



European Union Aviation Safety Agency

## Notification of a Proposal to issue an Airworthiness Directive

**PAD No.:** 26-024

**Issued:** 04 February 2026

Note: This Proposed Airworthiness Directive (PAD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

In accordance with the EASA Continuing Airworthiness Procedures, the Executive Director is proposing the issuance of an EASA Airworthiness Directive (AD), applicable to the aeronautical product(s) identified below.

All interested persons may send their comments, referencing the PAD Number above, to the e-mail address specified in the 'Remarks' section, prior to the consultation date indicated.

### Design Approval Holder's Name:

AIRBUS S.A.S.

### Type/Model designation(s):

A318, A319, A320 and A321 aeroplanes

**Effective Date:** [TBD - standard: 14 days after AD issue date]

**TCDS Number(s):** EASA.A.064

**Foreign AD:** Not applicable

**Supersedure:** This AD supersedes EASA AD 2016-0184 dated 13 September 2016 and AD 2025-0032 dated 10 February 2025.

## ATA 05 – Time Limits / Maintenance Checks – Airworthiness Limitations Section Part 4, System Equipment Maintenance Requirements – Amendment

### Manufacturer(s):

Airbus, formerly Airbus Industrie

### Applicability:

Airbus A318-111, A318-112, A318-121, A318-122, A319-111, A319-112, A319-113, A319-114, A319-115, A319-131, A319-132, A319-133, A319-151N, A319-153N, A319-171N, A319-173N, A320-211, A320-212, A320-214, A320-215, A320-216, A320-231, A320-232, A320-233, A320-251N, A320-252N, A320-253N, A320-271N, A320-272N, A320-273N, A321-111, A321-112, A321-131, A321-211, A321-212, A321-213, A321-231, A321-232, A321-251N, A321-251NX, A321-252N, A321-252NX, A321-253N, A321-253NX, A321-253NY, A321-271N, A321-271NX, A321-271NY, A321-272N and A321-272NX aeroplanes, all manufacturer serial numbers.

### Definitions:

For the purpose of this AD, the following definitions apply:

**The ALS:** Airbus A318/A319/A320/A321 Airworthiness Limitations Section (ALS) Part 4, System Equipment Maintenance Requirements (SEMR), Revision 11 Issue 2.



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**The AMP:** The Aircraft Maintenance Programme (AMP) contains the tasks on the basis of which the scheduled maintenance is conducted to ensure the continuing airworthiness of each operated aircraft. For aircraft operated under EU regulations, the operator or the owner ensures compliance with the AMP as stipulated in Commission Regulation (EU) [1321/2014](#).

**New and/or more restrictive tasks:** This includes all tasks that are new or for which a threshold or interval was reduced, which were introduced into the ALS, as defined in this AD, since the previous ALS revision that is currently incorporated in the AMP.

**Reason:**

The airworthiness limitations for Airbus A320 family aeroplanes, which are approved by EASA, are currently defined and published in the Airbus A318/A319/A320/A321 ALS documents. These instructions have been identified as mandatory for continued airworthiness. The airworthiness limitations applicable to the SEMR are specified in ALS Part 4.

Failure to accomplish these instructions could result in an unsafe condition.

EASA previously issued AD 2025-0032, to require accomplishment of all maintenance tasks as described in ALS Part 4, SEMR, at Revision 10, and AD 2016-0184, requiring inspection and replacement of the Trimmable Horizontal Stabilizer Actuator (THSA) for Current Engine Option (CEO) aeroplane models.

Since those ADs were issued, Airbus published the ALS (as defined in this AD) to introduce new and/or more restrictive tasks, including the inspection and replacement of the THSA for CEO aeroplane models, as required by EASA AD 2016-0184, and the changes published through the Variations 10.1 and 10.2, and also to add newly certified A320 family models.

For the reason described above, this AD retains the requirements of EASA AD 2025-0032 and AD 2016-0184, which are superseded, expands the Applicability and requires accomplishment of the actions specified in the ALS.

**Required Action(s) and Compliance Time(s):**

Required as indicated by this AD, unless the action(s) required by this AD have been already accomplished:

**Maintenance Tasks and Replacement of Life Limited Parts:**

- (1) From the effective date of this AD, accomplish the following actions, as specified in the ALS (see Note 1 of this AD), as applicable to aeroplane model and depending on aeroplane configuration:
  - (1.1) Replace each component before exceeding the applicable life limit, and
  - (1.2) Within the thresholds and intervals (see Note 1 of this AD), accomplish all applicable maintenance tasks.



Note 1: For the purpose of this AD, the thresholds and intervals as defined in the 'Compliance Time' pages of the ALS include specific compliance times for certain tasks.

**Corrective Action(s):**

- (2) In case of finding discrepancies during accomplishment of any task as required by paragraph (1) of this AD, before next flight, accomplish the applicable corrective action(s) in accordance with the applicable Airbus maintenance documentation. If a detected discrepancy cannot be corrected by using existing Airbus instructions, before next flight, contact Airbus for approved instructions and accomplish those instructions accordingly.

**AMP Revision:**

- (3) Within 12 months after the effective date of this AD, revise the approved AMP by incorporating the limitations, tasks and associated thresholds and intervals described in the ALS, as applicable to aeroplane model and depending on aeroplane configuration.

**Credit:**

- (4) If, before the effective date of this AD, the AMP has been revised to incorporate the maintenance tasks and life limitations as specified in the previous ALS revision and/or variations hereto, that action ensures the continued accomplishment of those tasks and limitations.

Consequently, for an aeroplane to which that AMP applies, it is acceptable to accomplish the new and/or more restrictive tasks and limitations as specified in the ALS, as applicable to aeroplane model and depending on aeroplane configuration, within the compliance times as specified in the ALS (see Note 1 of this AD), to comply with paragraph (1) of this AD.

For that AMP, it is acceptable to incorporate the new and/or more restrictive tasks and limitations as specified in the ALS, as applicable to aeroplane model and depending on aeroplane configuration, into the AMP to comply with paragraph (3) of this AD.

**Recording AD Compliance:**

- (5) When the AMP of an aeroplane has been revised as required by paragraph (3) or (4) of this AD, as applicable, that action ensures continued accomplishment of the tasks as required by the paragraphs (1) and (2) of this AD for that aeroplane. Consequently, after revising the AMP, as required by paragraph (3) or (4) of this AD, as applicable, it is not necessary that accomplishment of individual action is recorded for demonstration of AD compliance on a continued basis.

**Ref. Publications:**

Airbus A318/A319/A320/A321 ALS Part 4, SEMR, Revision 11 Issue 2 dated 03 December 2025.

The use of later approved variations or revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

**Remarks:**

1. This Proposed AD will be closed for consultation on 04 March 2026.



2. Enquiries regarding this PAD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: [ADS@easa.europa.eu](mailto:ADS@easa.europa.eu).
3. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this PAD, and which may occur, or have occurred on a product, part or appliance not affected by this PAD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this PAD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
4. For any question concerning the technical content of the requirements in this PAD, please contact: AIRBUS – Airworthiness Office – 1IASA; E-mail: [account.airworth-eas@airbus.com](mailto:account.airworth-eas@airbus.com).

