EASA EMERGENCY AIRWORTHINESS DIRECTIVE AD No.: 2007-0197-E Date: 24 July 2007 nce with the No person may operate an aircraft to which an Airworthiness Directive applies, except in according requirements of that Airworthiness Directive unless otherwise agreed with the Authority of the te of Regist Type/Model designation Type Approval Holder's Name: **EUROCOPTER** EC 225 LP helicopters TCDS Number: EASA R.002 Foreign AD: Not applicable ASA AD 2007-0083-E dated 29 March 2007, Supersedure: This Emergency AD (EAD) supersedes and including its 'Correction' dated 30 March 2007 Main Rotor System – Mrk Hub Coning Stop Support & Dome Fairing **ATA 62** - Check Replacement / Readjustment Support Attack **ne**n **PTER** Manufacturer(s): **EURO** Applicability: helico ers, all serial numbers EATA issued successively Emergency Airworthiness Directive (EAD) 2006-40-E and EAD 2007-0083-E following a few reported cases of cracks and Ture of the main rotor hub coning stop support lugs at their attachment points on e reinforcement ring where the dome fairing is secured. Some further cracks affecting the dome fairing support assemblies, in particular, those modified per modification 332A08.0929.00 (EUROCOPTER EC225 Service Bulletin No. 62-005) have been reported afterwards. These new cracks concern by now the lugs securing the dome fairing support itself on the reinforcement ring. Failure of the coning stop support attachment lugs or of the dome fairing Reason support attachment lugs may result in the loss of the dome fairing with severe damage to the blades, but be also a danger to persons on the ground. For the reason stated above, this new EAD retains the requirements of both EAD 2006-0040-E and EAD 2007-0083-E and requires in addition to ensure at each flight-related check (every 15 flight hours or 7 days) that there is no crack in the dome fairing support lugs, in the area of their attachment points on the reinforcement ring, and to remove from service at the latest by September 1, 2007, the components resulting from modification 332A08.0929.00, embodied according to EUROCOPTER EC225 Service Bulletin (SB) No. 62-005. Effective Date: 26 July 2007

	(1) At the latest at the next flight-related check (every 15 flight hours or 7 days, the first limit reached is applicable) after the effective date of this directive, or after the previous flight-related check as required by paragraph (1) of EASA EAD 2007-0083-E, whichever occurs first, and thereafter at each flight-related check, carry out a visual check, or if in doubt a dye penetrant check, for crack detection inspection in the area of the attachment points on the main rotor hub reinforcement ring of the lugs securing the coning stop support and also of the lugs securing the dome fairing support, in accordance with the instructions given in paragraph 2.B.1 of EUROCOPTER EC225 Alert Service Bulletin (ASB) No. 05A003 Revision 1;
	(2) If a crack is found in one of the five lugs of the coning stop support and / or in one of the five lugs of the dome fairing support, before resuming flights, remove and replace the coning stop support assembly and or remove and replace the dome fairing support before resuming flights.
Compliance:	 (3) To be able to reach the place where the coning stort upport a sembly or the dome fairing support will be replaced, flight without the dome fairing itself is permitted on the condition that: the dome fairing and its attachment screws are received in accordance with the instructions given in the Note in paragraph 2. 1. but the referenced ASB; and the flight crews are informed that whe flying with lit the dome fairing, the lateral vibration of the helicopter significantly deteriorates at speeds of 70 to 120 KIAS but that this phenomenon only after the comfort of the helicopter and in no way jeopardizes flight safety.
	(4) For helicopters that have been in diffied in accordance with EUROCOPTER EC225 SB No. 62-005 (modification 332, 08.0929.00), in addition to paragraphs (1), (2) and (3) after a latest at the next flight-related check (every 15 flight and or 7 lays, the first limit reached is applicable) after the effection late of this directive, or after the previous flight-related check as required by paragraph (4) of EASA EAD 2007-0083-E, whichever occurs first and the reafter at intervals not exceeding 25 flight hours, readjust the fall-rening torque of the screws securing the dome fairing support to the hinforcement ring, in accordance with the instructions given in paragraph 2.B. of EUROCOPTER EC225 ASB No. 05A003 Revision 1;
	(5) Figure effective date of this directive, modification of any helicopters in ccolorance with EUROCOPTER EC225 SB No. 62-005 (mod 32A08 29.00) is no longer authorized;
	Not later than 01 September 2007, all components installed and/or modified as a result of modification 332A08.0929.00 specified in EUROCOPTER EC225 SB No. 62-005 must be removed from service.
Pof Publications	EUROCOPTER EC225 Alert Service Bulletin No. 05A003 Revision 1 or later approved revisions.
	 If requested and appropriately substantiated the responsible EASA manager for the related product has the authority to accept Alternative Method of Compliance (AMOCs) for this AD.
	The safety assessment has requested not to implement the full consultation process and an immediate publication and notification.
Remarks:	 Enquiries regarding this AD should be addressed to Focal Point Certification Directorate, EASA; E-mail: Ads@easa.europa.eu
	 For any questions concerning the technical content of the requirements in this AD, please contact: EUROCOPTER (STDI)-Aéroport de Marseille Provence 13725 Marignane Cedex- France; Tel: +33 (4) 42 85 97 97; Fax: +33 (4) 42 85 99 66. E-mail: directive.technical-support@eurcopter.com