



## Notification of a Proposal to issue an Airworthiness Directive

**PAD No.: 18-130**

**Issued: 25 September 2018**

Note: This Proposed Airworthiness Directive (PAD) is issued by EASA, acting in accordance with Regulation (EC) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

In accordance with the EASA Continuing Airworthiness Procedures, the Executive Director is proposing the issuance of an EASA Airworthiness Directive (AD), applicable to the aeronautical product(s) identified below.

All interested persons may send their comments, referencing the PAD Number above, to the e-mail address specified in the 'Remarks' section, prior to the consultation date indicated.

**Design Approval Holder's Name:**

PILATUS AIRCRAFT Ltd

**Type/Model designation(s):**

PC-6 aeroplanes

**Effective Date:** [TBD - standard: 14 days after AD issue date]

**TCDS Number(s):** Switzerland No. F 56-10

**Foreign AD:** Not applicable

**Supersedure:** None

### ATA 27 – Flight Controls – Flap Actuator / Pushrod Assembly Taper Pins – Inspection

**Manufacturer(s):**

Pilatus Aircraft Ltd; and Fairchild Republic Company, formerly Fairchild Industries, Fairchild Heli Porter, Fairchild-Hiller Corporation

**Applicability:**

PC-6 aeroplanes, all models, all manufacturer serial numbers (MSN), except those equipped with electrical operated flaps.

**Definitions:**

For the purpose of this AD, the following definitions apply:

**The SB:** Pilatus Aircraft Ltd Service Bulletin (SB) No. 27-005.

**Affected part:** Left Hand (LH) and Right Hand (RH) flap actuator assemblies, having Part Number (P/N) 6132.0039.51 and P/N 6132.0039.52, and pushrod assemblies, having P/N 6132.0040.00.

**Serviceable part:** An affected part supplied as spare by Pilatus after 02 July 2018, or an affected part that has passed an inspection (no defects found), before installation, in accordance with the



instructions of Section 3.B of the SB, or has been corrected in accordance with the instructions of Section 3.C of the SB.

**Reason:**

During a recent overhaul, two new flap actuators were found to have taper pins installed that, apparently, had not been swaged. Investigation results identified that the taper pins had been incorrectly swaged during the manufacturing process.

This condition, if not detected and corrected, could lead to loss of one or both taper pins, consequent asymmetric flap deployment or flap surface flutter, possibly resulting in loss of control of the aeroplane.

To address this potential unsafe condition, Pilatus issued the SB to provide inspection instructions.

For the reason described above, this AD requires a one-time inspection of the taper pins of the affected parts for correct installation and, depending on findings, accomplishment of applicable corrective action(s). This AD also requires inspection of, and, depending on findings, corrective action(s) on, affected parts held as spare, prior to installation.

**Required Action(s) and Compliance Time(s):**

Required as indicated, unless accomplished previously:

**Inspection(s):**

- (1) During the next scheduled 100 flight hours maintenance or annual inspection, or within 12 months after the effective date of this AD, whichever occurs first, inspect the taper pins of each affected part in accordance with the instructions of section 3.B of the SB.

**Corrective Action(s):**

- (2) If, during the inspection as required by paragraph (1) of this AD, any taper pin is found damaged, incorrectly swaged, or not swaged, before next flight, accomplish the applicable corrective action(s) in accordance with the instructions of section 3.C of the SB.

**Parts Installation:**

- (3) From the effective date of this AD, it is allowed to install on any aeroplane an affected part, provided it is a serviceable part, as defined in this AD.

**Ref. Publications:**

Pilatus Aircraft Ltd SB No. 27-005 original issue dated 02 July 2018.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

**Remarks:**

1. This Proposed AD will be closed for consultation on 23 October 2018.
2. Enquiries regarding this PAD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: [ADs@easa.europa.eu](mailto:ADs@easa.europa.eu).



3. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this PAD, and which may occur, or have occurred on a product, part or appliance not affected by this PAD, can be reported to the [EU aviation safety reporting system](#).
4. For any question concerning the technical content of the requirements in this PAD, please contact: Pilatus Aircraft Ltd, Customer Support Manager, CH-6371 Stans, Switzerland  
Telephone: +41 41 619 33 33, Fax: +41 41 619 73 11  
E-mail: [SupportPC12@pilatus-aircraft.com](mailto:SupportPC12@pilatus-aircraft.com), Website: [www.pilatus-aircraft.com](http://www.pilatus-aircraft.com).

