EASA AD No.: 2014-0270R1

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EASA

AIRWORTHINESS DIRECTIVE

AD No.: 2014-0270R1

Date: 15 December 2014

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.

This AD is issued in accordance with EU 748/2012, Part 21.A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].

Design Approval AIRBUS	Holder's Name:	Type/Model designation(s): A319 and A320 aeroplanes
TCDS Number:	EASA.A.064	
Foreign AD:	Note applicable	
Revision:	This AD revises EASA Emerg	gency AD 2014-0270-E dated 11 December 2014.
ATA 57	Wings – Wing Bottom Skin at Main Landing Gear Rib 5 – Inspection	
Manufacturer(s):	Airbus (formerly Airbus Inc	dustrie)
Applicability:	aeroplanes, manufacture	33, A320-214, A320-216, A320-232 and A320-233 r serial numbers (MSN) 5817, 5826, 5837, 5848, 5896 and 5910, and MSN 5918 and up.
Reason:	after installation. The fast	s, a number of taperlok fasteners were found failed eners in question are located at the bottom skin of ILG) reinforcing plate, wing skin and Gear Support
	This condition, if not detected and corrected could reduce the design margin of the structure.	
0	A319 and A320 aeroplane aeroplanes are not affected	e preliminary investigation, this affects only certain e Models delivered since January 2014. A321 ed, as the wing assembly is done using parallel es are not affected, since none have been 014.
	require repetitive inspection	gs, EASA issued Emergency AD 2014-0270-E to ons of the bottom skin taperlok fasteners at the on and, depending on findings, accomplishment of on(s).
	Since that AD was issued clarification, as well as con	, operator comments have indicated the need for rrection.

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		For the reason described above, this AD is revised to add Notes for information and to correct paragraphs (1) and (2) of the AD.	
		This AD is still considered to be an interim action and further AD action may follow.	
	Effective Date:	Revision 1: 22 December 2014	
		Original issue: 15 December 2014	
	Required Action(s) and Compliance Time(s):	Required as indicated, unless accomplished previously:	
		(1) Within 8 calendar days after 15 December 2014 [the effective date of the original issue of this AD], or 8 calendar days after the aeroplane date of manufacture (see Note 1), or before next flight (see Note 2), whichever occurs later, and, thereafter, at intervals not to exceed 8 calendar days or before each flight (see Note 2), whichever occurs later, accomplish the actions specified in paragraphs (1.1) and (1.2) of this AD in accordance with the instructions of Airbus Alert Operators Transmission (AOT) A57N006-14.	
		(1.1) Accomplish a Detailed Visual Inspection (DVI) of the external surface of the Left Hand (LH) and Right Hand (RH) lower skin surface to detect missing or migrating fasteners.	
		(1.2) Accomplish a DVI of the inboard MLG support rib lower flange to detect broken nuts or/and fastener tails.	
		Note 1: For the purpose of this AD, the date of manufacture is the date of transfer of title, which is referenced in Airbus documentation at the time of first delivery to an operator.	
		Note 2: In case an aeroplane is not operated for more than 8 days, the next inspection is to be performed on that aeroplane before next flight.	
		(2) Within 4 months after the effective date of this AD, or after the aeroplane date of manufacture (see Note 1), or before next flight (see Note 3) whichever occurs later, and, thereafter, at intervals not to exceed 4 months or before each flight (see Note 3), whichever occurs later, accomplish a DVI of the outboard MLG support rib lower flange fasteners and nuts in accordance with the instructions of Airbus AOT A57N006-14.	
		Note 3: In case an aeroplane is not operated for more than 4 months, the next inspection is to be performed on that aeroplane before next flight.	
I		(3) If, during any DVI as required by paragraph (1) or (2) of this AD, as applicable, any discrepancy (missing, migrated or damaged fastener; or missing nut) is detected, before next flight, accomplish the applicable corrective actions in accordance with the instructions of Airbus AOT A57N006-14.	
		(4) Replacement of fasteners or nuts on an aeroplane, as required by paragraph (3) of this AD, does not constitute terminating action for the repetitive inspections as required by this AD for that aeroplane.	
	Ref. Publications:	AOT A57N006-14 original issue dated 04 December 2014.	
		The use of later approved revisions of this document is acceptable for compliance with the requirements of this AD.	
	Remarks:	If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.	
		 The results of the safety assessment have indicated the need for immediate publication and notification, without the full public consultation process. 	

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- 3. Enquiries regarding this AD should be referred to the Safety Information Section, Certification Directorate, EASA. E-mail: ADs@easa.europa.eu.
- For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – Airworthiness Office – EIAS; Fax +33 5 61 93 44 51; E-mail: account.airworth-eas@airbus.com.

