



Airworthiness Directive

AD No.: 2016-0196

Issued: 30 September 2016

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) 216/2008 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EC) 216/2008, Article 14(4) exemption].

Design Approval Holder's Name:

AIRBUS

Type/Model designation(s):

A310 aeroplanes

Effective Date: 14 October 2016

TCDS Number(s): EASA.A.172

Foreign AD: Not applicable

Supersedure: This AD supersedes EASA AD 2009-0057 dated 13 March 2009.

ATA 57 – Wings – Centre Wing Trellis Boom Drainage Holes & Frame 40 Upper Corner Fitting – Inspection / Repair

Manufacturer(s):

Airbus (formerly Airbus Industrie)

Applicability:

A310-203, A310-203C, A310-204, A310-221, A310-222, A310-304, A310-308, A310-322, A310-324 and A310-325 aeroplanes, all serial numbers.

Note 1: The detailed applicability for each of the actions required by this AD is identified in each relevant "Action" paragraph of the 'Required Action(s) and Compliance Time(s)' section of this AD.

Reason:

DGAC France issued AD F-1992-106-132R7 to require certain inspections and modifications which addressed JAR/FAR 25-571 requirements, related to damage-tolerance and fatigue evaluation of structure. Following the Extended Design Service Goal activities as part of the Structure Task Group for the Airbus A310 program, EASA published AD 2007-0053, which replaced DGAC France AD F-1992-106-132R7.

After EASA issued AD 2007-0053R1, the thresholds and the intervals of Airbus Service Bulletins (SB) A310-57-2050 and A310-57-2064 were updated, prompting EASA to issue AD 2009-0057 and AD



2007-0053 was revised (R2) accordingly. EASA AD 2009-0057 also required the accomplishment of the actions specified in Airbus SB A310-57-2048 at Revision 01.

After EASA issued AD 2009-0057, in the frame of the Widespread Fatigue Damage campaign, new analysis has indicated the need for additional work included in Revision 03 of Airbus SB A310-57-2050.

For the reason described above, this new AD retains the requirements of EASA AD 2009-0057, which is superseded, and requires inspection and corrective actions as specified in Airbus SB A310-57-2050 Revision 04.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

Action No. 1: Cold working of Trellis Boom Drainage Holes: Applicable to A310-203, A310-203C, A310-204, A310-222, A310-304, A310-322 and A310-324 aeroplanes, all serial numbers, except those aeroplanes on which:

- Airbus modification 06130 was embodied before delivery, or
- Airbus SB A310-57-2048 has been embodied in service, or
- any rework of cracked drain holes was previously performed in accordance with the instructions of Airbus SB A310-57-2050, as applicable.

- (1) Before exceeding the applicable threshold(s) defined in Airbus SB A310-57-2048 Revision 01, accomplish the cold working of trellis boom drainage holes in accordance with the instructions of Airbus SB A310-57-2048 Revision 01.

Action No. 2: Inspection of Trellis Boom Drainage Holes: Applicable to A310-203, A310-203C, A310-204, A310-221, A310-222, A310-304, A310-308, A310-322, A310-324 and A310-325 aeroplanes, all serial numbers.

- (2) Before exceeding the applicable threshold(s) defined in Airbus SB A310-57-2050 Revision 04, accomplish the initial inspection of the trellis boom drainage holes and, depending on findings, accomplish the applicable corrective action(s) within the time period defined in, and in accordance with the instructions of, Airbus SB A310-57-2050 Revision 04.

Aeroplanes, which have exceeded the applicable inspection threshold value(s) must be inspected within the grace periods defined in Airbus SB A310-57-2050 Revision 04, taking the effective date of this AD as a starting date, and without exceeding the previous values specified in Airbus SB A310-57-2050 Revision 03.

- (3) Thereafter, within the interval defined in Airbus SB A310-57-2050 Revision 04, repeat the inspection in accordance with the instructions of Airbus SB A310-57-2050 Revision 04 and, depending on findings, accomplish the applicable corrective action(s) within the time period defined in, and in accordance with the instructions of, Airbus SB A310-57-2050 Revision 04.



For the first repeat inspection to be accomplished after the effective date of this AD, a grace period can be applied, without exceeding the applicable interval value specified in Airbus SB A310-57-2050 at Revision 04.

Action No. 3: Inspection of Fuselage Frame 40 Upper Corner Fitting: Applicable to A310-203, A310-203C, A310-204, A310-221, A310-222, A310-304, A310-308, A310-322, A310-324 and A310-325 aeroplanes, all serial numbers.

- (4) Before exceeding the applicable threshold(s) defined in Airbus SB A310-57-2064 Revision 02, and, thereafter, within the applicable interval(s) as defined in Airbus SB A310-57-2064 Revision 02, inspect the frame 40 upper corner fitting left hand and right hand sides and, depending on findings, accomplish the applicable corrective action(s) within the time period defined in, and in accordance with the instructions of, Airbus SB A310-57-2064 Revision 02.

Aeroplanes, which have exceeded the applicable inspection threshold value(s) must be inspected within the grace period defined in Airbus SB A310-57-2064 Revision 02, taking 27 March 2009 as a starting date (i.e.: AD 2009-0057 effective date), and without exceeding the previous value(s) specified in Airbus SB A310-57-2064 at Revision 01.

Note 2: For aeroplanes previously inspected in accordance with the instructions of Airbus SB A310-57-2064 original issue or Revision 01, no further action is required by paragraph (4) of this AD.

- (5) Thereafter, within the interval defined in Airbus SB A310-57-2064 Revision 02, repeat the inspection in accordance with the instructions of Airbus SB A310-57-2064 Revision 02 and, depending on findings, accomplish the applicable corrective action(s) within the time period defined in, and in accordance with the instructions of, Airbus SB A310-57-2064 Revision 02.

For the first repeat inspection to be accomplished after the effective date of this AD, a grace period can be applied, without exceeding the applicable interval value specified in Airbus SB A310-57-2064 at Revision 01.

- (6) The accomplishment of corrective action(s) on an aeroplane as required by paragraph (2), (3) (4) or (5) of this AD, as applicable, does not constitute terminating action for the repetitive inspections required by this AD for that aeroplane.

Ref. Publications:

Airbus SB A310-57-2048 original issue dated 29 June 1990, or Revision 01 dated 22 May 2007.

Airbus SB A310-57-2050 original issue dated 23 April 1990, or Revision 01 dated 22 May 2007, or Revision 02 dated 27 August 2009, or Revision 03 dated 19 December 2014, or Revision 04 dated 13 March 2015.

Airbus SB A310-57-2064 original issue dated 24 August 1995, or Revision 01 dated 05 January 2001, or Revision 02 dated 21 December 2007.

The use of a later approved revision of these documents is acceptable for compliance with the requirements of this AD.



Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. This AD was posted on 30 August 2016 as PAD 16-124 for consultation until 27 September 2016. No comments were received during the consultation period.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – EIAW (Airworthiness Office)
E-mail: continued.airworthiness-wb.external@airbus.com.

