



## Emergency Airworthiness Directive

**AD No.:** 2020-0224-E

**Issued:** 16 October 2020

Note: This Emergency Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

### Design Approval Holder's Name:

AIRBUS HELICOPTERS

### Type/Model designation(s):

AS 350 and AS 355 helicopters

**Effective Date:** 20 October 2020

**TCDS Number(s):** EASA.R.008 and EASA.R.146

**Foreign AD:** Not applicable

**Supersedure:** None

## ATA 64 – Tail Rotor – Tail Rotor Blades – Inspection / Replacement / Modification

### Manufacturer(s):

Airbus Helicopters (AH), formerly Eurocopter, Eurocopter France, Aerospatiale

### Applicability:

AS 350 B, AS 350 BA, AS 350 BB, AS 350 B1, AS350 B2 and AS 350 D helicopters, all serial numbers (s/n); and

AS 355 E, AS 355 F, AS 355 F1, AS 355 F2 and AS 355 N helicopters, all s/n.

### Definitions:

For the purpose of this AD, the following definitions apply:

**Affected part:** Pre-mod 075580 tail rotor blades (TRB), having a Part Number (P/N) as listed in Appendix 1, Table 1, of this AD.

**Serviceable part:** Post-mod 075580 TRB, having a P/N as listed in Appendix 1, Table 2, of this AD.

**The applicable ASB:** AH AS 350 Emergency Alert Service Bulletin (ASB) 05.00.99 and AS 355 ASB 05.00.84, as applicable (single document).



**Groups:** Group 1 helicopters are those that have at least one affected part installed. Group 2 helicopters are those that do not have an affected part installed.

**Reason:**

Two occurrences have recently been reported of finding large debonding of TRB leading edge (LE) protection shields. The design and assembly procedure of the affected part is such that rapid debonding can occur if humidity/liquid water reaches the bonding surface between LE and blade spar.

This condition, if not detected and corrected, could lead to loss of tail rotor control, possibly resulting in loss of control of the helicopter.

To address this potential unsafe condition, AH published the applicable ASB to provide inspection instructions.

For the reason described above, this AD requires repetitive inspections of the affected parts and, depending on findings, replacement. This AD also requires modification of the helicopter by installation of serviceable parts, and prohibits (re)installation of affected parts.

**Required Action(s) and Compliance Time(s):**

Required as indicated, unless accomplished previously:

**Repetitive Inspection(s):**

- (1) For Group 1 helicopters: Before next flight after the effective date of this AD, and, thereafter, before each flight, visually inspect each TRB LE protection shield in accordance with the instructions of paragraph 3.B.1 of the applicable ASB.
- (2) For Group 1 helicopters: Within 30 flight hours (FH) after the effective date of this AD or within 100 FH since last tapping inspection of the TRB LE, whichever occurs first, and, thereafter, at intervals not to exceed 30 FH, accomplish a tapping inspection of the TRB LE in accordance with the instructions of paragraph 3.B.1 of the applicable ASB.

**Corrective Action(s):**

- (3) If, during any inspection as required by paragraph (1) or (2) of this AD, discrepancies, as defined in the applicable ASB, are detected on an affected part, before next flight, replace that affected part with a serviceable part.

**Parts Installation:**

- (4) For Group 1 and Group 2 helicopters: From the effective date of this AD, do not install on any helicopter an affected part, as defined in this AD.

**Modification:**

- (5) For Group 1 helicopters: Within 14 months after the effective date of this AD, replace each affected part with a serviceable part in accordance with the instructions of the applicable ASB.



**Terminating Action:**

- (6) Replacement of all affected parts on a helicopter in accordance with the instructions of the applicable ASB constitutes terminating action for the repetitive inspections as required by paragraph (1) and (2) of this AD for that helicopter.
- (7) Modification of a helicopter as required by paragraph (5) of this AD constitutes terminating action for the repetitive inspections as required by paragraph (1) and (2) of this AD for that helicopter.

**Ref. Publications:**

AH AS350 Emergency ASB 05.00.99 original issue date 14 October 2020.

AH AS355 Emergency ASB 05.00.84 original issue date 14 October 2020.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

**Remarks:**

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. The results of the safety assessment have indicated the need for immediate publication and notification, without the full consultation process.
3. Enquiries regarding this AD should be referred to the EASA Programming and Continued Airworthiness Information Section, Certification Directorate. E-mail: [ADs@easa.europa.eu](mailto:ADs@easa.europa.eu).
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
5. For any question concerning the technical content of the requirements in this AD, please contact: Airbus Helicopters (Technical Support), Aéroport de Marseille Provence 13725 Marignane Cedex, France, Telephone +33 (0)4 42 85 97 97, Fax +33 (0)4 42 85 99 66, Web portal: <https://keycopter.airbushelicopters.com> > Technical Requests Management, E-mail: [support.technical-dyncomp.ah@airbus.com](mailto:support.technical-dyncomp.ah@airbus.com), and [TechnicalSupport.Helicopters@airbus.com](mailto:TechnicalSupport.Helicopters@airbus.com).



## Appendix 1

Table 1 – Affected Parts (P/N)

350A120030.01, 350A120030.02, 350A120030.03, 350A120030.04, 350A120030.05, 350A120030.06, 350A120030.07, 350A120030.08, 350A120030.09 or 350A120030.10
355A120031.01, 355A120031.02, 355A120031.03, 355A120031.04, 355A120031.05, 355A120031.06, 355A120031.07, 355A120031.08, 355A120031.09 or 355A120031.10
355A120040.00, 355A120040.01, 355A120040.02, 355A120040.03, 355A120040.04, 355A120040.05, 355A120040.06, 355A120040.09 or 355A120040.12
355A120041.00, 355A120041.01, 355A120041.02, 355A120041.03, 355A120041.04, 355A120041.05, 355A120041.06, 355A120041.09 or 355A120041.12
355A120050.00, 355A120050.01, 355A120050.02, 355A120050.03, 355A120050.06 or 355A120050.07
355A120051.00, 355A120051.01, 355A120051.02, 355A120051.03, 355A120051.06 or 355A120051.07

Table 2 – Serviceable Parts (P/N)

350A120030.11, 350A120030.12, 350A120030.13, 350A120030.14 or 350A120030.15
355A120031.11, 355A120031.12, 355A120031.13, 355A120031.14, 355A120031.15, 355A120031.16, 355A120031.17 or 355A120031.18
355A120040.07, 355A120040.08, 355A120040.10, 355A120040.11, 355A120040.13, 355A120040.14 or 355A120040.15
355A120041.07, 355A120041.08, 355A120041.10, 355A120041.11, 355A120041.13, 355A120041.14 or 355A120041.15
355A120050.04, 355A120050.05, 355A120050.08, 355A120050.09, 355A120050.10, 355A120050.11, 355A120050.12 or 355A120050.13
355A120051.04, 355A120051.05, 355A120051.08, 355A120051.09, 355A120051.10, 355A120051.11, 355A120051.12 or 355A120051.13

