


EASA	AIRWORTHINESS DIRECTIVE	
	<p>AD No.: 2009-0057</p> <p>Date: 13 March 2009</p> <p>Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.</p>	
<p>This AD is issued in accordance with EC 1702/2003, Part 21A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].</p>		
<p>Type Approval Holder's Name :</p> <p>AIRBUS SAS</p>	<p>Type/Model designation(s) :</p> <p>A310 aeroplanes</p>	
<p>TCDS Number : France No 145</p>		
<p>Foreign AD : Not applicable</p>		
<p>Supersedure : None</p> <p>Note: This AD replaces the requirements of §§ 1.15 and 1.17 of EASA AD 2007-0053R1, which has been revised.</p>		
<p>ATA 57</p>	<p>Wings – Centre Wing Trellis Boom Drainage Holes & Frame 40 Upper Corner Fitting – Inspection / Repair</p>	
<p>Manufacturer(s):</p>	<p>Airbus (formerly Airbus Industrie)</p>	
<p>Applicability:</p>	<p>A310-203, A310-203C, A310-204, A310-221, A310-222, A310-304, A310-308, A310-322, A310-324 and A310-325 aeroplanes, all serial numbers.</p> <p>Note: The detailed applicability for each of the actions required by this AD is identified in each relevant "Action" paragraph of the 'Required Action(s) and Compliance Time(s)' section of this AD.</p>	
<p>Reason:</p>	<p>DGAC France AD 1992-106-132(B) original issue up to revision 7 was issued to require a set of inspection and modification tasks which addressed JAR/FAR 25-571 requirements related to damage-tolerance and fatigue evaluation of structure.</p> <p>Following the Extended Design Service Goal activities as part of the Structure Task Group for the Airbus A310 program, EASA published AD 2007-0053, which replaced DGAC France AD F-1992-106-132R7.</p> <p>Since the issuance of AD 2007-0053R1, the thresholds and the intervals of Airbus Service Bulletins (SB) A310-57-2050 and A310-57-2064 have been updated.</p> <p>Consequently, this new AD takes over the requirements of paragraphs 1.15 and 1.17 of EASA AD 2007-0053R1, which has been revised accordingly. In addition, and requires the accomplishment of Airbus SB A310-57-2048 at revision 01.</p>	
<p>Effective Date:</p>	<p>27 March 2009</p>	

<p>Required Action(s) and Compliance Time(s):</p>	<p>Required as indicated, unless accomplished previously:</p> <p><u>Action No.1: Cold working of Trellis Boom Drainage Holes.</u></p> <p>Applicable to A310-203, A310-203C, A310-204, A310-222, A310-304, A310-322 and A310-324 aeroplanes, all serial numbers, except those aeroplanes on which:</p> <ul style="list-style-type: none"> - Airbus modification 06130 was embodied before delivery, or - Airbus SB A310-57-2048 has been embodied in service, or - any rework of cracked drain holes was previously performed in accordance with Airbus SB A310-57-2050 instructions, as applicable. <p>(1) No later than the applicable thresholds defined in SB A310-57-2048 revision 01, perform the cold working of trellis boom drainage holes in accordance with Airbus SB A310-57-2048 Revision 01.</p> <p><u>Action No.2: Inspection of Trellis Boom Drainage Holes.</u></p> <p>Applicable to A310-203, A310-203C, A310-204, A310-221, A310-222, A310-304, A310-308, A310-322, A310-324 and A310-325 aeroplanes, all serial numbers .</p> <p>(2) No later than the applicable thresholds defined in SB A310-57-2050 revision 01, perform the initial inspection of trellis boom drainage holes and, depending on findings, apply the corrective measures within the time period defined, in accordance with Airbus SB A310-57-2050 revision 01.</p> <p>Aeroplanes which have exceeded the values of the inspection thresholds must be inspected within the grace periods defined in SB A310-57-2050 revision 01, taking the effective date of this directive as a starting date, and without exceeding the previous values quoted in the Airbus SB A310-57-2050 at original issue.</p> <p>Note 1: For aeroplanes previously inspected in accordance with Airbus SB A310-57-2050 at original issue, no further action is required by paragraph (2) of this AD.</p> <p>(3) Thereafter, repeat the inspection at the intervals and according to the instructions defined in Airbus SB A310-57-2050 revision 01 and, depending on findings, apply the corrective measures within the time period defined, in accordance with Airbus SB A310-57-2050 revision 01.</p> <p>For the first planned repeat inspection to occur after the effective date of this directive, a grace period can be applied, without exceeding the interval value quoted in the Airbus SB A310-57-2050 original issue.</p> <p><u>Action No.3: Inspection of Fuselage Frame 40 Upper Corner Fitting.</u></p> <p>Applicable to A310-203, A310-203C, A310-204, A310-221, A310-222, A310-304, A310-308, A310-322, A310-324 and A310-325 aeroplanes, all serial numbers.</p> <p>(4) No later than the applicable threshold defined in SB A310-57-2064 revision 02, perform the initial inspection of the FR40 upper corner fitting LR & RH sides and, depending on findings, apply the corrective measures within the time period defined, in accordance with Airbus SB A310-57-2064 revision 02.</p> <p>Aeroplanes which have exceeded the values of the inspection threshold must be inspected in the grace period defined in SB A310-57-2064 revision 02, taking the effective date of this directive as a starting date, and without exceeding the previous values quoted in SB A310-57-2064 at revision 01.</p>
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	<p>Note 2: For aeroplanes previously inspected in accordance with Airbus SB A310-57-2064 original issue or revision 01, no further action is required by paragraph (4) of this AD.</p> <p>(5) Thereafter, repeat the inspection at the intervals and according to the instructions defined in Airbus SB A310-57-2064 revision 02 and, depending on findings, apply the corrective measures within the time period defined, in accordance with Airbus SB A310-57-2064 revision 02.</p> <p>For the first planned repeat inspection which occurs after the effective date of this directive, a delay (grace period) can be applied, without exceeding the interval value quoted in Airbus SB A310-57-2064 revision 01.</p> <p>(6) The accomplishment of corrective measures in accordance with Airbus SB A310-57-2050 and A310-57-2064 (at any revision) does not constitute terminating action for the repetitive inspection requirements of this AD.</p>
Ref. Publications:	<p>AIRBUS Service Bulletins :</p> <p>A310-57-2048 original issue or revision 01</p> <p>A310-57-2050 original issue or revision 01</p> <p>A310-57-2064 original issue, revision 01 or revision 02</p> <p>The use of a later approved revision of these documents is acceptable for compliance with the requirements of this AD.</p>
Remarks :	<ol style="list-style-type: none"> 1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD 2. This AD was published on 13 January 2009 as PAD 09-013 for consultation until 03 February 2009. The Comment Response Document can be found at http://ad.easa.europa.eu/. 3. Enquiries regarding this AD should be referred to the Airworthiness Directives, Safety Management & Research Section, Certification Directorate, EASA. E-mail ADs@easa.europa.eu. 4. For any question concerning the technical content of the requirements in this AD, please contact: Airbus SAS – EAW (Airworthiness Office, Telephone:+ 33 5 61 93 36 96, Fax:+ 33 5 61 93 44 51)