



Notification of a Proposal to issue an Airworthiness Directive

PAD No.: 26-031

Issued: 20 February 2026

Note: This Proposed Airworthiness Directive (PAD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

In accordance with the EASA Continuing Airworthiness Procedures, the Executive Director is proposing the issuance of an EASA Airworthiness Directive (AD), applicable to the aeronautical product(s) identified below.

All interested persons may send their comments, referencing the PAD Number above, to the e-mail address specified in the 'Remarks' section, prior to the consultation date indicated.

Design Approval Holder's Name:

AIRBUS HELICOPTERS

Type/Model designation(s):

AS 350 and EC 130 helicopters

Effective Date: [TBD - standard: 14 days after AD issue date]

TCDS Number(s): EASA.R.008

Foreign AD: Not applicable

Supersedure: None

ATA 31 – Instruments – Fuse Cover Attachment of Circuit Breaker Panel – Modification

Manufacturer(s):

Airbus Helicopters (AH), formerly Eurocopter, Eurocopter France, Aerospatiale

Applicability:

AS 350 B2, AS 350 B3 all serial numbers (s/n), on which modification (mod) 073273 and mod 073274 have been embodied, except those which mod 074280 has been embodied; and

EC 130 B4 helicopters, all s/n, on which mod 073245 has been embodied.

Definitions:

For the purpose of this AD, the following definitions apply:

Affected part: Fuse cover having Part Number (P/N) 350A76-2395-20, P/N 350A76-2395-21 or P/N 350A76-2395-23, part of the control unit P/N as listed in Appendix 1 of this AD, except those modified and reidentified in accordance with the instructions of the ASB.

The ASB: AH Alert Service Bulletin (ASB) ASB AS350-31-10-0002 original issue or ASB EC130-31-10-0001 original issue, as applicable.



Reason:

An occurrence of partial blocking of the pilot pedal was reported on a helicopter. Further investigation indicated that this event was caused by an interference between the left pedal of the pilot pedal block and the partially detached fuse cover of the circuit breaker panel.

This condition, if not corrected, could result in reduced yaw control of the helicopter.

To address this potential unsafe condition, AH published the ASB to provide instructions for reinforcement of the fuse cover attachment on the circuit breaker panel.

For the reason described above, this AD requires modification of the installation of the affected part, and prohibits installation of affected parts.

Required Action(s) and Compliance Time(s):

Required as indicated by this AD, unless the actions required by this AD have been already accomplished:

Modification:

- (1) Within 330 flight hours (FH) or 6 months, whichever occurs first after the effective date of this AD, modify the installation of the affected part in accordance with the instructions of the ASB.

Part(s) Installation:

- (2) After modification of a helicopter as required by paragraph (1) of this AD, do not instal an affected part on that helicopter.

Ref. Publications:

AH ASB AS350-31-10-0002 original issue dated 11 February 2026.

AH ASB EC130-31-10-0001 original issue dated 11 February 2026.

The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.

Remarks:

1. This Proposed AD will be closed for consultation on 06 March 2026.
2. Enquiries regarding this PAD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
3. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this PAD, and which may occur, or have occurred on a product, part or appliance not affected by this PAD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this PAD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.



4. For any question concerning the technical content of the requirements in this PAD, please contact: Airbus Helicopters Customer Support, Telephone +33 (0)4.42.85.97.89, Fax + 33 (0)4.42.85.99.66, E-mail: Airframe.Technical-Support@airbus.com, Technical Request Management: TechnicalSupport.Helicopters@airbus.com.



Appendix 1

Control Unit Affected P/N

Helicopter Model	Manufacturer P/N	P/N
AS 350 B2 AS 350 B3	052706AA	704A46580115
	052706B or 052706B-A	704A46580118
	074102AA	704A46580122
	052706CA	704A46580123
	101801AA	704A46580129
	104101AA	704A46580130
	304-2613-00	704A46580140
	304-2614-00	704A46580141
	304-2615-00	704A46580142
	304-2616-00	704A46580143
	304-2617-00	704A46580144
EC 130 B4	024803AA	704A46580112
	024803B or 024803B-A	704A46580119
	074801A or 074801AA	704A46580124
	304-2612-00	704A46580139

