



Airworthiness Directive

AD No.: 2026-0072

Issued: 01 April 2026

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I Part M.A.301, or Annex Vb Part ML.A.301, as applicable, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I Part M.A.303, or Annex Vb Part ML.A.303, as applicable] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

LEONARDO S.p.A.

Type/Model designation(s):

AW189 helicopters

Effective Date: 15 April 2026

TCDS Number(s): EASA.R.510

Foreign AD: Not applicable

Supersedure: This AD supersedes EASA AD 2025-0213 dated 26 September 2025.

ATA 71 – Power Plant – Engine Mount Fitting – Inspections / Replacement

Manufacturer(s):

Leonardo S.p.A. Helicopters, formerly Finmeccanica S.p.A., AgustaWestland S.p.A.

Applicability:

AW189 helicopters, all serial numbers (s/n).

Definitions:

For the purpose of this AD, the following definitions apply:

The SB: Leonardo Alert Service Bulletin (SB) 189-402 Revision B.

Affected part: Engine external fitting right-hand (RH) Part Number (P/N) 8G5333A04351; and engine external fitting left-hand (LH) P/N 8G5333A04551.

Serviceable part: An affected part that is new (never previously installed on a helicopter).

Groups: Group 1 helicopters are s/n from 92001 to 92010 inclusive.
Group 2 helicopters are those that are not Group 1.



Reference date: The starting date, which is used to determine the due date for the first '8 years scheduled inspection' as specified in Leonardo Air Vehicle Maintenance Planning Information 89-A-AMPI-00-P or 89-E-AMPI-00-P, as applicable.

Reason:

Occurrences of heavy corrosion damage have been reported on LH and RH engine mount external fittings installed on AW189 helicopters.

This condition, if not detected and corrected, could reduce the structural integrity of the engine mount.

To address this potential unsafe condition, Leonardo issued the SB 189-402 original issue providing instructions for repetitive inspections of the engine mount external fittings of certain helicopters, and, depending on findings, replacement. Consequently, EASA published AD 2025-0213 requiring accomplishment of repetitive inspections of each affected part for those helicopters, and, depending on findings, its replacement.

Since that AD was issued, it has been determined that the potential unsafe condition could affect all AW189 helicopters.

For the reasons described above, this AD retains the requirements of EASA AD 2025-0213, which is superseded, and expands the Applicability to all s/n.

Required Action(s) and Compliance Time(s):

Required as indicated by this AD, unless the action(s) required by this AD have been already accomplished:

Inspections:

- (1) For Group 1 helicopters: Within 100 flight hours (FH) after 10 October 2025 [the effective date of EASA AD 2025-0213] and, thereafter, at intervals not to exceed 400 FH, inspect each affected part in accordance with the instructions of Part I of the Accomplishment Instructions of the SB.
- (2) For Group 1 helicopters: Within 24 months, or concurrently with the first engine 1 removal, whichever occurs first after 10 October 2025 [the effective date of EASA AD 2025-0213], inspect the LH affected part in accordance with the instructions of Part II of the Accomplishment Instructions of the SB.
- (3) For Group 1 helicopters: Within 24 months, or concurrently with the first engine 2 removal, whichever occurs first after 10 October 2025 [the effective date of EASA AD 2025-0213], inspect the RH affected part in accordance with the instructions of Part II of the Accomplishment Instructions of the SB.
- (4) For Group 2 helicopters: Before exceeding 8 years since the reference date or within 100 FH, whichever occurs later after the effective date of this AD, and, thereafter, at intervals not to exceed 400 FH, inspect each affected part in accordance with the instructions of Part I of the Accomplishment Instructions of the SB.



- (5) For Group 2 helicopters: Within the compliance time as defined in Table 1 of this AD, but not before having accumulated 7 years since the reference date, inspect the LH affected part in accordance with the instructions of Part II of the Accomplishment Instructions of the SB.
- (6) For Group 2 helicopters: Within the compliance time as defined in Table 1 of this AD, but not before having accumulated 7 years since the reference date, inspect the RH affected part in accordance with the instructions of Part II of the Accomplishment Instructions of the SB.

Table 1 – Inspection Compliance Time

A or B, whichever occurs later	
A	Before exceeding 8 years since the reference date
B	Within 24 months after the effective date of this AD or at first engine 1 or engine 2, as applicable, removal, whichever occurs first after the effective date of this AD

Corrective Action(s):

- (7) If, during any inspection as required by paragraph (1) to (6) of this AD, as applicable, any crack or corrosion is detected on an affected part, before next flight, contact Leonardo Product Engineering Support for applicable instructions and, within the compliance time specified therein, accomplish those instructions accordingly.

Terminating Action:

- (8) Replacement of the affected part (LH or RH, respectively) with a serviceable part, accomplished on a helicopter in accordance with the instructions of the SB, constitutes terminating action for the repetitive inspections (including the initial inspection) as required by paragraph (1) or (4) of this AD, as applicable, for the engine external fitting (LH or RH, respectively) of that helicopter.
- (9) If, during the inspection as required by paragraph (2), (3), (5) or (6) of this AD, as applicable, no crack or corrosion is detected on an affected part, that inspection constitutes terminating action for the repetitive inspections (including the initial inspection) as required by paragraph (1) or (4) of this AD for that affected part.

Credit:

- (10) For Group 1 helicopters: Inspections, corrective action(s) or replacement of an affected part, accomplished on a helicopter before the effective date of this AD in accordance with the instructions of Leonardo SB 189-402 original issue or Revision A, are acceptable to comply with the requirements of paragraphs (1), (2), (3), (7) or (8) of this AD, as applicable, for that helicopter.
- (11) For Group 2 helicopters: Inspections, corrective action(s) or replacement of an affected part, accomplished on a helicopter before the effective date of this AD in accordance with the instructions of Leonardo SB 189-402 Revision A, are acceptable to comply with the requirements of paragraphs (4), (5), (6), (7) or (8) of this AD, as applicable, for that helicopter.



Ref. Publications:

Leonardo SB 189-402 original issue dated 11 August 2025, Revision A dated 22 December 2025, and Revision B dated 16 March 2026.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. Based on the required actions and the compliance time, EASA have decided to issue a Final AD with Request for Comments, postponing the public consultation process until after publication. All interested persons may send their comments, referencing the AD Number, to the E-mail address specified in below Remark 3, prior to 29 April 2026. Only if any comment is received during the consultation period, a Comment Response Document will be published in the [EASA Safety Publications Tool](#), in a compressed ('zipped') file, attached to the record for this AD.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
5. For any question concerning the technical content of the requirements in this AD, please contact: Leonardo S.p.A. Helicopters, E-mail: engineering.support.lhd@leonardocompany.com.

