

Airworthiness Directive

AD No.: 2022-0019

Issued: 31 January 2022

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

AIRBUS

Type/Model designation(s):

A380 aeroplanes

Effective Date: 14 February 2022

TCDS Number(s): EASA.A.110

Foreign AD: Not applicable

Supersedure: This AD supersedes EASA AD 2019-0223 dated 06 September 2019.

ATA 57 – Wings – Front and Rear Spars – Inspection

Manufacturer(s):

Airbus

Applicability:

Airbus A380-841, A380-842 and A380-861 aeroplanes, all manufacturer serial numbers (MSN).

Definitions:

For the purpose of this AD, the following definitions apply:

The Airbus instructions: Airbus Service Bulletin (SB) A380-57-8263 Revision 01 and Airbus Alert Operator Transmission (AOT) A57R019-21, as applicable.

Affected areas (on both left-hand and right-hand sides):

- Wing outer rear spar (ORS) top and bottom flanges between ribs 33 and 49 (for aeroplanes in pre-modification (pre-mod) 77989 configuration, all MSN up to 270 inclusive)
- Wing outer inner front spar (OIFS) top and bottom flanges between ribs 8 and 14;
- Wing outer front spar (OFS) top and bottom flanges between ribs 38 and 49 (for aeroplanes in pre-mod 77990 configuration, all MSN up to 270 inclusive).

Applicable wing box assembly date: The date of wing box assembly, as applicable to MSN, listed in relevant Appendix of the Airbus instructions.

Reason:

Occurrences have been reported of finding cracks in the affected areas of the wing ORS on in-service A380 aeroplanes.

This condition, if not detected and corrected, could reduce the structural integrity of the wing.

To address this potential unsafe condition, Airbus issued SB A380-57-8263 original issue to provide inspection instructions, and EASA issued AD 2019-0223, as an interim action, to require repetitive special detailed inspections (SDI), using ultrasonic testing methods, of the affected area of the wing ORS on a limited batch of aeroplanes.

Since that AD was issued, it has been determined that additional areas may be affected by the same unsafe condition, and that all MSN must be inspected; consequently, Airbus issued the Airbus instructions, as defined in this AD, providing applicable instructions.

For the reason described above, this AD retains the requirements of EASA AD 2019-0223, which is superseded, expanding the Applicability to all aeroplanes, including those in pre-mod 77989 configuration (for ORS), and expanding the affected areas subject to inspection to include the wing OFS (for aeroplanes in pre-mod 77990 configuration) and the wing OIFS.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

Inspection(s):

- (1) Before exceeding 180 months since the applicable wing box assembly date, and, thereafter, at intervals not to exceed 36 months (see Note 1 of this AD), accomplish an SDI of the affected areas in accordance with the Airbus instructions.

Note 1: The 36-month inspection interval is applicable for unrepaired areas. For areas repaired as required by paragraph (2) of this AD, the intervals for post-repair inspection as required by paragraph (1) of this AD are those specified in the Airbus approved repair instructions.

Corrective Action(s):

- (2) If, during any inspection as required by paragraph (1) of this AD, any crack is detected, before next flight, contact Airbus for approved repair instructions and, within the compliance time specified therein, accomplish those instructions accordingly (see Note 1 of this AD). If no compliance time for the repair is identified in those instructions, accomplish those instructions before next flight.

Credit:

- (3) Inspections of the affected areas of the wing ORS, accomplished on an aeroplane before the effective date of this AD in accordance with the instruction of Airbus SB A380-57-8263 original issue, are acceptable to comply with the initial requirements of paragraph (1) of this AD, as applicable, for that aeroplane.

Terminating Action:

- (4) None.



Reporting:

- (5) Within 30 days after each inspection as required by paragraph (1) of this AD, report the inspection results (including no findings) to Airbus.

Ref. Publications:

Airbus SB A380-57-8263 original issue dated 23 August 2019, and Revision 01 dated 31 January 2022.

Airbus AOT A57R019-21 original issue dated 31 January 2022.

The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. Based on the required actions and the compliance time, EASA have decided to issue a Final AD with Request for Comments, postponing the public consultation process until after publication.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
5. For any question concerning the technical content of the requirements in this AD, please contact: Airbus – EIANA (Airworthiness Office), Telephone: +33 562 110 253, Fax: +33 562 110 307, E-mail: account.airworth-A380@airbus.com.

