


EASA	NOTIFICATION OF A PROPOSAL TO ISSUE AN AIRWORTHINESS DIRECTIVE	
	PAD No.: 15-073 Date: 02 June 2015 Note: This Proposed Airworthiness Directive (PAD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.	
	In accordance with the EASA Continuing Airworthiness Procedures, the Executive Director is proposing the issuance of an EASA Airworthiness Directive (AD), applicable to the aeronautical product(s) identified below. All interested persons may send their comments, referencing the PAD Number above, to the e-mail address specified in the 'Remarks' section, prior to the consultation closing date indicated.	
Design Approval Holder's Name: AIRBUS		Type/Model designation(s): A330 aeroplanes
TCDS Number: EASA.A.004		
Foreign AD: Not applicable		
Supersedure: This AD supersedes EASA AD 2013-0094 dated 16 April 2013.		
ATA 71	Power Plant – Upper Aft Mount Bolts – Inspection / Replacement	
Manufacturer(s):	Airbus (formerly Airbus Industrie)	
Applicability:	Airbus A330-201, A330-202, A330-203, A330-301, A330-302 and A330-303 aeroplanes, all manufacturer serial numbers, except those on which Airbus modification 203947 has been embodied in production.	
Reason:	<p>During a scheduled replacement of a CF6-80E1 engine on an A330 aeroplane, a bolt on the aft engine mount upper beam was found sheared. The affected bolt is one out of four bolts that attach the upper beam to the pylon.</p> <p>Investigation results revealed an unusual contact with the counter-bore edge of the beam which induced a significant groove on the bolt during its installation in production. It is suspected that the induced groove led to a fatigue crack initiation and subsequent quick propagation leading to the complete fracture of the bolt. In case of multiple bolt fractures, the remaining bolts would be insufficient to sustain the residual fatigue and limit loads.</p> <p>This condition, if not detected and corrected, could lead, in case of multiple bolt fracture, to loss of an engine mount structural integrity and possible in-flight engine detachment, resulting in reduced control of the aeroplane and/or injury to persons on the ground.</p> <p>To address this potential unsafe condition, EASA issued AD 2013-0094 to require a one-time ultrasonic (US) inspection of the four aft mount-pylon bolts of both engines to detect sheared bolts and, depending on findings, accomplishment of applicable corrective actions.</p>	

	<p>Since EASA AD 2013-0094 was issued, further investigation results revealed that the pylon bolt failure was caused by inappropriate upper beam installation during production. An abnormal bending load applied on the bolt during installation of the upper beam could have increased the stress close to or beyond the limit strength, high enough to fracture the bolt.</p> <p>Prompted by these findings, Airbus issued Service Bulletin (SB) A330-71-3031 providing instructions for repetitive inspections and the applicable corrective actions.</p> <p>For the reasons described above, this AD, which supersedes EASA AD 2013-0094, requires repetitive US inspections of the aft mount-pylons bolts of each engine and, depending on findings, corrective actions.</p>
Effective Date:	[TBD: 14 days after final AD issue date]
Required Action(s) and Compliance Time(s):	<p>Required as indicated, unless accomplished previously:</p> <ol style="list-style-type: none"> (1) Within 26 months after the effective date of this AD and, thereafter, at intervals not to exceed 24 months, accomplish an US inspection of each of the four aft mount-pylon bolts of each engine in accordance with the instructions of Airbus SB A330-71-3031. (2) If, during any inspection as required by paragraph (1) of this AD, a fractured bolt is detected, before next flight, accomplish the applicable corrective actions in accordance with the instructions of Airbus SB A330-71-3031. (3) Accomplishment of the applicable set of corrective actions on an aeroplane as required by paragraph (2) of this AD constitutes terminating action for the repetitive inspections required by this AD for that aeroplane. (4) If no fractured bolt is detected during an inspection as required by paragraph (1) of this AD, accomplishment of the applicable set of additional inspections and, depending on findings, corrective actions as defined in, and in accordance with, the instructions of Airbus SB A330-71-3031 constitutes terminating action for the repetitive inspections required by this AD for that aeroplane.
Ref. Publications:	<p>Airbus SB A330-71-3031 original issue dated 29 January 2015.</p> <p>The use of later approved revisions of this document is acceptable for compliance with the requirements of this AD.</p>
Remarks:	<ol style="list-style-type: none"> 1. This Proposed AD will be closed for consultation on 30 June 2015. 2. Enquiries regarding this PAD should be referred to the Safety Information Section, Certification Directorate, EASA. E-mail: ADs@easa.europa.eu. 3. For any question concerning the technical content of the requirements in this PAD, please contact: AIRBUS – Airworthiness Office – EAL. E-mail: airworthiness.A330-A340@airbus.com.