



Airworthiness Directive

AD No.: 2025-0116

Issued: 19 May 2025

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I Part M.A.301, or Annex Vb Part ML.A.301, as applicable, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I Part M.A.303, or Annex Vb Part ML.A.303, as applicable] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

AIRBUS HELICOPTERS

Type/Model designation(s):

AS 332 helicopters

Effective Date: 02 June 2025

TCDS Number(s): EASA.R.002

Foreign AD: Not applicable

Supersedure: This AD supersedes EASA AD 2024-0168 dated 22 August 2024.

ATA 62 – Main Rotor – Swashplate Assembly – Modification

Manufacturer(s):

Airbus Helicopters (AH), formerly Eurocopter, Eurocopter France, Aérospatiale

Applicability:

AS 332 C, C1, L, L1, L2 helicopters, all serial numbers (s/n).

Definitions:

For the purpose of this AD, the following definition apply:

The ASB: AH Alert Service Bulletin (ASB) AS332-62.00.77 Revision 1.

Groups: Group 1 are AS 332 L2 helicopters, all s/n except those in post-modification (mod) 0743714 configuration, which are not Group 3.

Group 2 are helicopters which are not Group 1 and are not Group 3.

Group 3 are helicopters that, before the effective date of this AD, have been modified in accordance with the instructions of AH ASB AS332-62.00.77 at original issue or on which a Main Rotor Hub (MRH) assembly, modified in accordance with the instructions of AH ASB AS332-62.00.77 at original issue, has been installed.

Reason:

An occurrence was reported of rupture of the scissors link of the rotating swashplate assembly. It was established that the link broke because the ball joint-cups assembly was not free to turn.



Investigations identified the following findings:

- The cups and the ball joint, that are normally paired, were probably mixed/unpaired during overhaul.
- The cups and the ball joint assembly had probably not been properly lubricated during last overhaul.
- As a contributing factor, the cups and the ball joint are made of tungsten carbide, whose failure mode can be a sudden seizure of the assembly rather than play, usually detected in case of different design.

This condition, if not corrected, could lead to loss of connection between rotor shaft and rotating swashplate and, consequently, loss of control of the helicopter.

To address this potential unsafe condition, AH developed mod 0728849 to replace the ball joint-cups assembly with a one-piece self-lubricated spherical-bearing, and issued the ASB providing modification instructions for in-service helicopters; EASA issued the AD 2024-0168 requiring embodiment of that modification.

Since that AD was issued, it was identified that the original issue of the ASB AS332-62.00.77 does not provide instructions to apply anti-corrosion agent on the parts during installation. Consequently, AH revised the ASB AS332-62.00.77 to address that omission.

For the reason described above, this AD partially retains the requirements of EASA AD 2024-0168, which is superseded, and requires accomplishment of additional work.

Required Action(s) and Compliance Time(s):

Required as indicated by this AD, unless the action(s) required by this AD have been already accomplished:

Modification:

- (1) For Group 1 helicopters: Within 36 months after 05 September 2024 [the effective date of EASA AD 2024-0168], replace the rotor shaft assembly with a rotor shaft assembly in post-mod 0743714 and post-mod 0728849 configuration. Replacing the rotor shaft assembly of a helicopter can be accomplished in accordance with the instructions of the applicable AS 332 Aircraft Maintenance Manual.
- (2) For Group 2 helicopters: Within 36 months after 05 September 2024 [the effective date of EASA AD 2024-0168], modify the rotating swashplate assembly in accordance with instructions of the ASB.
- (3) For AS 332 C and AS 332 L helicopters: Before or concurrently with the modification as required by paragraph (2) of this AD for that helicopter, modify that helicopter in accordance with the instructions of the AH SB 62.00.05.
- (4) For Group 2 helicopters: Replacing the MRH assembly of a helicopter with an MRH assembly in post-mod 0728849 and post-mod 0743046 configurations is an acceptable method to comply with the requirements of paragraphs (2) and (3) of this AD, as applicable, for that helicopter.



Replacing the MRH assembly of a helicopter can be accomplished in accordance with the instructions of the applicable AS 332 Aircraft Maintenance Manual.

Part(s) Installation:

- (5) From the effective date of this AD, it is allowed to install a MRH assembly or a Rotor Shaft Assembly on a helicopter, as applicable, provided it is in post-mod 0728849 configuration.

Additional Work:

- (6) For Group 3 helicopters: Unless it can be determined that anti-corrosion agent has been applied on the parts during installation, in accordance with a procedure technically equivalent to the instructions of paragraph 3.B.2.d. of the ASB, within 6 months after the effective date of this AD, accomplish the instructions of paragraph 3.B.2.d. of the ASB.

Ref. Publications:

AH ASB AS332-62.00.77 original issue dated 23 July 2024 or revision 1 dated 15 April 2025.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. This AD was posted on 16 April 2025 as PAD 25-064 for consultation until 14 May 2025. The Comment Response Document can be found in the [EASA Safety Publications Tool](#), in the compressed ('zipped') file, attached to the record for this AD.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
5. For any question concerning the technical content of the requirements in this AD, please contact: Airbus Helicopters (Technical Support), Aéroport de Marseille Provence 13725 Marignane Cedex, France, Telephone +33 (0)4 42 85 97 97, Fax +33 (0)4 42 85 99 66, Web portal: <https://keycopter.airbushelicopters.com> > Technical Requests Management, E-mail: TechnicalSupport.Helicopters@airbus.com.

