



Airworthiness Directive

AD No.: 2015-0215R1

Issued: 29 October 2015

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) 216/2008 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EC) 216/2008, Article 14(4) exemption].

Design Approval Holder's Name:

AIRBUS

Type/Model designation(s):

A380 aeroplanes

Effective Date: 06 November 2015 [same as original AD]

TCDS Number(s): EASA.A.110

Foreign AD: Not applicable

Revision: This AD revises EASA AD 2015-0215 dated 23 October 2015.

ATA 53 – Fuselage – Horizontal Tail Plane Pivot Fittings – Inspection

Manufacturer(s):

Airbus

Applicability:

Airbus A380-841, A380-842 and A380-861 aeroplanes, all manufacturer serial numbers, except those on which Airbus modification (mod) 75897 has been embodied in production.

Reason:

During a “health check” inspection, as part of a campaign focused on detecting fatigue damage, cracked sealant was found around the bushing installed on the outboard horizontal tail plane (HTP) pivot fitting. Subsequent investigation confirmed that the affected crack on sealant was a result of migration of bushings on Section 19 pivot fittings.

This condition, if not detected and corrected, could affect the structural integrity of the HTP, possibly resulting in reduced control of the aeroplane.

To address this unsafe condition, Airbus issued Alert Operators Transmission (AOT) A53R008-14, providing inspection instructions for aeroplanes not fitted with mechanical bush retainers on the left-hand (LH) and right-hand (RH) pivot fittings outboard side, and to install a bushing retainer as a repair. Airbus also issued Service Bulletin (SB) A380-53-8089, to provide inspection instructions.



Airbus also introduced in manufacturing process a mechanical retainer installation for aeroplanes manufactured after the issuance of AOT A53R008-14. The mechanical retainer installation is also embodied in aeroplanes modified in accordance with Airbus mod 74864.

Consequently, EASA issued AD 2015-0215 to require repetitive detailed inspections (DET) of the pivot fittings and surrounding parts and, depending on findings, accomplishment of applicable corrective action(s). That AD also required repetitive DET of each affected hinge pin assembly to check the tightening torque of nuts.

Since that AD was issued, it was determined that Airbus had issued additional Technical Adaptations (TA) applicable to aeroplanes which were equipped with a bush retainer. Additionally, it was recognised that an Airbus modification installing a mechanical retainer was incorrectly referred in the Reason paragraph.

For the reasons described above, this AD is revised to update Appendix 1 of this AD referring to additional Airbus TA's and to correct a reference of the Airbus modification in Reason paragraph.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

- (1) Within the compliance time as defined in Table 1 of this AD, depending on aeroplane configuration, and, thereafter, at intervals as defined in Airbus SB A380-53-8089, accomplish the following actions concurrently, as specified in paragraphs (1.1) and (1.2) of this AD in accordance with the instructions of Airbus SB A380-53-8089:
 - (1.1) A DET of the HTP pivot fitting (outboard and inboard) bush area, LH and RH side.
 - (1.2) A DET of each hinge pin assembly to check the tightening torque of each hinge pin nut on LH and RH side.

Table 1 – Initial DET

Aeroplane Configuration	Compliance Time
Pre-mod 74864 and Pre-AOT A53R008-14	Whichever occurs later, A or B : A : Before exceeding 300 FC or 2 200 flight hours (FH), whichever occurs first since the aeroplane first flight. B : Within 50 FC after the effective date of this AD.
Pre-mod 74864 and Post-AOT A53R008-14	Within 1 500 FH after installation of retainers in accordance with Airbus AOT A53R008-14, or Airbus Technical Documentation (TD) td_g15_s4_00151_2014, as applicable.
Post-mod 74864	Within 1 200 FC or 8 900 FH, whichever occurs first since the aeroplane first flight.

- (2) If, during any DET as required by paragraph (1) of this AD, any discrepancy is detected, within the compliance time and in accordance with instructions of Airbus SB A380-53-8089,



depending on finding(s), accomplish all the applicable corrective action(s), including repair(s), additional inspections, and follow-on repetitive inspections, as applicable.

- (3) Inspections and corrective actions, accomplished before the effective date of this AD in accordance with the instructions of Airbus TA as specified in Appendix 1 of this AD are acceptable to comply with the requirements of paragraphs (1) and (2) of this AD for that aeroplane. From the effective date of this AD, instructions of Airbus SB A380-53-8089 must be used.
- (4) Accomplishment of corrective action(s) on an aeroplane, as required by paragraph (2) of this AD, does not constitute terminating action for the repetitive DET as required by paragraphs (1) of this AD for that aeroplane.

Ref. Publications:

Airbus AOT A53R008-14 original issue dated 25 November 2014.

Airbus SB A380-53-8089 original issue dated 21 October 2015.

The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. Based on the required actions and the compliance time, EASA have decided to issue a Final AD with Request for Comments, postponing the public consultation process until after publication.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – Airworthiness Office – EIANA; Telephone +33 562 110 253. Fax +33 562 110 307
E-mail account.airworth-A380@airbus.com.



Appendix 1 – Airbus TA

Airbus TA	Applicable to MSN
TA_80011602_003-2014-1	00003, 00005, 00006, 00007, 00008, 00009, 00010, 00011, 00012, 00013, 00014, 00015, 00016, 00017, 00019, 00020, 00021, 00022, 00023, 00025, 00026, 00027, 00028, 00029, 00030, 00031, 00033, 00034, 00035, 00036, 00038, 00039, 00040, 00041, 00042, 00043, 00044, 00045, 00046, 00047, 00048, 00049, 00050, 00051, 00052, 00054, 00055, 00056, 00057, 00058, 00059, 00061, 00062, 00063, 00064, 00065, 00066, 00067, 00068, 00069, 00070, 00071, 00072, 00073, 00074, 00075, 00076, 00077, 00078, 00079, 00080, 00081, 00082, 00083, 00084, 00085, 00086, 00087, 00088, 0089, 00090, 00092, 00093, 00094, 00095, 00096, 00098, 00099, 00100, 00101, 00103, 00105, 00106, 00107, 00108, 00109, 00110, 00111, 00112, 00113, 00114, 00115, 00116, 00117, 00119, 00120, 00121, 00122, 00123, 00124, 00125, 00126, 00127, 00128, 00130, 00131, 00132, 00133, 00134, 00135, 00136, 00137, 00138, 00139, 00140, 00141, 00142, 00144, 00146, 00147, 00148, 00149, 00150, 00151, 00152, 00153, 00154, 00155, 00156, 00157, 00158, 00159, 00160, 00161, 00163 and 00171
TA-80039994_002_2015-1	00003, 00005, 00006, 00007, 00008, 00009, 00010, 00011, 00012, 00013, 00014, 00015, 00016, 00017, 00019, 00020, 00021, 00022, 00023, 00025, 00026, 00027, 00028, 00029, 00030, 00031, 00033, 00034, 00035, 00036, 00038, 00039, 00040, 00041, 00042, 00043, 00044, 00045, 00046, 00047, 00048, 00049, 00050, 00051, 00052, 00054, 00055, 00056, 00057, 00058, 00059, 00061, 00062, 00063, 00064, 00065, 00066, 00067, 00068, 00069, 00070, 00071, 00072, 00073, 00074, 00075, 00076, 00077, 00078, 00079, 00080, 00081, 00082, 00083, 00084, 00085, 00086, 00087, 00088, 0089, 00090, 00092, 00093, 00094, 00095, 00096, 00098, 00099, 00100, 00101, 00103, 00105, 00106, 00107, 00108, 00109, 00110, 00111, 00112, 00113, 00114, 00115, 00116, 00117, 00119, 00120, 00121, 00122, 00123, 00124, 00125, 00126, 00127, 00128, 00130, 00131, 00132, 00133, 00134, 00135, 00136, 00137, 00138, 00139, 00140, 00141, 00142, 00144, 00146, 00147, 00148, 00149, 00150, 00151, 00152, 00153, 00154, 00155, 00156, 00157, 00158, 00159, 00160, 00161, 00163 and 00171
TA-80028753_007_2015	00117
TA-80031894_004_2015	00155
TA-80013852_029_2014	00160

