


<b>EASA</b>	<b>AIRWORTHINESS DIRECTIVE</b>	
	<p><b>AD No.: 2009-0258</b></p> <p><b>Date: 10 December 2009</b></p> <p>Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.</p>	
<p>This AD is issued in accordance with EC 1702/2003, Part 21A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].</p>		
<b>Type Approval Holder's Name :</b>		<b>Type/Model designation(s) :</b>
AIRBUS		A310 and A300-600 aeroplanes
TCDS Number : France TCDS No 145		
Foreign AD : Not applicable		
Supersedure : None		
<b>ATA 29</b>	<b>Hydraulic Power – Ram Air Turbine (RAT) Balance Weight Screws – Inspection / Replacement</b>	
Manufacturer(s):	Airbus (formerly Airbus Industrie)	
Applicability:	A310 and A300-600 aeroplanes, all certified models, all manufacturer serial numbers, if equipped with a Hamilton Sundstrand RAT Turbine Assembly, as identified by Part Number (P/N) in Hamilton Sundstrand Service Bulletin (SB) 730816-29-15 for A310 aeroplanes, and in Hamilton Sundstrand SB 732365-29-7 for A300-600 aeroplanes.	
Reason:	<p>Hamilton Sundstrand (HS), the manufacturer of the RAT, reported the failure during a wind tunnel test of a balance weight fastening screw on the RAT turbine cover. After investigation, it has been discovered that a batch of screws, which are used to attach the balance washers of the HS RAT Turbine Assembly, has not been subject to the correct heat treatment and are consequently exposed to potential fracture.</p> <p>This condition, if not corrected, might lead to the ejection of screw heads and consequently to the detachment of the associated balance washers. The loss of balance washers could increase RAT vibrations, which might lead to a possible detachment of RAT parts and consequent loss of RAT functionality. The loss of the RAT, in combination with a total engine flame out, could result in loss of control of the aeroplane.</p> <p>For the reasons described above, this AD requires the identification of the affected RAT turbine assemblies and replacement of all balance weight screws or, in case balance washer detachment is found, replacement of the RAT turbine assembly.</p>	
Effective Date:	24 December 2009	

<p>Required Action(s) and Compliance Time(s):</p>	<p>Required as indicated, unless accomplished previously:</p> <p>(1) Before the next RAT spin test, or within 1 500 flight hours (FH) or 9 months after the effective date of this AD, whichever occurs first:</p> <ul style="list-style-type: none"> <li>- Identify the P/N and s/n of the RAT turbine assembly, and</li> <li>- In case the P/N and s/n of the RAT turbine assembly are listed in the applicable HS SB, or if the P/N or s/n cannot be identified, inspect the RAT turbine assembly in accordance with the instructions of Airbus All Operator Telex (AOT) A300-29A6062 or Airbus AOT A310-29A2098, as applicable to aeroplane model:</li> </ul> <p>(1.1) If all balance screws are fitted on the turbine, within 1 500 FH or 9 months, whichever occurs first after the effective date of this AD, either:</p> <ul style="list-style-type: none"> <li>- replace the RAT turbine assembly, or</li> <li>- replace all balance screws on the RAT turbine assembly.</li> </ul> <p>(1.2) If one or more screws are fractured but the associated balance washers are still fitted on the RAT turbine assembly, before next flight, perform one of the actions as specified in paragraph (1.1) of this AD.</p> <p>(1.3) If one or more screws are fractured and any balance washer is missing, before next flight, replace the RAT turbine assembly.</p> <p>(2) Within 10 days after accomplishment of the inspection required by paragraph (1) of this AD, report the results, including no findings, to Airbus.</p> <p>(3) After the effective date of this AD, do not install on an aeroplane a RAT turbine assembly as identified by P/N in HS SB 730816-29-15 or HS SB 732365-29-7, as applicable to aeroplane model, unless it has been inspected and, if necessary, corrected in accordance with the requirements of this AD.</p>
<p>Ref. Publications:</p>	<p>Airbus AOT A300-29A6062 at original issue. Airbus AOT A310-29A2098 at original issue.</p> <p>The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.</p> <p>Hamilton Sundstrand SB 730816-29-15. Hamilton Sundstrand SB 732365-29-7.</p>
<p>Remarks :</p>	<ol style="list-style-type: none"> <li>1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.</li> <li>2. This AD was posted on 07 October 2009 as PAD 09-120 for consultation until 04 November. No comments were received during the consultation period.</li> <li>3. Enquiries regarding this AD should be referred to the Airworthiness Directives, Safety Management &amp; Research Section, Certification Directorate, EASA. E-mail <a href="mailto:ADs@easa.europa.eu">ADs@easa.europa.eu</a>.</li> <li>4. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS SAS – EAW (Airworthiness Office, Telephone: + 33 5 61 93 36 96, Fax: + 33 5 61 93 44 51).</li> </ol>