EASA

EMERGENCY AIRWORTHINESS DIRECTIVE

AD No.: 2010-0026-E

Date: 19 February 2010

Note: This Emergency Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.

This EAD is issued in accordance with EC 1702/2003, Part 21A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].

Type Approval Holder's Name: EUROCOPTER		Type/Model designation(s): EC 120 B Helicopters	
TCDS Number:	DGAC 189		
Foreign AD: Not Applicable			
Supersedure: None			
ATA 62	Main Rotor Head- Rotor	Hub Inspection	
Manufacturer(s):	Eurocopter (formerly Euroco	opter France, Aerospatiale)	
Applicability:	Eurocopter helicopter mode Rotor Head with the followin	ls EC 120 B all serial numbers, if equipped with Main g Hub part numbers:	
	C622A1002103, C622A1003	2104, C622A1002105	
		ed that an EC 120 B helicopter has performed an set of amplitude vibrations originating from the main	
Reason:		on, it was determined that the Main Rotor Head attachment area of one of the three drag damper	
		ed above, this Emergency AD requires a repetitive ng eventual cracks to preclude any risk of MRH hub	
Effective Date:	21 February 2010		
Required Action(s) and Compliance	Required as indicated, unles	ss already accomplished:	

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Time(s):	(1) Within the next 15 flight hours after the effective date of this AD and thereafter, at intervals not exceeding 15 flight hours, perform a visual inspection to detect cracks in the inspection areas of the rotor hub in accordance with the instructions of paragraph 2.B.2 of Emergency Alert Service Bulletin Eurocopter EC 120 05A012 - Revision 1 - (EASB).	
	(2) If no crack is found, re-install the dome fairing if it was removed previously and resume flight in accordance with the instructions of § 2.B.2, of Eurocopter EASB.	
	(3) If one or more cracks are found, before next flight, contact Eurocopter as described in the NOTE 1 of paragraph 2.A of Eurocopter EASB, and replace the affected rotor hub with a new rotor hub in accordance with § 2.B.2 of Eurocopter EASB.	
	(4) If during inspection accomplished in accordance with § (1) of this AD local deterioration is found, remove the finish paint until the P05 primer coat become visible and visually inspect the concerned areas in accordance with § 2.B.2.b.3. Accomplish the relevant corrective actions in accordance with § 2.B.2.b.1 or § 2.B.2.b.2.	
	(5) Replacement of the rotor hub in compliance with § (3) or § (4) of this AD does not terminate the repetitive inspection requirements of § (1) of this AD.	
	Eurocopter EC 120 B EASB 05A012 – Revision 1, dated 19 February 2010.	
Ref. Publications:	The use of later approved revisions of this document is acceptable for compliance with the requirements of this AD.	
	If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.	
	The safety assessment has requested not to implement the full consultation process and an immediate publication and notification.	
Remarks:	3. Enquiries regarding this AD should be referred to the Airworthiness Directives, Safety Management & Research Section, Certification Directorate, EASA; E-mail: ADs@easa.europa.eu .	
	4. For any question concerning the technical content of the requirements in this AD, please contact: EUROCOPTER (STDI), Aéroport de Marseille Provence, 13725 Marignane Cedex, France Telephone: +33 (0) 4 42 85 97 97 - Fax +33 (0) 4 42 85 99 66	
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