EASA	AIRWORTHINESS DIRECTIVE
	AD No.: 2007-0091 **[Corrected: 23 June 2008]** **Date: 10 April 2007** Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.

This AD is issued in accordance with EC 1702/2003, Part 21A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].

Type Approval Holder's Name :	**Type/Model designation(s) :**
AIRBUS SAS	A300 aircraft

TCDS Number :	France No. 145
Foreign AD :	Not applicable
Supersedure:	This AD supersedes DGAC France AD 1989-061-092(B) R4.

ATA 53	**Fuselage – Sections 13 to 18 Longitudinal Lap Joints – Inspection**
Manufacturer(s):	AIRBUS (formerly AIRBUS INDUSTRIE).
Applicability:	AIRBUS A300 aircraft, all certified models, serial numbers (MSN) 0003 through 0156 included.
Reason:	This Airworthiness Directive (AD) is issued in order to prevent cracks development in sections 13 to 18 of the fuselage between rivets of longitudinal lap joints between frames 18 and 80 which could affect structural integrity of the fuselage if not corrected. This new AD: - Retains the requirements of DGAC AD 1989-061-092(B)R4, which is cancelled; - Takes into account a new inspection program as detailed in AIRBUS Service Bulletins (SB) A300-53-0211 Revision 7, which will allow A300 aircraft to reach the Limit of Validity (LOV). This AD has been republished to correctly refer to SB A300-53-0211 in Note 2 of the Compliance section.
Effective Date:	24 April 2007
Required action(s) and Compliance Time(s):	**New requirements of this AD:** **1. <u>Inspect longitudinal lap joints for cracks</u>** Unless already accomplished, within 2 000 flight cycles, counted from 01 December 2006, the following actions are rendered mandatory :

	1.1. In order to prevent cracks development in sections 13 to 18 of the fuselage between rivets of longitudinal lap joints, perform the inspection program on : - the non repaired longitudinal lap joints, inner skin, - the repaired or modified longitudinal joints, in accordance with the instructions of SB A300-53-0211 revision 7 at the threshold and interval values specified either in Tables 1A, 1B, 2 of section 3.A.(3), or in Section 3.E. figure 3 of SB A300-53-0211 revision 7, as applicable. Note 1: an application tolerance of 2000 Flight Cycles counted from 01 December 2006 [date of release of SB A300-53-0211 revision 7] is applicable, as long as these values do not exceed 50% of the applicable intervals. Note 2: for special and standard areas, the areas with an application period given in Figure 3 must be modified by accomplishment of Modification Service Bulletins when the application period has been reached. Refer to Section 3.E. figure 3 of SB A300-53-0211 revision 7. **1.2.** In case of crack discovery, carry out relevant corrective actions within imposed time limits in accordance with the SB A300-53-0211 revision 7. **1.3.** Send inspection results to AIRBUS, whatever they are. **Requirements of AD 1989-061-092(B)R4** **2. Inspect fuselage inner doublers** In order to detect debonding and corrosion, inspect from inside, the bonded inner doublers of longitudinal lap joints between frames 18 and 80, in accordance with instructions of SB A300-53-0229 revision 5, at the specified threshold and interval values. In case of debonding and/or corrosion detected during inspections, perform relevant corrective actions within imposed time limits in accordance with instructions of SB A300-53-0229 revision 5.
Ref. Publications:	AIRBUS SB A300-53-0211 Revision 7 and SB A300-53-0229 Revision 5. The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.
Remarks :	1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD. 2. This AD was posted on 07 March 2007 as PAD 07-039 for consultation until 04 April 2007. No comments were received during the consultation period. 3. Enquiries regarding this AD should be referred to the Airworthiness Directives, Safety Management & Research Section, Certification Directorate, EASA; E-mail ADs@easa.europa.eu. 4. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS SAS – EAW, Airworthiness Office, Telephone + 33 5 61 93 36 96; Facsimile + 33 5 61 93 44 51.