



Notification of a Proposal to issue an Airworthiness Directive

PAD No.: 20-076

Issued: 11 May 2020

Note: Proposed Airworthiness Directive (PAD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

In accordance with the EASA Continuing Airworthiness Procedures, the Executive Director is proposing the issuance of an EASA Airworthiness Directive (AD), applicable to the aeronautical product(s) identified below.

All interested persons may send their comments, referencing the PAD Number above, to the e-mail address specified in the 'Remarks' section, prior to the consultation date indicated.

Design Approval Holder's Name:

CFM INTERNATIONAL S.A.

Type/Model designation(s):

LEAP-1A engines

Effective Date: [TBD - standard: 14 days after AD issue date]

TCDS Number(s): EASA.E.110

Foreign AD: Not applicable

Supersedure: None

ATA 72 – Engine – High Pressure Turbine Rotor Stage 2 Disc – Inspection

Manufacturer(s):

SAFRAN Aircraft Engines, formerly SNECMA (France); General Electric Aviation (United States)

Applicability:

LEAP-1A23, LEAP-1A24, LEAP-1A24E1, LEAP-1A26, LEAP-1A26CJ, LEAP-1A26E1, LEAP-1A29, LEAP-1A29CJ, LEAP-1A30, LEAP-1A32, LEAP-1A33, LEAP-1A33B2 and LEAP-1A35A engines, all serial numbers (s/n).

These engines are known to be installed on, but not limited to, certain Airbus A319, A320 and A321 aeroplanes.

Definitions:

For the purpose of this AD, the following definitions apply:

The SB: CFM International (CFMI) Service Bulletin (SB) LEAP-1A-72-00-0405-01A-930A-D.

Affected part: High pressure turbine (HPT) rotor stage 2 discs, having Part Number (P/N) 2466M52G03 and P/N 2788M26G01, and s/n as identified in Table 1 of the SB.



Serviceable part: Any HPT rotor stage 2 disc which is not an affected part; or an affected part which passed (no indication outside inspection limits found) an immersion ultrasonic inspection as defined in the SB.

Group 1 engines: Any engine with an affected part installed.

Group 2 engines: Any engine with a serviceable part installed.

Qualified shop visit: Any engine shop visit involving the removal of an affected part from the engine.

Reason:

It has been determined that a certain batch of HPT rotor stage 2 discs may have subsurface anomalies developed during the manufacturing process.

This condition, if not detected and corrected, could lead to failure of the HPT rotor stage 2 disc, possibly resulting in high energy debris release, with consequent damage to, and reduced control of, the aeroplane.

To address this potential unsafe condition, CFM issued the SB, listing the affected parts, and providing inspection instructions.

For the reason described above, this AD requires the replacement of the affected parts, and prohibits (re)installation of the affected parts on any engine.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

Replacement:

- (1) For Group 1 engines: During the next qualified shop visit after the effective date of this AD, replace the affected part with a serviceable part, as defined in this AD. This can be accomplished in accordance with the instructions of the applicable Engine Shop Manual (see Note 1 of this AD).

Note 1: After passing an immersion ultrasonic inspection as defined in the SB, an affected part is effectively a serviceable part, as defined in this AD.

Part Installation:

- (2) Do not install on any engine an affected part, as required by paragraph (2.1) or (2.2) of this AD, as applicable.
 - (2.1) For Group 1 engines: After replacement of the affected part as required by paragraph (1) of this AD.
 - (2.2) For Group 2 engines: From the effective date of this AD.



Ref. Publications:

CFMI SB LEAP-1A-72-00-0405-01A-930A-D issue 001 dated 05 March 2020.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

Remarks:

1. This Proposed AD will be closed for consultation on 08 June 2020.
2. Enquiries regarding this PAD should be referred to the EASA Programming and Continued Airworthiness Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
3. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this PAD, and which may occur, or have occurred on a product, part or appliance not affected by this PAD, can be reported to the [EU aviation safety reporting system](#).
4. For any question concerning the technical content of the requirements in this PAD, please contact: CFM International S.A., Customer Support Centre, Telephone: +33 1 64 14 88 66, Fax: +33 1 64 79 85 55, E-mail: cfm.csc@safrangroup.com,

or

CFM Inc. Aviation Operations Centre, Telephone: +1 513-552-3272, or +1 877-432-3272, Fax: +1 877-432-3329, E-mail: geae.aoc@ge.com, or aviation.fleetsupport@ge.com.

