



Notification of a Proposal to issue an Airworthiness Directive

PAD No.: 23-103

Issued: 19 September 2023

Note: This Proposed Airworthiness Directive (PAD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

In accordance with the EASA Continuing Airworthiness Procedures, the Executive Director is proposing the issuance of an EASA Airworthiness Directive (AD), applicable to the aeronautical product(s) identified below.

All interested persons may send their comments, referencing the PAD Number above, to the e-mail address specified in the 'Remarks' section, prior to the consultation date indicated.

Design Approval Holder's Name:

PILATUS AIRCRAFT Ltd

Type/Model designation(s):

PC-12 aeroplanes

Effective Date: [TBD - standard: 14 days after AD issue date]

TCDS Number(s): EASA.A.089

Foreign AD: Not applicable

Supersedure: This AD supersedes EASA AD 2008-0163 dated 27 August 2008 and EASA AD 2022-0103 dated 09 June 2022.

ATA 05 – Time Limits / Maintenance Checks – Airworthiness Limitations Section – Amendment

Manufacturer(s):

Pilatus Aircraft Ltd (Pilatus)

Applicability:

PC-12, PC-12/45, PC-12/47 and PC-12/47E aeroplanes, all Manufacturer Serial Numbers (MSN).

Definitions:

For the purpose of this AD, the following definitions apply:

The applicable ALS: Pilatus PC-12 Aircraft Maintenance Manual (AMM) Airworthiness Limitations Section (ALS) Chapter 04-00-00, Document Number 02049 Issue 01 Revision 46 (PC-12, PC-12/45 and PC-12/47), Document Number 02300 Issue 01 Revision 31 (PC-12/47E) and Document Number 02436 Issue 01 Revision 09 (PC-12/47E), as applicable to aeroplane model/configuration.

The AMP: The Aircraft Maintenance Programme (AMP) contains the tasks on the basis of which the scheduled maintenance is conducted to ensure the continuing airworthiness of each operated aeroplane. For aeroplanes operated under EU regulations, the operator or the owner ensures



compliance with the AMP as stipulated in Commission Regulation (EU) [1321/2014](#).

New and/or more restrictive tasks and limitations: This includes all tasks and limitations that are new and all tasks for which a threshold or interval was reduced, which were introduced into the ALS since the previous ALS revision that is currently incorporated in the AMP.

Reason:

The airworthiness limitations and certification maintenance instructions for Pilatus PC-12 aeroplanes, which are approved by EASA, are currently defined and published in Pilatus PC-12 AMM Chapter 04-00-00. These instructions have been identified as mandatory for continued airworthiness.

Failure to accomplish these instructions could result in an unsafe condition.

Previously, EASA issued AD 2022-0103, requiring the actions described in the Pilatus PC-12 AMM Chapter 04-00-00, Document Number 02049 Issue 01 Revision 43, Document Number 02300 Issue 01 Revision 28 and Document Number 02436 Issue 01 Revision 06.

Since that AD was issued, Pilatus published the applicable ALS, as defined in this AD, which contains new and/or more restrictive tasks and limitations, as specified in the Miscellaneous Limitations section, to introduce new repetitive inspections of the main landing gear yoke fitting, applicable to aeroplanes with hydraulic landing gear.

Previously, EASA also issued AD 2008-0163, the requirements of which have been fully included in the applicable ALS.

For the reasons described above, this AD retains the requirements of EASA AD 2008-0163 and AD 2022-0103, which are superseded, and requires accomplishment of the actions specified in the applicable ALS.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

Maintenance Tasks and Replacement of Life Limited Parts:

(1) From the effective date of this AD, accomplish the following actions, as specified in the applicable ALS (see Note 1 of this AD), as applicable to aeroplane model and depending on aeroplane configuration:

(1.1) Replace each component before exceeding the applicable life limit, and

(1.2) Within the thresholds and intervals accomplish all applicable maintenance tasks.

Note 1: For the purpose of this AD, the thresholds and intervals as defined in the 'Compliance Time' pages of the ALS include specific compliance times for certain tasks.



Corrective Action(s):

- (2) In case of finding discrepancies (as defined in the applicable ALS) during accomplishment of any task as required by paragraph (1) of this AD, within the compliance time specified in the applicable ALS, accomplish the applicable corrective action(s) in accordance with the applicable Pilatus maintenance documentation. If no compliance time is identified in the applicable ALS, accomplish the applicable corrective action(s) before next flight. If a detected discrepancy is not identified in the applicable ALS, before next flight, contact Pilatus for approved instructions and accomplish those instructions accordingly.

AMP Revision:

- (3) Within 12 months after the effective date of this AD, revise the AMP by incorporating the limitations, tasks and associated thresholds and intervals described in the applicable ALS, as applicable to aeroplane model and depending on aeroplane configuration.

Credit:

- (4) If, before the effective date of this AD, the AMP has been revised to incorporate the maintenance tasks and limitations as specified in a previous revision of the applicable ALS, that action ensures the continued accomplishment of those tasks and limitations.

Consequently, for an aeroplane to which that AMP applies, it is acceptable to accomplish the new and/or more restrictive tasks and limitations as specified in the applicable ALS, as applicable to aeroplane model and depending on aeroplane configuration, within the compliance times as specified in the applicable ALS to comply with paragraph (1) of this AD.

For that AMP, it is acceptable to incorporate the new and/or more restrictive tasks and limitations as specified in the applicable ALS, as applicable to aeroplane model and depending on aeroplane configuration, into the AMP to comply with paragraph (3) of this AD.

Recording AD Compliance:

- (5) When the AMP of an aeroplane has been revised as required by paragraph (3) or (4) of this AD, as applicable, that action ensures continued accomplishment of the tasks as required by paragraphs (1) and (2) of this AD for that aeroplane. Consequently, after revising the AMP, as required by paragraph (3) or (4) of this AD, as applicable, it is not necessary that accomplishment of individual action is recorded for demonstration of AD compliance on a continued basis.

Ref. Publications:

Pilatus PC-12, PC-12/45 and PC-12/47 AMM Chapter 04-00-00 (DMC-12-A-04-00-00-00A-000A-A issue 19), Document Number 02049 Issue 01 Revision 46 dated 02 October 2023.

Pilatus PC-12/47E AMM Chapter 04-00-00 (DMC-12-B-04-00-00-00A-000A-A issue 22), Document Number 02300 Issue 01 Revision 31 dated 02 October 2023.

Pilatus PC-12/47E AMM Chapter 04-00-00 (DMC-12-C-04-00-00-00A-000A-A issue 7), Document Number 02436 Issue 01 Revision 09 dated 02 October 2023.



The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.

Remarks:

1. This Proposed AD will be closed for consultation on 17 October 2023.
2. Enquiries regarding this PAD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
3. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this PAD, and which may occur, or have occurred on a product, part or appliance not affected by this PAD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this PAD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
4. For any question concerning the technical content of the requirements in this PAD, please contact: Pilatus Aircraft Ltd, Customer Support PC-12, CH-6371 Stans, Switzerland, Telephone: +41 41 619 33 33, Fax: +41 41 619 73 11
E-mail: SupportPC12@pilatus-aircraft.com, Website: www.pilatus-aircraft.com.

