



Airworthiness Directive

AD No.: 2018-0233

Issued: 26 October 2018

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

AIRBUS

Type/Model designation(s):

A319, A320 and A321 aeroplanes

Effective Date: 09 November 2018

TCDS Number(s): EASA.A.064

Foreign AD: Not applicable

Supersedure: This AD supersedes EASA AD 2013-0310 dated 20 December 2013.

ATA 53 – Fuselage – Cargo Compartment Fitting Brackets, Tack and Rivet Holes – Inspection / Repair

Manufacturer(s):

Airbus, formerly Airbus Industrie

Applicability:

Airbus A319-111, A319-112, A319-113, A319-114, A319-115, A319-131, A319-132 and A319-133, aeroplanes, all manufacturer serial numbers, except those on which Airbus modification (mod) 161306 has been embodied in production.

Airbus A320-211, A320-212, A320-214, A320-215, A320-216, A320-231, A320-232, A320-233, A321-111, A321-112, A321-131, A321-211, A321-212, A321-213, A321-231 and A321-232 aeroplanes, all manufacturer serial numbers, except those on which Airbus mod 161271 has been embodied in production.

Definitions:

For the purpose of this AD, the following definitions apply:

The inspection SB: Airbus Service Bulletin (SB) A320-53-1257 Revision (Rev.) 02.

The applicable modification SB: Airbus SB A320-53-1261 Rev. 04, SB A320-53-1360 (any revision), SB A320-53-1364 and SB A320-53-1365, as applicable.



Reason:

During a full scale fatigue test, several broken frames in the cargo compartment area between Frame (FR) 50 and FR63 have been found, especially on the cargo floor support fittings and open tack holes on left hand (LH) side.

This condition, if not detected and corrected, could affect the structural integrity of the aeroplane.

To address this unsafe condition, Airbus issued SB A320-53-1257, providing inspection instructions, and SB A320-53-1261, providing modification instructions.

Consequently, EASA published AD 2013-0310, requiring repetitive inspections of the frames in the cargo compartment area and of the cargo floor support fittings and open tack holes on the LH side and, depending on findings, accomplishment of corrective action(s). That AD also required a modification, which constituted terminating action for the required repetitive inspections.

Since that AD was issued, further analyses and widespread fatigue damage evaluations identified the need to reduce the threshold and intervals for the repetitive inspections for certain configurations, and Airbus issued the inspection SB accordingly. Airbus issued SB A320-53-1360, SB A320-53-1364 and SB A320-53-1365 to supplement SB A320-53-1261, and Service Bulletin Information Transmission (SBIT) 16-0070 providing additional information.

For the reason described above, this AD retains the requirements of EASA AD 2013-0310, which is superseded, but requires accomplishment of the repetitive inspections within reduced compliance times for certain configurations. This AD also requires additional work for those aeroplanes that have already been modified in accordance with the instructions of Airbus SB A320-53-1261 Rev. 02.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

Repetitive Inspection(s):

- (1) Within the compliance time(s) as defined in Table 1 or Table 2 of Appendix 1 of this AD, as applicable, and, thereafter, at intervals not to exceed the values as defined in Table 3 of Appendix 1 of this AD, as applicable, accomplish a rototest inspection of open tack holes and rivet holes at the cargo floor support fittings between FR50 and FR63, between Stringer (STG) 33 and STG39 (LH side only), for A320 and A321 aeroplanes, and between FR53 and FR63, between STG33 and STG39 (LH side only), for A319 aeroplanes, in accordance with the instructions of the inspection SB.

Corrective Action(s):

- (2) If, during any inspection as required by paragraph (1) of this AD, a crack is detected, before next flight, contact Airbus for approved repair instructions and, within the compliance time identified in those instructions, accomplish those instructions accordingly.



Modification:

- (3) Except as required by paragraph (4) of this AD, before exceeding 48 000 flight cycles (FC) or 96 000 flight hours (FH), whichever occurs first since aeroplane first flight, modify the aeroplane in accordance with the instructions of the applicable modification SB.

Additional Work:

- (4) For an aeroplane that was modified, before the effective date of this AD, in accordance with the instructions of Airbus SB A320-53-1261 Rev. 02, within 30 days after the effective date of this AD, contact Airbus for additional work instructions and, within the compliance time specified in those instructions, accomplish the additional work accordingly.

Credit:

- (5) Modification of an aeroplane, before the effective date of this AD, in accordance with the instructions of Airbus SB A320-53-1261 original issue, or Rev. 01, or Rev. 03, is acceptable to comply with the requirements of paragraph (3) of this AD for that aeroplane.
- (6) Inspections on an aeroplane, accomplished before the effective date of this AD, in accordance with the instructions of Airbus SB A320-53-1257 at original issue or Rev. 01, are acceptable to comply with the initial requirements of paragraph (1) of this AD for that aeroplane.

Terminating Action:

- (7) Repair of an aeroplane as required by paragraph (2) of this AD does not constitute terminating action for the repetitive inspections as required by paragraph (1) of this AD for that aeroplane, unless otherwise stated in the repair instructions.
- (8) Modification of an aeroplane as required by paragraph (3) or (4) of this AD, as applicable, constitutes terminating action for the repetitive inspections as required by paragraph (1) of this AD for that aeroplane.

Ref. Publications:

Airbus SB A320-53-1257 original issue dated 21 December 2012, or Rev. 01 dated 28 April 2016, or Rev. 02 dated 29 November 2016.

Airbus SB A320-53-1261 original issue dated 21 December 2012, or Rev. 01 dated 30 June 2015, or Rev. 02 dated 01 March 2016, or Rev. 03 dated 08 August 2016, or Rev. 04 dated 03 May 2017.

Airbus SB A320-53-1360 original issue dated 01 December 2016, or Rev. 01 dated 01 March 2018.

Airbus SB A320-53-1364 original issue dated 04 May 2018.

Airbus SB A320-53-1365 original issue dated 17 April 2018.

The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.



Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. This AD was posted on 11 July 2018 as PAD 18-093 for consultation until 08 August 2018. The Comment Response Document can be found in the [EASA Safety Publications Tool](#), in the compressed (zipped) file attached to the record for this AD.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#).
5. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – Airworthiness Office – EIAS; Fax +33 5 61 93 44 51; E-mail: account.airworth-eas@airbus.com.

Revised



Appendix 1

Table 1 - Initial Inspection

Model and Configuration	FH and FC Accumulated on 03 January 2014 [the effective date of EASA AD 2013-0310] since first flight of the aeroplane	Compliance Time
A320 and A319 PAX, pre-mod 160001 and pre-SB A320-57-1193 (mod 160080) A321 pre-mod 160021	36 200 FC or 72 400 FH, or less	Before exceeding 38 200 FC or 76 400 FH, whichever occurs first
	More than 36 200 FC or 72 400 FH, but not more than 45 000 FC	Within 2 000 FC or 4 000 FH whichever occurs first after 03 January 2014 [the effective date of AD 2013-0310]
	More than 45 000 FC	Within 1 000 FC or 2 000 FH, whichever occurs first after 03 January 2014 [the effective date of AD 2013-0310]

Table 2 - Initial Inspection

Model and Configuration	FH and FC Accumulated on the effective date of this AD since first flight of the aeroplane	Compliance Time (see Note 1 of this AD)
A320 and A319 PAX, post-mod 160001 or post-SB A320-57-1193 (mod 160080) A321 post-mod 160021	33 400 FC or 66 900 FH, or less	Before exceeding 35 400 FC or 70 900 FH, whichever occurs first since first flight of the aeroplane (see Note 2 of this AD)
	More than 33 400 FC or 66 900 FH, but no more than 40 000 FC	Within 2 000 FC or 4 000 FH, whichever occurs first since the effective date of this AD (see Note 2 of this AD)
	More than 40 000 FC but no more than 43 000 FC	Within 1 000 FC or 2 000 FH, whichever occurs first since the effective date of this AD (see Note 2 of this AD)
	More than 43 000 FC since aeroplane first flight	Within 30 days after the effective date of this AD (see Note 2 of this AD)
A319 CJ, post-mod 160001 or post-SB A320-57-1193 (mod 160080)	Not applicable	Before exceeding 19 800 FC or 85 300 FH, whichever occurs first since first flight of the aeroplane

Note 1: For A320 and A319 in post-SB A320-57-1193 configuration, refer to Airworthiness Limitations Section (ALS) Part 2 variation 3.6 or ALS Part 2 revision 4, or later further ALS Part 2 revision, for determination of the threshold when sharklet is installed.



Note 2: For A320 and A319 PAX, post-SB A320-57-1193 (mod 160080): Without exceeding the time at which inspection is required through the threshold or compliance time for A320 and A319-PAX aeroplanes in pre-SB A320-57-1193 (pre mod 160080) configuration.

Table 3 – Repetitive Inspections

Model and Configuration	Interval (FC or FH, whichever occurs first)
A320, A319 and A321	5 000 FC or 10 000 FH
A319 CJ, post mod 160001 or post-SB 57-1193 [mod 160080]	2 900 FC or 12 800 FH

Revised

