


EASA	AIRWORTHINESS DIRECTIVE	
	<p>AD No: 2006 – 0292</p> <p>Date: 29 September 2006</p>	
<p>No person may operate an aircraft to which an Airworthiness Directive applies, except in accordance with the requirements of that Airworthiness Directive unless otherwise agreed with the Authority of the State of Registry.</p>		
<p>Type Approval Holder's Name:</p> <p>Enstrom Helicopter Corporation</p>	<p>Type/Model designation(s):</p> <p>Enstrom F-28 and 280 Series Helicopters</p>	
<p>TCDS Number: FAA H1CE</p>		
<p>Foreign AD: None</p>		
<p>Supersedure: United Kingdom CAA Additional AD 007-12-83 Revision 1</p>		
ATA 64	Tail Rotor – Pitch Link & Pitch Arm Plate – Inspection/Replacement	
<p>Manufacturer:</p>	<p>Enstrom Helicopter Corporation</p>	
<p>Applicability:</p>	<p>All Enstrom F-28 and 280 series helicopters.</p>	
<p>Reason:</p>	<p>This AD results from an EASA review of UK CAA Additional AD's submitted for consideration under EC 1592/2002 Article 10.1 to become EU requirements.</p> <p>In this case EASA concurs with the UK CAA that an unsafe condition exists.</p> <p>Inspection of the tail rotor pitch change links is necessary to prevent loss of yaw control.</p>	
<p>Effective Date:</p>	<p>09 October 2006</p>	
<p>Compliance:</p>	<p>Required as indicated, unless already accomplished in accordance with CAA-UK AAD 007-12-83 Revision 1:</p> <ol style="list-style-type: none"> 1) Within the next 10 flight hours (FH) after the effective date of this directive, inspect the tail rotor pitch change links for tightness of attachment between the link rod end and the drive arm on the tail rotor blade grip; 2) If loss of torque at the bolted joint is such that the AN4 pitch link bolt can be freely turned by hand, before next flight, replace the bolt with a new bolt of the same Part Number and check the pitch link bolt hole in the pitch arm plate for wear. Thereafter, at intervals not to exceed 100 FH, re-inspect the subject bolted joints for wear. 3) If the hole is worn to a dimension in excess of 0.251 inches, before next flight, replace the pitch arm plate with a serviceable part of the same Part Number. 	

Ref. Publications:	None
Remarks:	<ol style="list-style-type: none">1. If requested and appropriately substantiated the responsible EASA manager for the related product has the authority to accept Alternative Methods of Compliance (AMOCs) for this AD.2. This AD was posted as PAD 06-160 for consultation on 23 June 2006 with a comment period until 14 July 2006. No comments were received during the consultation period.3. Enquiries regarding this Airworthiness Directive should be referred to the AD Focal Point - Certification Directorate, EASA. E-mail: ADs@easa.europa.eu.4. For any question concerning the technical content of the requirements in this AD, please contact Enstrom Helicopter Corporation U.S.A. - P.O. Box 490, 2209 22nd Street - Menominee, Michigan 49858; United States of America; e-mail: customerservice@enstromhelicopter.com.