


EASA	AIRWORTHINESS DIRECTIVE	
	<p>AD No.: 2009-0081R1</p> <p>Date: 30 July 2010</p> <p>Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.</p>	
<p>This AD is issued in accordance with EC 1702/2003, Part 21A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008 Article 1(4) exemption].</p>		
<p>Type Approval Holder's Name :</p> <p>AIRBUS</p>	<p>Type/Model designations (s) :</p> <p>A300 and A300-600 aeroplanes</p>	
<p>TCDS Number : France No. 145</p>		
<p>Foreign AD : Not applicable</p>		
<p>Revision : This AD revises AD 2009-0081, dated 06 April 2009.</p>		
<p>ATA 57</p>	<p>Wings – Main Landing Gear (MLG) Rib 5 Attachment Fitting Lower Flange Inspection and Repair</p>	
<p>Manufacturer(s):</p>	<p>Airbus (formerly Airbus Industrie).</p>	
<p>Applicability:</p>	<p>Airbus A300B2-1C, A300B2-203, A300B2K-3C, A300B4-103, A300B4-120, A300B4-203, A300B4-2C, A300C4-203 and A300F4-203 aeroplanes, all serial numbers,</p> <p>Airbus A300B4-601, A300B4-603, A300B4-605R, A300B4-620, A300B4-622, A300B4-622R, A300C4-620 and A300F4-605R aeroplanes, all serial numbers,</p> <p>except aeroplanes modified in production with Airbus modification 11912 or aeroplanes on which Airbus modification 11932 has been embodied before initial entry into service.</p> <p>In addition, the requirements of this AD do not apply to:</p> <ul style="list-style-type: none"> - aeroplanes without Airbus modification 11912 on which MLG ribs 5 have been replaced in service on both the left hand (LH) and the right hand (RH) wings. - aeroplanes that have had the holes 47 and 54 on both the LH and RH wings repaired in accordance with repair drawing R57240226. <p>Note: If a new MLG rib 5 has been installed on one wing only, or if repair drawing R57240226 has been performed on one wing only, then the mandatory actions specified in this AD are still required on the MLG rib 5 installed on the other wing.</p>	

Reason:	<p>Following the occurrence of cracks on the MLG Rib 5 RH and LH attachment fitting lower flanges, DGAC France AD 2003-318(B) was issued to require repetitive inspections and, as terminating action, the embodiment of Airbus Service Bulletins (SB) A300-57-0235 and A300-57-6088 not later than 31 December 2004.</p> <p>Subsequently, new cases of cracks were discovered during scheduled maintenance checks by operators of A300B4 and A300-600 type aeroplanes on which the terminating action SB's were embodied. This condition, if not corrected, could affect the structural integrity of those aeroplanes.</p> <p>To address and correct this condition, Airbus developed an inspection programme for aeroplanes modified in accordance with SB A300-57-0235 or A300-57-6088. This inspection programme was required to be implemented by DGAC France AD F-2005-113, original issue and later revision 1.</p> <p>A new EASA AD 2008-0111, superseding DGAC France AD F-2005-113R1, was issued to reduce the applicability. For aeroplanes already compliant with DGAC France AD F-2005-113R1, no further action was required.</p> <p>Since EASA AD 2008-0111 issuance, Airbus reviewed the inspection programmes of SB A300-57A0246 and SB A300-57A610 to introduce repetitive inspections including a new inspection technique for holes 47 and 54 and to reduce inspections threshold and intervals from 700 Flight Cycles (FC) to 400 FC until a revised terminating action is made available.</p> <p>For the reasons stated above, AD 2009-0081 superseded EASA AD 2008-0111 and required operators to comply with the new inspection programme introduced in Revisions 3 of Airbus SB A300-57A0246 and Airbus SB A300-57A6101.</p> <p>This AD is revised to introduce an optional terminating action which consists of spot-facing the sensitive holes of the MLG Rib 5 (LH and RH) bottom flanges.</p>
Effective Date:	<p>Revision 1 : 17 August 2010</p> <p>Original issue : 20 April 2009</p>
Required Action(s) and Compliance Time(s):	<p>Required as indicated:</p> <p>(1) Within 400 FC following embodiment of Airbus SB A300-57-0235 or Airbus SB A300-57-6088 (at any revision), as applicable to aeroplane model, or</p> <p>Within 400 FC or 4 months, whichever occurs first after the effective date of this AD at original issue, for those aeroplanes that have already exceeded 400 FC from embodiment of Airbus SB A300-57-0235 or Airbus SB A300-57-6088 (at any revision), as applicable to aeroplane model, perform a Detailed Visual Inspection (DVI) for cracks at:</p> <ul style="list-style-type: none"> (a) the bottom flange and vertical web in the area between the wing rear spar/gear rib 5 attachment and the forward reaction-rod pick up lug, (b) on the inboard side, around the fastener holes at locations 43, 47 to 50, 52 and 54, (c) on the outboard side, the lower flange, the vertical web and around the fastener, holes at locations 43, 47 to 50, 52 and 54, <p>followed, if no is crack detected, by a Fluorescent Penetrant Inspection (FPI) at holes location 47 and 54, in the RH and/or LH MLG rib 5 attachment fitting lower flange, in accordance with the instructions of Airbus SB A300-57A0246 Revision 03 or Airbus SB A300-57A6101 Revision 03, as applicable to aeroplane model.</p> <p>Actions performed prior to the effective date of this AD at original issue in</p>

	<p>accordance with the instructions of any previous issues of Airbus SB A300-57-0246 or Airbus SB A300-57-6101 are acceptable for compliance with the corresponding paragraph (1)(a), (1)(b) or (1)(c) of this AD, as long as the inspected areas and the methods used are equivalent to the ones specified in Airbus SB A300-57A0246 Revision 03 or Airbus SB A300-57A6101 Revision 03, as applicable to aeroplane model, and paragraph (1) of this AD.</p> <p>Note: A High Frequency Eddy Current inspection can be carried out as secondary inspection at holes location 47 and 54, in accordance with the instructions of Airbus SB A300-57A0246 Revision 03 or Airbus SB A300-57A6101 Revision 03 at the operator discretion.</p> <p>(2) Thereafter, at intervals not to exceed 400 FC, repeat the DVI and FPI inspections, in accordance with the instructions of Airbus SB A300-57A0246 Revision 03 or Airbus SB A300-57A6101 Revision 03, as applicable to aeroplane model.</p> <p>(3) If any cracks are detected during any of the inspections as required by paragraph (1) or (2) of this AD, before next flight, contact Airbus for approved repair solutions and repair the aeroplane accordingly.</p> <p><u>OPTIONAL TERMINATING ACTION:</u></p> <p>(4) After modification of an aeroplane by spot-facilit of the sensitive holes on the bottom flange MLG Rib 5 LH and RH in accordance with the instructions of Airbus SB A300-57-0254 or Airbus A300-57-6110, as applicable to aeroplane model, the inspections of this AD are no longer required for that aeroplane.</p>
Ref. Publications:	<p>Airbus Service Bulletins:</p> <p>A300-57-0235 original issue up to revision 05 A300-57-6088 original issue up to revision 04</p> <p>A300-57A0246 Revision 03 A300-57A6101 Revision 03</p> <p>A300-57-0254 original issue A300-57-6110 original issue</p> <p>The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.</p>
Remarks :	<ol style="list-style-type: none"> 1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD. 2. The required actions and the risk allowance have granted the issuance of a Final AD with Request for Comments, postponing the public consultation process after publication. 3. Enquiries regarding this AD should be referred to the Airworthiness Directives, Safety Management & Research Section, Certification Directorate, EASA; E-mail ADs@easa.europa.eu. 4. For any question concerning the technical content of the requirements in this AD, please contact: Airbus SAS – EAW (Airworthiness Office, Telephone: + 33 5 61 93 36 96, Fax:+ 33 5 61 93 44 51).