



Airworthiness Directive

AD No.: 2015-0033R1

Issued: 03 May 2016

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) 216/2008 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EC) 216/2008, Article 14(4) exemption].

Design Approval Holder's Name:

AIRBUS HELICOPTERS

Type/Model designation(s):

EC 130 helicopters

Effective Date: Revision 1: 17 May 2016
Original issue: 26 February 2015

TCDS Number(s): EASA.R.008

Foreign AD: Not applicable

Revision: This AD revises EASA Emergency AD 2015-0033-E dated 24 February 2015, which superseded EASA AD 2014-0145R1 dated 13 June 2014.

ATA 53 – Fuselage – Tail Boom / Fenestron Junction Frame – Inspection

Manufacturer(s):

Airbus Helicopters (formerly Eurocopter, Eurocopter France)

Applicability:

EC 130 B4 and EC 130 T2 helicopters, all serial numbers.

Reason:

Two events of crack propagation through the junction frame of the tail boom / Fenestron were reported following non-scheduled inspections of EC 130 B4 helicopters. The investigation revealed that the cracks initiated in the lower right hand (RH) part of the frame between the web and the flange where the lower spar of the tail boom is joined. Although the cracks were of significant length, no deterioration was visible from the outside of the helicopter.

This condition, if not detected and corrected, could lead to structural failure, possibly resulting in Fenestron detachment and consequent loss of control of the helicopter.

To address this unsafe condition, Airbus Helicopters (AH) issued EC130 Emergency Alert Service Bulletin (ASB) No. 05A017 to provide instructions for detailed visual checks on the inside of the tail boom.



Prompted by these findings, EASA issued Emergency AD 2014-0145-E (later revised) to require repetitive inspections of the affected area and, depending on findings, accomplishment of applicable corrective actions.

Since EASA AD 2014-0145R1 was published, it was determined that, apart from the existing flight hours (FH) inspection interval, an inspection interval defined in sling cycles is necessary. In addition, AH issued Service Bulletin (SB) EC130-53-029 which provides a modification that will allow the inspections to be accomplished from outside the tailboom. AH have revised EC130 ASB No. 05A017 (Revision 2) accordingly, to define this alternative method.

Consequently, EASA issued Emergency AD 2015-00033-E, retaining the requirements of EASA AD 2014-0145R1, which was superseded, to allow, for post-MOD 350A087421 and post-SB EC130-53-029 helicopters, an alternative visual (external) inspection method, although with reduced inspection intervals, in combination with the internal inspections at extended intervals. That AD also introduced an additional sling cycle interval, as defined in AH EC130 ASB 05A017 Revision 2. That AD also specified that certain tasks can be accomplished by the pilot.

Since EASA AD 2015-0033-E was issued, it was determined that, for helicopters with a tail boom / Fenestron junction frame installed which has accumulated less than 1 200 FH, inspection is not necessary. For this reason, AH have revised EC130 ASB No. 05A017 (Revision 3) accordingly.

This AD is revised to confirm the change to the inspection threshold.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

- (1) Within the compliance time as specified in Table 1 of this AD, as applicable, accomplish a detailed visual inspection of the frame web in the radius between the web and the flange on the tail cone side in accordance with the instructions of paragraph 3.B.1 of AH EC130 Emergency ASB 05A017.

Table 1 – Inspection Threshold

FH accumulated (see Note 1)	Compliance Time
Less than 1 190 FH	Before exceeding 1 200 FH
1 190 FH or more	Within 10 FH after 09 June 2014 [the effective date of EASA AD 2014-0145-E]

Note 1: The FH referenced in Table 1 of this AD are those accumulated by the junction frame since first installation on a helicopter.

- (2) Within 25 FH or 390 sling cycles (see Note 2 of this AD), whichever occurs first after the inspection as required by paragraph (1) of this AD, and, thereafter, at intervals not to exceed 25 FH or 390 sling cycles, whichever occurs first, accomplish an inspection in accordance with the instructions of paragraph 3.B.1 or 3.B.2 of AH EC130 Emergency ASB 05A017.



Note 2: The definition of sling cycles can be found in paragraph 1.E.2 (b) of AH EC130 ASB No. 05A017 Revision 3.

- (3) For helicopters modified in production by AH MOD. 350A087421, or in service in accordance with the instructions of AH SB EC130-53-029, as an alternative to the inspections as required by paragraphs (1) and (2) of this AD, it is acceptable to accomplish the inspections as specified in paragraphs (3.1) and (3.2) of this AD.
 - (3.1) Within 10 FH or 250 sling cycles, whichever occurs first after modification of the helicopter, and, thereafter, at intervals not exceeding 10 FH or 250 sling cycles, whichever occurs first, accomplish a visual inspection (see Note 2) of the tailboom from outside in accordance with the instructions of paragraph 3.B.3 of AH Emergency ASB EC130 05A017.
 - (3.2) Within 660 FH after modification of the helicopter, and, thereafter, at intervals not to exceed 660 FH, accomplish a detailed visual inspection of the frame web in the radius between the web and the flange on the tail cone side in accordance with the instructions of paragraph 3.B.1 or 3.B.2 of AH EC130 Emergency ASB 05A017.

Note 3: The visual inspection as specified in paragraph 3.B.3 (a) of AH Emergency ASB EC130 No. 05A017 Revision 3 may be carried out by the pilot.

- (4) If, during any inspection as required by paragraph (1), (2), or (3) of this AD, a crack is detected, before next flight, contact AH for approved repair instructions and accomplish those instructions accordingly.
- (5) Repair of a helicopter, as required by paragraph (4) of this AD, does not constitute terminating action for the repetitive inspections as required by this AD for that helicopter.
- (6) Installation of a replacement (new) junction frame on a helicopter is allowed, provided that, before exceeding 1 200 FH after installation a detailed visual inspection of the frame web in the radius between the web and the flange on the tail cone side is accomplished in accordance with the instructions of paragraph 3.B.1 of Airbus Helicopters EC130 Emergency ASB No. 05A017, and, following installation, the helicopter is inspected as required by paragraph (2) or paragraph (3) of this AD.

Ref. Publications:

Airbus Helicopters EC130 Emergency ASB No. 05A017 original issue, dated 06 June 2014, or Revision 1 dated 13 June 2014, or Revision 2 dated 20 February 2015, or Revision 3 dated 07 March 2016.

The use of later approved revisions of this documents is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.



2. Based on the required actions and the compliance time, EASA have decided to issue a Final AD with Request for Comments, postponing the public consultation process until after publication.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. For any question concerning the technical content of the requirements in this AD, please contact: Airbus Helicopters (EBSESB) – Aéroport de Marseille Provence, 13725 Marignane Cedex, France; telephone +33 (4) 12 85 97 97; facsimile +33 (4) 85 99 66; E-mail: Directive.technical-support@airbus.com.

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