



Notification of a Proposal to issue an Airworthiness Directive

PAD No.: 24-119

Issued: 16 October 2024

Note: This Proposed Airworthiness Directive (PAD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

In accordance with the EASA Continuing Airworthiness Procedures, the Executive Director is proposing the issuance/cancellation of an EASA Airworthiness Directive (AD), applicable to the aeronautical product(s) identified below.

All interested persons may send their comments, referencing the PAD Number above, to the e-mail address specified in the 'Remarks' section, prior to the consultation date indicated.

Design Approval Holder's Name:

CFM INTERNATIONAL S.A.

Type/Model designation(s):

LEAP-1B engines

Effective Date: [TBD – standard: 14 days after AD issue date]

TCDS Number(s): EASA.E.115

Foreign AD: Not applicable

Supersedure: This AD supersedes EASA AD 2023-0109 dated 26 May 2023.

ATA 72 – Engine – High-Pressure Compressor Stages 6-10 Spools and High-Pressure Turbine Rotor Stage 1 Disks – Replacement

Manufacturer(s):

SAFRAN Aircraft Engines, formerly SNECMA (France); General Electric (United States)

Applicability:

LEAP-1B21, LEAP-1B23, LEAP-1B25, LEAP-1B27, LEAP-1B28, LEAP-1B28B1, LEAP-1B28B2, LEAP-1B28B2C, LEAP-1B28B3, LEAP-1B28BBJ1 and LEAP-1B28BBJ2 engines, all serial numbers (s/n).

These engines are known to be installed on, but not limited to, Boeing 737-8, 737-8200 and 737-9 aeroplanes.

Definitions:

For the purpose of this AD, the following definitions apply:

The SB: CFM International (CFM) Service Bulletin (SB) LEAP-1B-72-00-0392-01A-930A-D, and SB LEAP-1B-72-00-0402-01A-930A-D.



The SB tables: Tables 1, 2 and 3 of CFM SB LEAP-1B-72-00-0392-01A-930A-D and Table 1 of CFM SB LEAP-1B-72-00-0402-01A-930A-D.

Affected part: High-pressure compressor (HPC) Stage 6-10 spools and high-pressure turbine (HPT) Stage 1 Disks, having a Part Number (P/N) and s/n as listed in the SB Tables.

Serviceable part: Any HPC stage 6-10 spool or HPT stage 1 disk, eligible for installation, that is not an affected part.

Groups: Group 1 engines are those that have an affected part installed.
Group 2 engines are those that do not have an affected part installed.

Reason:

Three HPT rotor disks from a different engine type were found to contain iron inclusions. This has been attributed to specific deficiencies in the manufacturing process. Iron inclusion may lead to reduced mechanical properties and failure of an affected part prior to achieving its approved life as published in the Airworthiness Limitations Section (ALS) of the Engine Manual. It has been determined that the affected parts identified in the SB were manufactured using the same processes, and may also have reduced mechanical properties due to iron inclusion.

This condition, if not corrected, could lead to failure of affected parts, possibly resulting in high-energy debris release, with consequent damage to, and reduced control of, the aeroplane.

To address this potential unsafe condition, CFM published LEAP-1B-72-00-0392-01A-930A-D to provide replacement instructions for certain affected parts, and EASA issued AD 2023-0109, to require replacement of those affected parts and to prohibit (re)installation.

Since that AD was issued, additional parts have been determined to be affected, and CFM issued SB LEAP-1B-72-00-0402-01A-930A-D, to provide a list of the newly identified affected parts and instructions for replacement.

For the reasons described above, this AD retains the requirements of EASA AD 2023-0109, which is superseded, additionally requires replacement of affected parts listed in CFM SB LEAP-1B-72-00-0402-01A-930A-D, and prohibits reinstallation.

Required Action(s) and Compliance Time(s):

Required as indicated by this AD, unless the action(s) required by this AD have been already accomplished:

Replacement:

- (1) For Group 1 engines: Within the compliance time as specified in Table A of this AD, replace each affected part with a serviceable part in accordance with the instructions of the SB.

Note 1: The cycles since new (CSN) specified in the SB tables are those accumulated by the affected part since its first installation on an engine.



Note 2: The engine shop manual provides an acceptable method to determine the remaining cycles available for an affected part which has been operated on different engine models / thrust ratings.

Table A – Affected Part Replacement

Compliance Time (A or B, whichever occurs first)	
A	Within the compliance time as specified in the SB tables, as applicable (see Notes 1 and 2 of this AD), or within 50 engine cycles after the effective date of this AD, whichever occurs later
B	During the next piece-part exposure after the effective date of this AD

Parts Installation:

(2) From the effective date of this AD, do not install any affected part on any engine.

Ref. Publications:

CFM SB LEAP-1B-72-00-0392-01A-930A-D Issue 002 dated 05 September 2023.

CFM SB LEAP-1B-72-00-0402-01A-930A-D original issue (Issue 001) dated 23 January 2024.

The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.

Remarks:

1. This Proposed AD will be closed for consultation on 13 November 2024.
2. Enquiries regarding this PAD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
3. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this PAD, and which may occur, or have occurred on a product, part or appliance not affected by this PAD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this PAD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
4. For any question concerning the technical content of the requirements in this PAD, please contact: CFM International S.A., Customer Support Centre, Telephone: +33 1 64 14 88 66, Fax: +33 1 64 14 87 65, E-mail: cfm.csc@safrangroup.com,

or

CFM Inc., GE Aerospace Fleet Support, Telephone: +1 513-552-3272 or +1 877-432-3272, E-mail: aviation.fleetsupport@ge.com.

