



Airworthiness Directive

AD No.: 2021-0177

Issued: 23 July 2021

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

AIRBUS

Type/Model designation(s):

A319, A320 and A321 aeroplanes

Effective Date: 06 August 2021

TCDS Number(s): EASA.A.064

Foreign AD: Not applicable

Supersedure: None

ATA 29 – Hydraulic Power – Pylon / Engine Interface Rods – Inspection / Replacement

Manufacturer(s):

Airbus, formerly Airbus Industrie

Applicability:

Airbus A319-171N, A320-271N, A320-272N, A320-273N, A321-271N, A321-272N, A321-271NX and A321-272NX aeroplanes, all manufacturer serial numbers.

Definitions:

For the purpose of this AD, the following definitions apply:

The AOT: Airbus Alert Operators Transmission (AOT) A29N008-16.

The SB: Airbus Service Bulletin (SB) A320-29-1192.

Affected part: Pylon/engine interface rods having part number (P/N) D0003107000000, P/N D0003107000200, P/N D0003107000400 or P/N D0003107000600.

Serviceable part: Pylon/engine interface rods, eligible for installation, which are not affected parts; or affected parts which are new (not previously installed); or affected parts that, following installation, pass the inspections as required by this AD, as applicable.



Original rods: Affected parts, having P/N D0003107000000 or P/N D0003107000200.

Updated rods: Affected parts, having P/N D0003107000400 or P/N D0003107000600.

Groups: Group 1 are aeroplanes that have original rods installed.

Group 2 are aeroplanes that have updated rods installed.

Aeroplanes having Airbus modification 163348 embodied in production are Group 2.

Several pylon/engine interface rods are installed on an aeroplane; an aeroplane having original rods and updated rods installed is in both Group 1 and Group 2.

Airbus date of manufacture: The date of transfer of title (ownership) of the aeroplane at the time of first delivery to an operator, which is referenced in Airbus documentation.

Reason:

An occurrence was reported where, during inspection of the engines on an aeroplane, two original rods (as defined in this AD), installed to maintain an interface plate between pylon and nacelle, were found damaged at both rod-eye ends.

This condition, if not detected and corrected, could, in case of rod end rupture, lead to fuel pipe and hydraulic pipe chafing, possibly resulting in a fuel or hydraulic leak and consequent fire.

To address this potential unsafe condition, Airbus issued the AOT, providing instructions for repetitive inspections of the original rods, and designed updated rods (as defined in this AD), embodied in production with modification (mod) 163348, or in service through the accomplishment of Airbus SB A320-29-1189.

Following new findings on updated rods in service, Airbus published the SB, providing inspection instructions for the updated rods.

For the reasons described above, this AD requires repetitive detailed inspections (DET) of affected parts and, depending on findings, replacement.

This AD is considered an interim action and further AD action may follow.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

Inspection(s):

- (1) For Group 1 aeroplanes: Before exceeding 750 FH since aeroplane date of manufacture, or within 375 FH after the effective date of this AD, whichever occurs later, and, thereafter, at intervals not exceeding 750 FH, accomplish a DET of each original rod in accordance with the instructions of the AOT.
- (2) For Group 2 aeroplanes: Before exceeding 750 FH, but not before 650 FH since aeroplane date of manufacture, or after Airbus SB A320-29-1189 embodiment, as applicable, or within 750 FH after the effective date of this AD, whichever occurs later, and, thereafter, at intervals not



exceeding 1 500 FH, accomplish a DET of each updated rod, in accordance with the instructions of the SB.

Corrective Action(s):

- (3) If, during any inspection of an original rod as required by paragraph (1) of this AD, any defect is detected, as identified in the AOT, before next flight, replace that original rod with a serviceable part and accomplish the applicable corrective action(s) in accordance with the instructions of the AOT.
- (4) If, during any inspection of an updated rod as required by paragraph (2) of this AD, any defect is detected, as identified in the SB, before next flight, except as specified in paragraph (5) of this AD, replace that affected part with a serviceable part and accomplish the applicable corrective action(s) in accordance with the instructions of the SB, or contact Airbus for approved instructions and, within the compliance time(s) specified in those instructions, accomplish those instructions accordingly.
- (5) If, during any inspection of an updated rod as required by paragraph (2) of this AD, axial play/gap is detected not exceeding 1 mm, the part replacement, as required by paragraph (4) of this AD for that affected part, can be deferred until 100 FH after that inspection, provided that, within 24 hours before each flight (see Note 1 of this AD), the affected part passes a visual inspection in accordance with the instructions of the SB.

Note 1: Accomplishment of a daily visual inspection of the affected part in accordance with the instructions of the SB is acceptable to comply with the inspection requirement of paragraph (5) of this AD for that affected part.

Terminating Action:

- (6) None.

Reporting:

- (7) Within 90 days after each inspection as required by this AD, or after the effective date of this AD, whichever occurs later, report the results (including no findings for the initial inspection only) to Airbus. This can be accomplished in accordance with the instructions of the SB, or the AOT, as applicable.

Part(s) Installation:

- (8) For all aeroplanes: From the effective date of this AD, it is allowed to install an affected part on an aeroplane provided, following installation, it is inspected within the compliance time as identified in Table 1 of this AD, as applicable, and, thereafter, as required by paragraph (1) or (2) of this AD, as applicable.



Table 1 – Compliance time

	Compliance time
Original Rods	Before accumulating 750 FH since first installation on an aeroplane, or since last DET, or before next flight after installation, whichever occurs later
Updated Rods	<p>Whichever occurs later: A, B or C</p> <p>A) Within 750 FH but not before 650 FH since first installation on an aeroplane</p> <p>B) Within 1 500 FH since last DET accomplished after having accumulated 650 FH since first installation on an aeroplane</p> <p>C) Before next flight after installation</p>

Ref. Publications:

Airbus AOT A29N008-16 at original issue dated 03 January 2017, or Revision 01 dated 05 January 2017, or Revision 02 dated 23 January 2019.

Airbus SB A320-29-1192 original issue dated 31 May 2021.

The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. Based on the required actions and the compliance time, EASA have decided to issue a Final AD with Request for Comments, postponing the public consultation process until after publication.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
5. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – Airworthiness Office – IIASA; E-mail: account.airworth-eas@airbus.com.

