

EASA	AIRWORTHINESS DIRECTIVE
	AD No.: 2013-0005 Date: 10 January 2013 Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.
This AD is issued in accordance with EU 748/2012, Part 21.A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].	
Design Approval Holder's Name: AIRBUS	Type/Model designation(s): A380 aeroplanes
TCDS Number:	EASA.A.110
Foreign AD:	Not applicable
Supersedure:	None
ATA 32	Landing Gear – Body Landing Gear Retraction Actuator Pin Collar – Replacement
Manufacturer(s):	Airbus
Applicability:	Airbus A380-841, A380-842, and A380-861 aeroplanes, all manufacturer serial numbers.
Reason:	During fatigue testing of the A380 body landing gear (BLG), the retraction actuator pin collar failed on several occasions. This condition, if not corrected, could result in collapse of a BLG unit, adversely affecting the structural integrity of the aeroplane. Further analyses of the retraction actuator pin collar failures demonstrated a reduced fatigue life, equal to 2 895 accumulated flight cycles (FC). For the reasons described above, this AD requires the replacement of the retraction actuator pin collar fitted on both Left Hand (LH) and Right Hand (RH) BLG with a new design retraction actuator pin collar, which restores the aeroplane to a condition that meets the Design Service Goal.
Effective Date:	24 January 2013

Required Action(s) and Compliance Time(s):	Required as indicated, unless accomplished previously: (1) Within 2 895 FC from LH or RH BLG first installation on an aeroplane, as applicable, modify the LH and RH BLG in accordance with the instructions of Airbus Service Bulletin (SB) A380-32-8015 at revision 01: (1.1) Replace the retraction actuator pin collar having Part Number (P/N) 2144M5279-1 with a retraction actuator pin collar having P/N 2144M5279-2, and (1.2) Re-identify the BLG pintle frame by striking number 17 on the BLG pintle frame nameplate. (2) Aeroplanes which have been modified in production in accordance with Airbus modification 68762, or aeroplanes which have been modified in service before the effective date of this AD in accordance with the instructions of Airbus SB A380-32-8015 at original issue or revision 01, are not affected by the requirements of paragraph (1) of this AD, provided it can be demonstrated that retraction actuator pin collar having P/N 2144M5279-1 are not currently installed on that aeroplane. (3) After modification of an aeroplane as required by paragraph (1) of this AD, or from the effective date of this AD for aeroplanes which have been modified in production in accordance with Airbus modification 68762, as applicable, do not install any retraction actuator pin collar having P/N 2144M5279-1 on a BLG pintle frame assembly on that aeroplane. In addition, from the effective date of this AD, do not install a BLG unit on any aeroplane, unless it has been determined that retraction actuator pin collar having P/N 2144M5279-1 is not fitted on that BLG. Note: Retraction actuator pin collar having P/N 2144M5279-1 is not fitted on pintle frame assemblies having P/N (first 9 digits) 2144A5305 or 2244A5305 which have been modified in accordance with the instructions of Goodrich SB 2144A5305-32-014 at original issue or revision 01, provided that no replacement of retraction actuator pin collar has been done after that modification.
Ref. Publications:	Airbus SB A380-32-8015 original issue dated 15 June 2009 or revision 01 dated 21 December 2012. The use of later approved revisions of this document is acceptable for compliance with the requirements of this AD. Goodrich SB 2144A5305-32-014 original issue dated 05 May 2009 or revision 01 dated 07 August 2009.
Remarks:	1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD. 2. This AD was posted on 28 November 2012 as PAD 12-150 for consultation until 12 December 2012. The Comment Response Document can be found at http://ad.easa.europa.eu. 3. Enquiries regarding this AD should be referred to the Safety Information Section, Executive Directorate, EASA. E-mail: ADs@easa.europa.eu. 4. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS SAS - EIANA (Airworthiness Office), E-mail: account.airworth-A380@airbus.com.