


EASA	AIRWORTHINESS DIRECTIVE	
	<p>AD No.: 2008-0107</p> <p>Date: 23 June 2008</p> <p>Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation</p>	
<p>This AD is issued in accordance with EC 1702/2003, Part 21A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].</p>		
<p>Type Approval Holder's Name :</p> <p>328 Support Services GmbH</p>		<p>Type/Model designation(s) :</p> <p>328-100 aircraft</p>
<p>TCDS Number: EASA A.096</p>		
<p>Foreign AD: Not applicable</p>		
<p>Supersedure: This EASA AD supersedes Luftfahrt-Bundesamt (LBA) AD 2003-383 dated 13 November 2003, EASA Approval Number 2003-922.</p>		
<p>ATA 27</p>		
Flight Controls – Rudder Spring Tab Lever Assembly – Inspection / Replacement		
<p>Manufacturer(s): Dornier Luftfahrt GmbH; Fairchild-Dornier GmbH; AvCraft Aerospace GmbH</p>		
<p>Applicability: Model 328-100 aircraft, all serial numbers, if a Part Number (P/N) 001A272A4020-002 rudder spring tab lever assembly is installed.</p>		
<p>Reason:</p> <p>On 14 March 2002, an incident occurred with a Dornier 328-100 where the captain reported that the rudder was unresponsive. The aircraft landed without any further difficulties. A visual inspection of the rudder assembly was carried out and the spring tab assembly was found to be cracked and partially missing. During subsequent inspections of other aircraft, a number of additional rudder spring tab lever assemblies were found cracked.</p> <p>This condition, if not corrected, could lead to failure of the rudder flight control system and consequent loss of control of the aircraft. To address and correct this unsafe condition, LBA issued AD 2003-383 and 2003-384 for the Dornier 328-100 and 328-300 respectively, to require the initial and repetitive inspection of the rudder spring tab lever assembly and, in case cracks were found, the replacement of the rudder spring tab lever assembly with a serviceable unit.</p> <p>The current TC holder of this type design, 328 Support Services GmbH, has recently published Alert Service Bulletin ASB-328-27-036 Revision 2, which reduces the inspection interval to A-check [400 FH]. In addition, Service Bulletin SB-328-27-459 was revised to change the compliance status from 'optional' to 'mandatory' and instructs operators to replace the rudder spring tab lever assembly with an improved unit P/N 001A272A4020-004, ending the need for the repetitive inspections.</p> <p>For the reasons described above, this EASA AD retains the repetitive</p>		

	inspection requirements of LBA AD 2003-383, which is superseded, expands the applicability to all serial numbers, reduces the inspection interval to 400 FC and requires the replacement of the rudder spring tab lever assembly with an improved unit P/N 001A272A4020-004 as specified in SB-328-27-459.
Effective Date:	07 July 2008
Required action(s) and Compliance Time(s):	<p>Required as indicated, unless accomplished previously:</p> <ol style="list-style-type: none"> (1) Within the next 400 FH after 13 November 2003 [the effective date of LBA AD 2003-383] and thereafter at intervals not to exceed the C-Check interval, inspect the rudder spring tab lever assembly in accordance with ASB-328-27-036. (2) Within 400 FH after the last inspection as required by paragraph (1) of this AD [and previously required by LTA 2003-383], or within 400 FH after the effective date of this AD, whichever occurs later, and thereafter at intervals not to exceed 400 FH, inspect the rudder spring tab lever assembly in accordance with ASB-328-27-036 Revision 2. (3) When cracks are found in the rudder spring tab lever assembly during any inspection as required by paragraph (1) or (2) of this AD, before next flight, replace the rudder spring tab lever assembly in accordance with the instructions of ASB-328-27-036. (4) Inspections and corrective actions [replacement as required by paragraph (3) of this AD] accomplished prior to the effective date of this AD in accordance with ASB-328-27-036 at original issue or Revision 1 are acceptable to comply with the initial requirements of paragraph (1) of this AD. After the effective date of this AD, repetitive inspections and corrective actions must be accomplished in accordance with ASB-328-27-036 at Revision 2. (5) Within the next 6 months after the effective date of this AD, replace the P/N 001A272A4020-002 rudder spring tab lever assembly with an improved unit P/N 001A272A4020-004 in accordance with the instructions of SB-328-27-459. (6) After accomplishment of the replacement as required by paragraph (5) of this AD, the repetitive inspections of paragraph (2) are no longer required.
Ref. Publications:	<p>AvCraft Aerospace GmbH ASB-328-27-036 dated 12 February 2003 and Revision 1 dated 07 May 2004 and 328 Support Services GmbH ASB-328-27-036 Revision 2 dated 24 January 2008.</p> <p>AvCraft Aerospace GmbH SB-328-27-459 dated 03 May 2004 and 328 Support Services GmbH SB-328-27-459 Revision 1 dated 24 January 2008.</p> <p>The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.</p>
Remarks :	<ol style="list-style-type: none"> 1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD. 2. The required actions and the risk allowance have granted the issuance of a Final AD with Request for Comments, postponing the public consultation process until after publication. 3. Enquiries regarding this AD should be referred to the Airworthiness Directives, Safety Management & Research Section, Certification Directorate, EASA; E-mail: ADs@easa.europa.eu. 4. For any question concerning the technical content of the requirements in this AD, please contact: 328 Support Services GmbH Global Support Center, P.O. Box 1252 D-82231 Wessling, Federal Republic of Germany; Telephone: +49 8153 88111 6666 ; Fax 49 8153 88111 6565; E-mail: gsc.op@328support.de