 1457 1926">For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS - Airworthiness Office - EIAS. Fax +33 5 61 93 44 51. E-mail: account.airworth-eas@airbus.com. 								