



Airworthiness Directive

AD No.: 2019-0291

Issued: 29 November 2019

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

AIRBUS

Type/Model designation(s):

A330 and A340 aeroplanes

Effective Date: 13 December 2019

TCDS Number(s): EASA.A.015, EASA.A.004

Foreign AD: Not applicable

Supersedure: This AD supersedes EASA AD 2017-0224 dated 10 November 2017.

ATA 28 – Fuel – Fuel Pump – Inspection

Manufacturer(s):

Airbus, formerly Airbus Industrie

Applicability:

Airbus A330-201, A330-202, A330-203, A330-223, A330-223F, A330-243, A330-243F, A330-301, A330-302, A330-303, A330-321, A330-322, A330-323, A330-341, A330-342, A330-343 and A330-941 aeroplanes, all manufacturer serial numbers (MSN), and

Airbus A340-211, A340-212, A340-213, A340-311, A340-312, A340-313, A340-541, A340-542, A340-642 and A340-643 aeroplanes, all MSN.

Definitions:

For the purpose of this AD, the following definitions apply:

The AOT: Airbus Alert Operators Transmission (AOT) A28L006-17 Revision 04.

The applicable SB: Airbus Service Bulletin (SB) A330-28-3132, SB A340-28-4142 and SB A340-28-5062, as applicable, which refer to Eaton Aerospace Ltd SB 8810-28-06 Revision 2.

Affected part: Fuel pumps, having Part Number (P/N) 568-1-28300-101, or P/N 568-1-28300-103, or P/N 568-1-28300-200.



Locations:

Location A affected parts are installed at 600QL1(2), 112QA1(2), 608QL1(2), 711QN1(2)(3)(4) and 712QN1(2). These were affected by EASA AD 2017-0224.

Location B affected parts are installed at the collector cell, 121QA1(2), 122QA1(2), 100QA1(2)(3)(4) and 101QA1(2)(3)(4). These were not affected by EASA AD 2017-0224.

Erosion cases and breakthrough: The erosion cases 1, 2, 3 and breakthrough are defined and described with several pictures in Eaton Aerospace Ltd SB 8810-28-06 Revision 2 (or later revisions).

Serviceable part for installation at A locations: An affected part which is new (not previously installed) or has been repaired/overhauled by Eaton Aerospace (housing was replaced); or which, before installation, has passed an inspection (no erosion detected, or only Case 1 erosion) in accordance with the instructions of the applicable SB or AOT A28L006-17 (any revision); or a fuel pump which is not an affected part.

Serviceable part for installation at B locations: An affected part which is new (not previously installed) or has been repaired/overhauled by Eaton Aerospace (housing was replaced); or which, before installation, has passed an inspection (no erosion detected, or Case 1 or Case 2 erosion) in accordance with the instructions of the applicable SB or AOT A28L006-17 (any revision); or a fuel pump which is not an affected part.

Groups: Group 1 aeroplanes are those that have an affected part installed. Group 2 aeroplanes are those that do not have an affected part installed.

Reason:

An occurrence was reported of a fuel pump showing cavitation erosion which breached the fuel pump housing through the inlet webs and exposed the fuel pump power supply wires. Inspections accomplished on fuel pumps removed from other aeroplanes identified signs of erosion in varying degrees. A list of potentially affected fuel pump P/N was established.

This condition, if not detected and corrected, could result, in case the pump is running dry, in an ignition source in the fuel tank, which may result in a fuel tank explosion and consequent loss of the aeroplane.

To address this potential unsafe condition, Airbus issued the AOT to provide instructions to inspect the affected parts when installed at specific positions, and to update the applicable Master Minimum Equipment List (MMEL). EASA published AD 2017-0224 to require accomplishment of these actions.

Since that AD was published, Airbus issued the applicable SB and the AOT, introducing repetitive inspections of all affected parts, regardless of their position on the aeroplane.

For the reasons described above, this AD partially retains the requirements of EASA AD 2017-0224, which is superseded, expands the Applicability to include A330-941 aeroplanes and requires repetitive inspections of affected parts on all affected locations and, depending on findings, replacement of damaged affected parts with serviceable parts.



This AD is an interim action as it is expected that a new pump, more erosion resistant, will be developed and the installation of these pumps may be required by a new AD.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

Inspection of Affected Parts Installed at A Locations:

- (1) For Group 1 aeroplanes: Before an affected part exceeds 10 000 flight hours (FH) since first installation on an aeroplane and, thereafter, at intervals not to exceed the value specified in Table 1 of this AD, as applicable, depending on the detected erosion level, inspect each affected part at affected A locations in accordance with the instructions of the applicable SB or Airbus AOT A28L006-17 (any revision).

Table 1 – Fuel Pump Repetitive Inspection Intervals for Affected Parts at A Locations

Erosion	Compliance Time
No erosion	5 000 FH
Case 1 erosion	
Case 2 erosion	1 000 FH

Inspection of Affected Parts Installed at B Locations:

- (2) For Group 1 aeroplanes: Within the compliance times specified in Table 2 of this AD, depending on the FH accumulated by an affected part located at B locations on the effective date of the AD, and thereafter, at intervals not to exceed 30 months or 10 000 FH, whichever occurs first, inspect each affected part in accordance with the instructions of the applicable SB.

Table 2 – Fuel Pump Inspection Threshold for Affected Parts at B Locations

FH Accumulated (see Note 1 of this AD)	Compliance Time
50 000 or more, or FH unknown	Within 12 months after the effective date of this AD
40 000 or more	Within 18 months after the effective date of this AD, but not earlier than 6 months after the effective date of this AD
30 000 or more	Within 24 months after the effective date of this AD, but not earlier than 12 months after the effective date of this AD
20 000 or more	Within 30 months after the effective date of this AD, but not earlier than 18 months after the effective date of this AD
Less than 20 000	Within 10 000 FH or 30 months, whichever occurs first after exceeding 20 000 FH



Note 1: The FH specified in Table 1 and Table 2 of this AD are those accumulated by an affected part, on the effective date of this AD, since first installation on an aeroplane, or since last repair or overhaul by Eaton Aerospace (housing was replaced).

Corrective Action(s):

- (3) If, during any inspection as required by paragraph (1) of this AD, Case 3 erosion or breakthrough is found on an affected part, before next flight, replace that part with a serviceable part, or de-activate that fuel pump, deferring replacement in accordance with the provisions as specified in the applicable operator MEL, in accordance with the instructions of the applicable SB or the AOT.
- (4) If, during any inspection as required by paragraph (2) of this AD, breakthrough is found on an affected part, before next flight, replace that part with a serviceable part, or de-activate that fuel pump, deferring replacement in accordance with the provisions as specified in the applicable operator MEL, in accordance with the instructions of the applicable SB or the AOT.

Terminating Action:

- (5) None.

Part Installation / Switching Pump Location:

- (6) For Group 1 and Group 2 aeroplanes: From the effective date of this AD, it is allowed to install on any aeroplane an affected part in an A or B location, provided it is a serviceable part (see Definitions and restrictions in Table 3 of this AD) and that, following installation, it is inspected as required by paragraphs (1) and (2) of this AD.

Table 3 – Fuel Pump Installation

Affected Part Condition	Installation on A or B Location	Affected Part Installation
Breakthrough	A or B	Not allowed
Case 3	A or B	
Case 2	A	
	B	Allowed
Case 1	A or B	

Note 2: For the purpose of this AD, removal of an affected part from an aeroplane for inspection as required by this AD and subsequent re-installation of that part on that same aeroplane and at the same location after the inspection is not considered an “installation” as specified in paragraph (6) or Table 3 of this AD.

- (7) For Group 1 and Group 2 aeroplanes: Removing an affected part from an A location and installation of that affected part at an B location, or removing an affected part from an B location and installation at an A location, as applicable, is allowed (see paragraph (6) and restrictions in Table 3 of this AD), provided that, concurrently with installation, the affected part passes an inspection for the new location (see Definitions – serviceable part of this AD).



MMEL Changes - Dispatch Restrictions:

- (8) For Group 1 aeroplanes, except A330-941 aeroplanes: Within 30 days after 17 November 2017 [the effective date of EASA AD 2017-0224], amend the applicable MMEL, on the basis of which the operator's MEL is established, in accordance with the instructions of Airbus AOT A28L006-17 (any revision), inform all flight crews, and, thereafter, operate the aeroplane accordingly. This can be accomplished by inserting a copy of Airbus AOT A28L006-17 (any revision) into the applicable MMEL.
- (9) For Group 1 A340-500 and A340-600 aeroplanes: Concurrently with the MMEL amendment as required by paragraph (8) of this AD, amend the applicable MMEL, on the basis of which the operator's MEL is established, as indicated in Table 4 of this AD, inform all flight crews and, thereafter, operate the aeroplane accordingly.

Amendment of the MMEL can be accomplished by inserting a copy of this AD into the applicable MMEL.

Table 4 – A340-500 and A340-600 MMEL Amendment

MMEL Amendment
MMEL Item 28-27-06 and 28-27-07 can be applied, provided the related circuit breaker is pulled and tagged for the duration of the inoperative period

- (10) For Group 1 aeroplanes: Within 30 days after the effective date of this AD, amend the applicable MMEL, on the basis of which the operator's MEL is established, to include the additional items specified in the AOT in accordance with the instructions of the AOT, inform all flight crews, and, thereafter, operate the aeroplane accordingly.

Amendment of the MMEL can be accomplished by inserting a copy of the AOT or the applicable SB into the applicable MMEL.

Maintenance Action:

- (11) For all aeroplanes: From the effective date of this AD, each time defueling and ground fuel transfer operations, as specified in the AOT, are accomplished on an aeroplane, accomplish the actions and implement the restrictions in accordance with the instructions of the AOT on that aeroplane. Using the applicable Aircraft Maintenance Manual or Weight and Balance Manual task, provided that contains the instructions of the AOT for these actions, is acceptable to comply with this requirement.

Ref. Publications:

Airbus AOT A28L006-17 original issue dated 03 November 2017, Revision 01 dated 16 November 2017, Revision 02 dated 08 February 2018, Revision 03 dated 28 May 2019, or Revision 04 dated 04 September 2019.

Airbus SB A330-28-3132 original issue dated 06 March 2019.

Airbus SB A340-28-4142 original issue dated 06 March 2019.



Airbus SB A340-28-5062 original issue dated 06 March 2019.

Eaton Aerospace Ltd SB 8810-28-06 Revision 2 dated 01 March 2019.

The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. This AD was published on 24 May 2019 as PAD 19-093 for consultation until 21 June 2019 and republished on 17 October 2019 as PAD 19-093R1 for additional consultation until 31 October 2019. The Comment Response Documents can be found in the [EASA Safety Publications Tool](#), in the compressed (zipped) file attached to the record for this AD.
3. Enquiries regarding this AD should be referred to the EASA Programming and Continued Airworthiness Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#).
5. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – Airworthiness Office – EIAL; E-mail: airworthiness.A330-A340@airbus.com.

