

## Airworthiness Directive

**AD No.:** 2021-0145**Issued:** 18 June 2021

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

**Design Approval Holder's Name:**

AIRBUS

**Type/Model designation(s):**

A380 aeroplanes

**Effective Date:** 02 July 2021**TCDS Number(s):** EASA.A.110**Foreign AD:** Not applicable**Supersedure:** None

### ATA 57 – Wings – Wing Tip Assembly – Modification

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**Manufacturer(s):**

Airbus

**Applicability:**

Airbus A380-841, A380-842 and A380-861 aeroplanes, all manufacturer serial numbers, except those that have embodied Airbus modification (mod) 76718 and mod 76719 in production.

**Definitions:**

For the purpose of this AD, the following definitions apply:

**The applicable SB:** Airbus Service Bulletin (SB) A380-57-8132 (for right-hand (RH) wing tip) and SB A380-57-8133 (for left-hand (LH) wing tip), as applicable.

**Aeroplane date of manufacture:** The date of transfer of title (ownership), which is referenced in Airbus documentation at the time of first delivery to an operator.

**Reason:**

Following investigation after an in-service event, it was established that parts manufactured from 7449 aluminium alloy in the wing tip assembly are susceptible to cracking.

This condition, if not corrected, could lead to crack initiation and propagation, possibly resulting in reduced structural integrity of the wing.

To address this potential unsafe condition, Airbus initially issued SB A380-57-8145 and SB A380-57-8146 as interim action and then developed a new wing tip assembly made from 7010 aluminium alloy, which is embodied in production through Airbus mod 76718 and mod 76719, and issued the applicable SB to provide modification instructions for in-service aeroplanes.

For the reasons described above, this AD requires modification of the wing tip assemblies, both LH and RH sides.

#### Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

#### Modification:

Within the compliance time as defined in Table 1 of this AD, as applicable, modify the wing tip assemblies in accordance with the instructions of the applicable SB.

Table 1 – Wing Tip Modification

Aeroplane Configuration	Compliance Time
Post-SB A380-57-8145 (for LH wing)	Within 7 years after SB A380-57-8145 embodiment
Post-SB A380-57-8146 (for RH wing)	Within 7 years after SB A380-57-8146 embodiment
Pre-SB A380-57-8145 (for LH wing)	Before exceeding 7 years since aeroplane date of manufacture
Pre-SB A380-57-8146 (for RH wing)	

#### Ref. Publications:

Airbus SB A380-57-8132 original issue dated 03 February 2017, or Revision 01 dated 10 October 2019.

Airbus SB A380-57-8133 original issue dated 03 February 2017, or Revision 01 dated 23 January 2020.

The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.

Airbus SB A380-57-8145 original issue dated 22 July 2016, or Revision 01 dated 26 June 2017.

Airbus SB A380-57-8146 original issue dated 22 July 2016, or Revision 01 dated 26 June 2017.

#### Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.



2. This AD was posted on 26 April 2021 as PAD 21-057 for consultation until 24 May 2021. The Comment Response Document can be found in the [EASA Safety Publications Tool](#), in the compressed (zipped) file attached to the record for this AD.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: [ADs@easa.europa.eu](mailto:ADs@easa.europa.eu).
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
5. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS SAS - EIANA (Airworthiness Office), Telephone: +33 562 110 253, Fax: +33 562 110 307, E-mail: [account.airworth-A380@airbus.com](mailto:account.airworth-A380@airbus.com).

