EASA AD No.: 2015-0035-E

EASA

EMERGENCY AIRWORTHINESS DIRECTIVE

AD No.: 2015-0035-E

Date: 27 February 2015

Note: This Emergency Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.

This AD is issued in accordance with EU 748/2012, Part 21.A.3B. In accordance with EU 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EU 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].

Design Approval Holder's Name:		Type/Model designation(s):	
AGUSTAWESTLAND S.p.A.		A109, AW109, A119 and AW119 helicopters	
TCDS Number:	EASA.R.005		
Foreign AD:	Not applicable		
Supersedure:	None		
ATA 64	Tail Rotor – Pitch Control Link Assembly – Inspection / Replacement		
Manufacturer(s):	AgustaWestland S.p.A (formerly Agusta S.p.A.), AgustaWestland Philadelphia Corporation (formerly Agusta Aerospace Corporation).		
Applicability:	A109A, A109AII, A109C, A109E, A109K2, A109LUH, A109S, AW109SP, A119 and AW119MKII helicopters, all serial numbers.		
Reason:	An event was reported of in-flight breaking of the tail rotor (TR) pitch control link assembly Part Number (P/N) 109-0130-05-117 on an AW119MkII helicopter.		
	Further investigation determined excessive friction of one of the spherical bearings of the link as the most probable cause for the seizure. This may have been caused by incorrect installation when the affected bearing was replaced during overhaul a few flight hours (FH) before the occurrence.		
	This condition, if not detected and corrected, could lead to further cases of breaking of the TR pitch control link assembly, possibly resulting in loss of control of the helicopter.		
	To address this unsafe condition, AgustaWestland issued instructions to inspect of the spherical bearings installed on the TR pitch control link assembly.		
	For the reason described above, this AD requires two one-time inspections to detect freedom of movement of the spherical bearings installed on recently overhauled TR pitch control link assemblies P/N 109-0130-05-117 (one inspection with link installed on helicopter, the other with link removed) and, in case discrepancies are found, corrective action(s).		
	This AD is considered	to be an interim action and further AD action may follow.	
Effective Date:	03 March 2015		

EASA AD No.: 2015-0035-E

Required Action(s)	Required as indicated, unless accomplished previously:	
and Compliance Ó Time(s):	This AD requires actions on each overhauled TR pitch control link assembly, identified by P/N 109-0130-05-117, having accumulated (on the effective date of this AD) less than 100 flight hours (FH) since overhaul, and those for which the accumulated FH since last overhaul cannot be determined.	
	(1) Before next flight after the effective date of this AD, inspect the spherical bearings in accordance with the instructions of <u>Part I</u> (link remains installed on helicopter) of AgustaWestland Bolletino Tecnico (BT) 109-145 Rev. A, BT 109EP-141 Rev. A, BT 109K-65 Rev. A, BT 109L-079 Rev. A, BT 109S-065 Rev. A, BT109SP-087 Rev. A or BT 119-072 Rev. A, as applicable to helicopter model.	
	(2) Within 5 FH after the effective date of this AD, re-inspect the spherical bearings in accordance with the instructions of <u>Part II</u> (link to be removed from helicopter) of AgustaWestland BT 109-145 Rev. A, BT 109EP-141 Rev. A, BT 109K-65 Rev. A, BT 109L-079 Rev. A, BT 109S-065 Rev. A, BT109SP-087 Rev. A or BT 119-072 Rev. A, as applicable to helicopter model.	
	(3) If, during an inspection as required by paragraph (1) or (2) of this AD, any discrepancy is found, before next flight, accomplish the applicable corrective action(s) in accordance with the instructions of AgustaWestland BT 109-145 Rev. A, BT 109EP-141 Rev. A, BT 109K-65 Rev. A, BT 109L-079 Rev. A, BT 109S-065 Rev. A, BT109SP-087 Rev. A or BT 119-072 Rev. A, as applicable to helicopter model.	
	(4) From the effective date of this AD, installation on a helicopter of an overhauled TR pitch control link assembly P/N 109-0130-05-117 is allowed, provided that, prior to installation, the assembly has passed an inspection in accordance with the instructions of Part II of AgustaWestland BT 109-145 Rev. A, BT 109EP-141 Rev. A, BT 109K-65 Rev. A, BT 109L-079 Rev. A, BT 109S-065 Rev. A, BT109SP-087 Rev. A or BT 119-072 Rev. A, as applicable to helicopter model.	
Ref. Publications:	AgustaWestland BT 109-145 Rev. A, dated 27 February 2015.	
	AgustaWestland BT 109EP-141 Rev. A, dated 27 February 2015.	
	AgustaWestland BT 109K-65 Rev. A, dated 27 February 2015.	
	AgustaWestland BT 109L-079 Rev. A, dated 27 February 2015.	
	AgustaWestland BT 109S-065 Rev. A, dated 27 February 2015.	
	AgustaWestland BT 109SP-087 Rev. A, dated 27 February 2015.	
	AgustaWestland BT 119-072 Rev. A, dated 27 February 2015.	
	The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.	
Remarks:	If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.	
	 The results of the safety assessment have indicated the need for immediate publication and notification, without the full public consultation process. 	
	3. Enquiries regarding this AD should be referred to the Safety Information Section, Certification Directorate, EASA. E-mail: ADs@easa.europa.eu .	
	 For any question concerning the technical content of the requirements in this AD, please contact: Agusta S.p.A. Customer Support, Via del Gregge, 100, 21015 Lonate Pozzolo (VA) – Italy Telephone: +39 0331 664600; Fax: +39 0331 664684 E-mail: AW109.mbx@agustawestland.com. 	