

## Airworthiness Directive

**AD No.:** 2022-0252

**Issued:** 16 December 2022

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I Part M.A.301, or Annex Vb Part ML.A.301, as applicable, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I Part M.A.303, or Annex Vb Part ML.A.303, as applicable] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

### Design Approval Holder's Name:

ROLLS-ROYCE DEUTSCHLAND Ltd & Co KG

### Type/Model designation(s):

BR700-715 engines

**Effective Date:** 30 December 2022

**TCDS Number(s):** EASA.E.023

**Foreign AD:** Not applicable

**Supersedure:** None

## ATA 72 – Engine – Low Pressure Compressor Booster Rotor – Inspection

### Manufacturer(s):

Rolls-Royce Deutschland Ltd & Co KG (RRD)

### Applicability:

BR700-715A1-30, BR700-715B1-30 and BR700-715C1-30 engines, all serial numbers (s/n).

These engines are known to be installed on, but not limited to, Boeing 717 aeroplanes.

### Definitions:

For the purpose of this AD, the following definitions apply:

**The NMSB:** RRD Alert Non-Modification Service Bulletin (NMSB) SB-BR700-72-A900738.

**Affected part:** Low pressure (LP) compressor booster rotors, having Part Number (P/N) BRH19871 or P/N BRH20241, and an s/n in the range of 256 to 493 (inclusive).

**Flight mission:** Hawaiian Flight missions (FM 1); all other Flight missions (FM 2); and combined Hawaiian and other Flight missions (FM 3).

**Reason:**

Occurrences have been reported of finding malformed scallop edge geometry and surface conditions at the front flange scallops of certain LP compressor booster rotors.

This condition, if not detected and corrected, could lead to failure of an affected part, possibly resulting in release of high-energy debris with consequent engine in-flight shut-down and reduced control of the aeroplane.

To address this potential unsafe condition, RRD issued the NMSB to provide inspection instructions and accept/reject criteria.

For the reason described above, this AD requires repetitive fluorescent-penetrant inspections (FPI) of the front flange scallops of the affected part and, depending on findings, removal from service of the engine for corrective action(s).

**Required Action(s) and Compliance Time(s):**

Required as indicated, unless accomplished previously:

**Inspections:**

- Within the compliance time as specified in Table 1 of this AD, as applicable, or within 500 engine flight cycles (EFC) after the effective date of this AD, whichever occurs later, and, thereafter, at intervals not to exceed the value as specified in Table 2 of this AD, as applicable, accomplish an FPI (on-wing or in-shop) of the front flange scallops of the affected part in accordance with the instructions of the NMSB.

Table 1 – Affected Parts Initial FPI

Flight Mission	Compliance Time
FM 1	Before the affected part exceeds 36 000 EFC since new (first installation on an engine)
FM 2 and FM 3	Before the affected part exceeds 18 000 EFC since new (first installation on an engine)

Table 2 – Affected Parts FPI Interval

Flight Mission	Interval (not to exceed)
FM 1	6 000 EFC
FM 2 and FM 3	5 000 EFC

**Corrective Action(s):**

- If, during any on-wing FPI as required by paragraph (1) of this AD, any cracks are detected, before next flight, remove the engine from service and, before return to service of the engine, contact RRD for approved corrective action(s) and accomplish those actions accordingly.



- (3) If, during any in-shop FPI as required by paragraph (1) of this AD, any cracks are detected, before return to service of the engine, contact RRD for approved corrective action(s) and accomplish those actions accordingly.

#### Parts Installation:

- (4) From the effective date of this AD, it is allowed to install an affected part on any engine, provided that the part has not exceeded 36 000 EFC (for FM 1) or 18 000 EFC (for FM 2 and FM 3), as applicable, or has passed an FPI (no crack detected) in accordance with the instructions of the NMSB and that, following installation, the affected part is inspected as required by this AD.

#### Terminating Action:

- (5) Inspection of an engine in accordance with the instructions of Part C of the NMSB, including validation of the results from RRD, confirming that the affected part has no malformed scallop edge geometry or surface conditions at the front flange scallops, constitutes terminating action for the repetitive FPI as required by paragraph (1) of this AD (Table 2 intervals) for that engine.

#### Ref. Publications:

RRD Alert NMSB SB-BR700-72-A900738 original issue dated 15 December 2022.

The use of later approved revisions of the above-mentioned document are acceptable for compliance with the requirements of this AD.

#### Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. Based on the required actions and the compliance time, EASA have decided to issue a Final AD with Request for Comments, postponing the public consultation process until after publication.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: [ADs@easa.europa.eu](mailto:ADs@easa.europa.eu).
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
5. For any question concerning the technical content of the requirements in this AD, please contact: Rolls-Royce Deutschland Ltd & CoKG, Eschenweg 11, Dahlewitz, 15827 Blankenfelde-Mahlow, Germany, Telephone: +49 (0) 337086 1200, E-mail: [rrd.techhelp@rolls-royce.com](mailto:rrd.techhelp@rolls-royce.com).

