EASA AD No.: 2012-0013

AD No.: 2012-0013 Date: 20 January 2012 Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation

This AD is issued in accordance with EC 1702/2003, Part 21A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency IEC 2042/2003 Appex I. Part M A 303I or agreed with the Authority of the State of Registry IEC 216/2003 Appex I. Part M A 303I or agreed with the Authority of the State of Registry IEC 216/2003 Appex I. Part M A 303I or agreed with the Authority of the State of Registry IEC 216/2003 Appex I. Part M A 303I or agreed with the Authority of the State of Registry IEC 216/2003 Appex I. Part M A 303I or agreed with the Authority of the State of Registry IEC 216/2003 Appex I. Part M A 303I or agreed with the Authority of the State of Registry IEC 216/2003 Appex I. Part M A 303I or agreed with the Authority of the State of Registry IEC 216/2003 Appex I. Part M A 303I or agreed with the Authority of the State of Registry IEC 216/2003 Appex I. Part M A 303I or agreed with the Authority of the State of Registry IEC 216/2003 Appex I. Part M A 303I or agreed with the Authority of the State of Registry IEC 216/2003 Appex I. Part M A 303I or agreed with the Authority of the State of Registry IEC 216/2003 Appex I. Part M A 303I or agreed with the Authority of the State of Registry IEC 216/2003 Appex I. Part M A 303I or agreed with the Authority of the State of Registry IEC 216/2003 Appex I. Part M A 303I or agreed with the Authority of the State of Registry IEC 216/2003 Appex II.

Type Approval Holder's Name :		Type/Model designation(s):
AIRBUS		A380 aeroplanes
TCDS Number:	EASA.A.110	
Foreign AD:	Not applicable	
Supersedure:	None	
ATA 57	Wings – Wing F	Ril Foot Inspection
Manufacturer(s):	Airbus	
Applicability:	Airbus A38C 341, A380-842, and A380-861 aeroplanes, serial numbers (S/N) 01 5/N 03, S/N 04, S/N 05, S/N 06, S/N 07, S/N 08, S/N 09, S/N 10, S/N 12, S 113, S/N 16, S/N 17, S/N 19, S/N 20, S/N 21, S/N 23, S/N 33, S/N 34 and S/N 5	
Reason:	have been found v	cheduled internal inspection of an A380 wing, some rib feet with cracks originating from the rib to skin panel attachment cks according to Airbus All Operator Telex (AOT)
	aeroplanes where inspections, a new edges of the vertice according to Airbu significant than the	ling, inspections were carried out on a number of other further cracks have been found. During one of those of form of rib foot cracking originating from the forward and a cal web of the rib feet has been identified (Type 2 cracks as AOT terminology). The new form of cracking is more be original rib foot hole cracking. It has been determined that may develop on other aeroplanes after a period of time in
	This condition, if n structural integrity	not detected and corrected, could potentially affect the of the aeroplane.
		escribed above, this AD requires a Detailed Visual Inspecting rib feet. This AD also requires reporting the inspection

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	This AD is considered to be an interim action to immediately address this condition. As a result of the on-going investigation, further mandatory actions might be considered.			
Effective Date:	24 January 2012			
Required Action(s) and Compliance Time(s):	Required as indicated, unless already accomplished: (1) Within the compliance time defined in Table 1 of this AD, as applicable, depending on the number of flight cycles (FC) accumulated by the aeroplane at the effective date of this AD, accomplish a DVI of the Left Hand and Right Hand wing in accordance with the instructions of Airbus AOT A380-57A8058. Table 1			
	FC accumulated by the aeroplane at the effective date of this AD	Compliance and or the DVI		
	Aeroplane has accumulated between 1 300 FC and 1 799 FC since aeroplane first flight	Within the section of the section whichever occurs first after the effective date of this Ab.		
	Aeroplane has accumulated 1 800 FC or more since aeroplane first flight	Vithin 4 days or 14 FC, whichever occurs first after the effective date of this A		
	(2) If any crack is detected change the DVI as required by paragraph (1) of this AD, before next flight, contact Airbus for approved instructions and accomplish this instructions within the specified compliance time.			
	(3) Within 2 lays a conscomplishment of the DVI as required by paragraph (1) of this Astroport inspection results (including no findings) to Airbus.			
Ref. Publications:	Airbu, AO7 A380-57A8058 original issue dated 20 January 2012. The se solate approved revisions of this document is acceptable for compliance with the requirements of this AD.			
Remarks:	If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.			
6	The required actions and the risk allowance have granted the issuance of a Final AD with Request for Comments, postponing the public consultation process after publication.			
	 Enquiries regarding this AD should be referred to the Safety Information Section, Executive Directorate, EASA. E-mail: ADs@easa.europa.eu. 			
	 For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS SAS - EANA (Airworthiness Office), Phone: +33 562110253; Fax:+33 562 110 307 E-mail: account.airworth-A380@airbus.com and Nicolas.Cordeau@airbus.com. 			

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