



Emergency Airworthiness Directive

AD No.: 2016-0059-E

Issued: 22 March 2016

Note: This Emergency Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) 216/2008 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EC) 216/2008, Article 14(4) exemption].

Design Approval Holder's Name:

AIRBUS HELICOPTERS

Type/Model designation(s):

SA 330 J helicopters

Effective Date: 24 March 2016

TCDS Number(s): EASA.R.002

Foreign AD: Not applicable

Supersedure: This AD supersedes DGAC France AD 87-032-052(B)R3 dated 23 January 1991.

ATA 64 – Tail Rotor – Metal Blades / Skin – Inspection

Manufacturer(s):

Airbus Helicopters (formerly Eurocopter, Eurocopter France, Aerospatiale)

Applicability:

SA 330 J helicopters, all manufacturer serial numbers equipped with metal blades P/N 330A12-0005, all dash numbers (non de-iced blades) or 330A12-0006, all dash numbers (de-iced blades) on the tail rotor.

Reason:

Two occurrences of cracking of metal tail rotor blade skin P/N 330A12-0005 have been reported. These cracks subsequently led to rotor blade vibrations and forced landing of the helicopter.

This condition, if not detected and corrected, could lead to additional occurrences of TR blade structural damage, possibly resulting in significant vibrations and reduced control of the helicopter.

To address this potentially unsafe condition, Airbus Helicopters issued Alert Service Bulletin (ASB) SA330-05.101 to provide inspection instructions, superseding Aerospatiale SA330 SB 05-71 Revision 4 (originally referenced in DGAC France AD 87-032-052(B)R3) and extending the area affected by the inspection.



For the reasons described above, this AD retains the requirements of DGAC France AD 87-032-052(B)R3, which is superseded, but requires those action(s) in accordance with an improved service instructions.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

- (1) Within the compliance time specified in Table 1 of this AD, as applicable, and, thereafter, at intervals not to exceed the value as specified in Table 1 of this AD, as applicable, inspect each tail rotor blade in accordance with the instructions of paragraph 3 of Airbus Helicopters ASB SA330-05.101. The compliance time for the initial inspection must be counted from the latest inspection in accordance with Aerospatiale SA330 SB 05-71 Revision 4, as previously required by DGAC France AD 87-032-052(B)R3.

Table 1 – Tail Rotor Blade Inspections

Metal Tail Rotor Blade	Compliance Time flight hours (FH)	Inspection Interval, FH
P/N 330A12-0006, all dash numbers (with a de-icing system)	15	15
P/N 330A12-0005, all dash numbers (without a de-icing system)	30	30

- (2) If, during any inspection as required by paragraph (1) of this AD, any discrepancy is detected, as described in Airbus Helicopters ASB SA330-05.101, before next flight, accomplish the applicable corrective action(s) in accordance with the instructions of Airbus Helicopters ASB SA330-05.101.
- (3) From the effective date of this AD, installation of a metal tail rotor blade P/N 330A12-0006 or P/N 330A12-0005 is allowed, provided that, before installation, the blade has passed an inspection in accordance with the instructions of paragraph (3) of Airbus Helicopters ASB SA330-05.101 and that, following installation, the blade is inspected as required by this AD.

Ref. Publications:

Airbus Helicopters ASB SA330-05.101 original issue, dated 21 March 2016.

The use of later approved revisions of this document is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. The results of the safety assessment have indicated the need for immediate publication and notification, without the full consultation process.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.



4. For any question concerning the technical content of the requirements in this AD, please contact:
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