



Airworthiness Directive

AD No.: 2018-0045

Issued: 15 February 2018

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) 216/2008 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EC) 216/2008, Article 14(4) exemption].

Design Approval Holder's Name:

AIRBUS

Type/Model designation(s):

A350 aeroplanes

Effective Date: 01 March 2018

TCDS Number(s): EASA.A.151

Foreign AD: Not applicable

Supersedure: None

ATA 53, 55 – Fuselage / Stabilizer – Vertical Stabilizer Tension Bolts – Modification

Manufacturer(s):

Airbus

Applicability:

Airbus A350-941 aeroplanes, all manufacturer serial numbers, except those on which Airbus modification (mod) 108307 and mod 110696, or mod 110898 has been embodied in production.

Definitions:

For the purpose of this AD, the following definitions apply:

The aeroplane date of manufacture: The date of transfer of title (ownership) which is referenced in Airbus documentation at the time of first delivery to an operator.

The SB: Airbus Service Bulletin (SB) A350-55-P002.

Reason:

It was identified that the section 19 holes for the Vertical Tail Plane (VTP) tension bolts connection are not properly protected against corrosion.

This condition, if not corrected, could reduce the structural integrity of the VTP.



To address this unsafe condition, Airbus developed production mod 108307 and mod 110696 to improve protection against corrosion, and issued the SB to provide in-service modification instructions.

For the reasons described above, this AD requires a modification by adding sealant and protective treatment to the head of the section 19 VTP tension bolts connection, at the barrel nut cavities and in the surrounding area.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

Modification:

Before exceeding 36 months since the aeroplane date of manufacture, or within 6 months after the effective date of this AD, whichever occurs later, accomplish concurrently the actions as required by paragraphs (1) and (2) of this AD, in accordance with the instructions of the SB.

- (1) Apply sealant and protective treatment at tension bolt heads, on Left Hand (LH) and Right Hand (RH) side.
- (2) Remove the barrel nut plate(s) (if installed) and install shouldered cap(s) in the inner hole of the root joint fitting and apply overlapping sealant, on LH and RH side.
- (3) Apply sealant and protective treatment on barrel nut holes, on LH and RH side.

Ref. Publications:

Airbus SB A350-55-P002 original issue dated 21 December 2017.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. This AD was posted on 17 January 2018 as PAD 18-005 for consultation until 31 January 2018. The Comment Response Document can be found in the [EASA Safety Publications Tool](#), in the compressed (zipped) file attached to the record for this AD.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. For any question concerning the technical content of the requirements in this AD, please contact: continued-airworthiness.a350@airbus.com.

