



# Notification of a proposal to issue an Airworthiness Directive

**PAD No.: 16-108**

**Issued: 25 July 2016**

Note: This Proposed Airworthiness Directive (PAD) is issued by EASA, acting in accordance with Regulation (EC) 216/2008 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.

In accordance with the EASA Continuing Airworthiness Procedures, the Executive Director is proposing the issuance of an EASA Airworthiness Directive (AD), applicable to the aeronautical product(s) identified below. All interested persons may send their comments, referencing the PAD Number above, to the e-mail address specified in the 'Remarks' section, prior to the consultation date indicated.

**Design Approval Holder's Name:**

PILATUS AIRCRAFT Ltd

**Type/Model designation(s):**

PC-6 aeroplanes

**Effective Date:** [TBD - standard: 14 days after AD issue date]

**TCDS Number(s):** Switzerland No. F 56-10

**Foreign AD:** Not applicable

**Supersedure:** None

## ATA 57 – Wings – Aileron Mass-Balance Counterweight Attachment – Inspection

**Manufacturer(s):**

Pilatus Aircraft Ltd. and Fairchild Republic Company (formerly Fairchild Industries, Fairchild Heli Porter and Fairchild-Hiller Corporation).

**Applicability:**

PC-6 aeroplanes, all models, all serial numbers delivered before 01 January 2016.

**Reason:**

The proper installation of the aileron counterweight requires a combination, peculiar to each aileron, of anchor nut types, bolt types, number of washers, and the definition of the bolt torque. Some combinations of counterweight and attaching parts, which could result in reduced thread engagement, have been reported on a PC-6 aeroplane.

This condition, if not detected and corrected, may lead to a disconnection of the aileron counterweight from the aileron, possibly resulting in reduced control of the aeroplane.

To address this potential unsafe condition, Pilatus issued Service Bulletin (SB) No. 57-006 (hereafter referred to as 'the SB' in this AD) to provide inspection instructions.



For the reason described above, this AD requires identification and inspection of the affected aileron mass-balance counterweight attachment parts and, depending on findings, accomplishment of applicable corrective action(s).

**Required Action(s) and Compliance Time(s):**

Required as indicated, unless accomplished previously:

- (1) Within 12 months after the effective date of this AD, remove each aileron counterweight to determine the type and number of washers required for the installation of a counterweight on each aileron in accordance with the instructions of paragraphs 3.B.(2) and 3.B.(3) of the SB.
- (2) Before next flight after the determination as required by paragraph (1) of this AD, (re)install each aileron counterweight on the aeroplane in accordance with the instructions of paragraph 3.B.(3) of the SB.
- (3) From the effective date of this AD, it is allowed to install an aileron on an aeroplane, provided that, before installation, the aileron passes an inspection in accordance with the instructions of the SB. Installing an aileron on an aeroplane in accordance with the instructions of Pilatus AMM 01975, Rev.23 or later, is an acceptable alternative method to comply with this requirement.
- (4) From the effective date of this AD, it is allowed to install an aileron counterweight on an aeroplane, provided that, before installation, an inspection in accordance with the instructions of paragraphs 3.B.(2) of the SB is accomplished and the installation is accomplished in accordance with the instructions of paragraphs 3.B.(3) of the SB. Installing an aileron counterweight on an aeroplane in accordance with the instructions of Pilatus AMM 01975, Rev.23 or later, is an acceptable alternative method to comply with this requirement.

**Ref. Publications:**

Pilatus Aircraft Ltd SB No. 57-006, dated 13 May 2016.

The use of later approved revisions of this document is acceptable for compliance with the requirements of this AD.

**Remarks:**

1. This Proposed AD will be closed for consultation on 22 August 2016.
2. Enquiries regarding this PAD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: [ADs@easa.europa.eu](mailto:ADs@easa.europa.eu).
3. For any question concerning the technical content of the requirements in this PAD, please contact Pilatus Aircraft Ltd, Customer Support Manager, CH-6371 Stans, Switzerland  
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E-mail: [SupportPC12@pilatus-aircraft.com](mailto:SupportPC12@pilatus-aircraft.com), Website: [www.pilatus-aircraft.com](http://www.pilatus-aircraft.com).

