



Airworthiness Directive

AD No.: 2024-0105

Issued: 30 May 2024

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I Part M.A.301, or Annex Vb Part ML.A.301, as applicable, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I Part M.A.303, or Annex Vb Part ML.A.303, as applicable] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

AIRBUS S.A.S.

Type/Model designation(s):

A319 aeroplanes

Effective Date: 13 June 2024

TCDS Number(s): EASA.A.064

Foreign AD: Not applicable

Supersedure: None

ATA 53 – Fuselage – Potable and Waste Water Service Panels – Modification

Manufacturer(s):

Airbus, formerly Airbus Industrie

Applicability:

Airbus A319-111, A319-112, A319-113, A319-114, A319-115, A319-131, A319-132, A319-133 aeroplanes, all manufacturer serial numbers except those on which:

- Airbus modification (mod) 160001 was embodied in production or SB A320-57-1193 was embodied in service;

and/or

- Airbus mod 28238, mod 28162 and mod 28342 were embodied in production.

Definitions:

For the purpose of this AD, the following definitions apply:

Affected areas: Potable water panel at stringer (STR) 43 between frame (FR) 65 and FR66 and waste water panel at STR40 between FR65 and FR66.

The SB: Airbus Service Bulletin (SB) A320-53-1517.



Reason:

During the Widespread Fatigue Damage (WFD) scenario review in the frame of the Airbus Extra Long Range (XLR) certification, it has been identified that some previous findings of the Full Scale Fatigue Test in WFD areas were not properly correlated. The WFD criteria in terms of number of findings per location and end-of-test demonstrated life have to be assessed accordingly. Several cracks found around the potable and waste water service panels (WFD areas) shall be considered as WFD instead of typical fatigue findings and, therefore, WFD fatigue calculations must be updated accordingly.

This condition, if not corrected could affect the structural integrity of the fuselage.

To address this potential unsafe condition, Airbus issued the SB to provide modification instructions.

For the reason described above, this AD requires accomplishment of the modification.

Required Action(s) and Compliance Time(s):

Required as indicated by this AD, unless the action(s) required by this AD have been already accomplished:

Modification:

- (1) Before exceeding 56 800 flight cycles (FC) since aeroplane first flight, accomplish the modification of each affected area, as applicable, in accordance with the instructions of the SB.

Note 1: The SB requires, as one of the steps of the modification, an inspection and potential corrective action, depending on the findings.

ALI Tasks Superseded:

- (2) Accomplishment of the modification on an aeroplane as required by paragraph (1) of this AD, supersedes ALI tasks 534141-02-2 or 534142-02-1, as applicable, for that aeroplane, unless specified otherwise in the instructions provided by Airbus in case of findings (see Note 1 of the AD).

Ref. Publications:

Airbus SB A320-53-1517 original issue dated 12 December 2023.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. This AD was posted on 25 April 2024 as PAD 24-046 for consultation until 23 May 2024. The Comment Response Document can be found in the [EASA Safety Publications Tool](#), in the compressed ('zipped') file, attached to the record for this AD.



3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
5. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – Airworthiness Office – 1IASA; E-mail: account.airworth-eas@airbus.com.

