

# Airworthiness Directive

## Alpha HR200 Series and R2000 Series Aircraft



Issued by the Civil Aviation Authority of New Zealand in accordance with section 72I(3A) of the Civil Aviation Act. An Airworthiness Directive (AD) contains regulatory information which is mandatory. An operator of an aircraft must not operate the aircraft unless the operator complies with every applicable AD issued by the Director in accordance with section 72I(3A) of the Civil Aviation Act. An AD is issued where the Director believes on reasonable grounds that an unsafe condition exists in an aircraft or aeronautical product.

### DCA/R2000/15A      Nose Landing Gear Leg Bracket – Inspection and Repair

- Applicability:** Model HR 200/100, HR 200/100S, HR 200/120, HR 200/120B and HR 200/160 aircraft, S/Ns 001 through 378.
- Model R 2100, R 2100 A, R 2112, R 2120 U and R 2160 D aircraft, S/Ns 001 through 378.
- Model R 2160 aircraft, S/Ns 001 through 378 and S/Ns 160A-06001 onward.
- Model R 2160i aircraft, S/Ns 001 through 378 and S/Ns 160Ai-07007 onward.
- Note 1:** The applicability of this AD revised. Alpha Aviation manufactured model R2160i aircraft added.
- Requirement:** To prevent cracks developing in the NLG leg lower bracket, inspect the plate for cracks and determine the width of the plate, per the instructions in Pierre Robin Service Bulletin No. 101, revision 3.
- Note 2:** Accomplishment of DGAC F-1983-206 satisfies the requirement of this AD.
- Compliance:** For plates with a width of less than 84mm:
- Within the next 100 hours TIS, unless already accomplished within the last 100 hours TIS. If no cracks are found inspect thereafter at intervals not to exceed 100 hours TIS.
- If cracked, determine the crack dimensions per SB No. 101, revision 3.
- If cracks are within the allowable criteria per SB No. 101, revision 3, continue with inspections at intervals not to exceed 25 hours TIS.
- If the cracks are outside the allowable crack dimensions per SB No. 101, revision 3, accomplish an approved manufacturer repair, per SB No. 101, revision 3, before further flight.
- For plates with a width of 84mm:
- Within the next 500 hours TIS, unless already accomplished within the last 500 hours TIS. If no cracks are found inspect thereafter at intervals not to exceed 500 hours TIS.
- If cracked, determine the crack dimensions per SB No. 101, revision 3.
- If cracks are within the allowable criteria per SB No. 101, revision 3, continue with inspections at intervals not to exceed 25 hours TIS.
- If the cracks are outside the allowable crack dimensions per SB No. 101, revision 3, accomplish an approved manufacturer repair, per SB No. 101, revision 3, before further flight.
- Effective Date:** DCA/R2000/15    -    29 June 2006
- DCA/R2000/15A    -    28 June 2007