



Airworthiness Directive

AD No.: 2026-0195

Issued: 13 October 2026

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I Part M.A.301, or Annex Vb Part ML.A.301, as applicable, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I Part M.A.303, or Annex Vb Part ML.A.303, as applicable] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

AIRBUS HELICOPTERS

Type/Model designation(s):

SA 365 N helicopters

Effective Date: 27 October 2026

TCDS Number(s): EASA.R.105

Foreign AD: Not applicable

Supersedure: None

ATA 65 – Tail Rotor Drive – Tail Gear Box Control Rod Bearings – Replacement / Life Limitation

Manufacturer(s):

Airbus Helicopters (AH), formerly Eurocopter (EC), Eurocopter France, Aerospatiale

Applicability:

SA 365 N helicopters, all serial numbers.

Definitions:

For the purpose of this AD, the following definitions apply:

The ASB: Airbus Helicopters (AH) Alert Service Bulletin (ASB) AS365-05-00-0005.

Affected part: Tail gear box (TGB) control rod bearing, having Part Number 360A33-4052-00.

Serviceable part: A TGB control rod bearing, eligible for installation in accordance with AH instructions, that is not an affected part; or an affected part that has accumulated less than 500 flight hours (FH) since new (first installation on a helicopter).



Reason:

An occurrence was reported, where metal particles were found on the control rod bearings. Subsequent investigation identified premature bearing spalling as the root cause of the metallic particles.

This condition, if not corrected, could lead to loss of yaw control of the helicopter.

To address this potential unsafe condition, AH issued the ASB to provide instructions for replacement of the affected parts at reduced life limits.

For the reason described above, this AD requires replacement of affected parts before exceeding the reduced life limit.

Required Action(s) and Compliance Time(s):

Required as indicated by this AD, unless the action(s) required by this AD have been already accomplished:

Replacement / Life Limit:

- (1) Within 500 FH after the effective date of this AD or before the affected part accumulates 500 FH since first installation on a helicopter, whichever occurs later, replace that affected part with a serviceable part, as defined in this AD, in accordance with the instructions of the ASB.

Part(s) Installation:

- (2) From the effective date of this AD, it is allowed to install an affected part on any helicopter, provided that the part is a serviceable part, and, following installation, it is replaced with a serviceable part before it accumulates 500 FH since first installation on a helicopter.

Ref. Publications:

AH ASB AS365-05-00-0005 issue 001 dated 22 June 2026.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. The original issue of this AD was posted on 11 September 2026 as PAD 26-105 for consultation until 09 October 2026. No comments were received during the consultation period.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than



those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.

5. For any question concerning the technical content of the requirements in this AD, please contact: Airbus Helicopters (Technical Support) – Aéroport de Marseille Provence, 13725 Marignane Cedex, France, Telephone: +33 (4) 42 85 97 97, Fax: +33 (4) 42 85 99 66, Web portal: <https://airbusworld.helicopters.airbus.com> Technical Requests Management, or E-mail: TechnicalSupport.Helicopters@airbus.com

