


<b>EASA</b>	<b>NOTIFICATION OF A PROPOSAL TO ISSUE AN AIRWORTHINESS DIRECTIVE</b>
	<p><b>PAD No.: 13-019</b></p> <p><b>Date: 22 January 2013</b></p> <p>Note: This Proposed Airworthiness Directive (PAD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.</p>
<p>In accordance with the EASA Continuing Airworthiness Procedures, the Executive Director is proposing the issuance of an EASA Airworthiness Directive (AD), applicable to the aeronautical product(s) identified below. All interested persons may send their comments, referencing the PAD Number above, to the e-mail address specified in the 'Remarks' section, prior to the consultation closing date indicated.</p>	
<b>Design Approval Holder's Name:</b> SOCATA	<b>Type/Model designation(s):</b> TBM 700 aeroplanes
TCDS Number: EASA A.010	
Foreign AD: Not applicable	
Supersedure: This AD supersedes DGAC France AD 2000-307(A) dated 09 August 2000.	
<b>ATA 55</b>	<b>Stabilizers – Horizontal Stabiliser Outboard Fittings – Modification</b>
<b>Manufacturer(s):</b>	SOCATA (formerly EADS SOCATA)
<b>Applicability:</b>	SOCATA TBM 700 aeroplanes, manufacturer serial numbers (MSN): 1 through 161 inclusive, except MSN 99 and 157, which were already modified during production.
<b>Reason:</b>	<p>During the 1990s, several occurrences were reported of finding cracks in the outboard hinge fittings of the horizontal stabiliser on TBM 700 aeroplanes.</p> <p>This condition, if not detected and corrected, could result in rupture of the outboard hinge fittings, which would adversely affect the structural integrity of the horizontal stabiliser. The in-flight loss of the horizontal stabiliser would result in reduced control of the aeroplane.</p> <p>To address this unsafe condition, DGAC France issued AD 1999-060(A), requiring repetitive inspections of the fittings and, depending on findings, corrective action.</p> <p>After that AD was issued, SOCATA determined that the cause of the cracks was a wrong installation of the fittings during production, inducing stress. Consequently, DGAC France issued AD 2000-307(A), partially retaining the requirements of DGAC France AD 1999-060(A), which was superseded, and required, depending on findings, that the installation of the fittings of in-service aeroplanes be rectified by introduction of adjusting shims, a modification which was introduced as standard on the production line from MSN 162. The periodical inspection of the fittings for cracks was still required, pending a better understanding of the cause of the cracks.</p>



	<p>Since DGAC France AD 2000-307(A) was issued, the results of the further analysis revealed that the final design (installation of shims on the outboard hinge fittings of the horizontal stabiliser) guarantees a service fatigue life which exceeds the one established for the TBM 700 during certification. Consequently, for aeroplanes with this modification, the repetitive inspections of the fittings can be discontinued. However, as the installation of the fittings was only required depending on findings, this modification may not have been accomplished on all affected aeroplanes.</p> <p>For the reasons described above, this AD supersedes (and thereby cancels the requirements of) DGAC France AD 2000-307(A) and requires installation of shims on the outboard hinge fittings of the horizontal stabiliser.</p>
Effective Date:	[TBD: 14 days after final AD issue date]
Required Action(s) and Compliance Time(s):	<p>Required as indicated, unless accomplished previously:</p> <p>Within 110 flight hours after the effective date of this AD, install shims on the outboard hinge fittings of the horizontal stabiliser, in accordance with the instructions of SOCATA Service Bulletin (SB) 70-080-55.</p>
Ref. Publications:	<p>SOCATA SB SB 70-080-55 Original issue dated July 2000, or Revision 1 dated August 2000, or Revision 2 dated August 2012.</p> <p>The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.</p>
Remarks:	<ol style="list-style-type: none"> <li>1. This Proposed AD will be closed for consultation on 19 February 2013.</li> <li>2. Enquiries regarding this PAD should be referred to the Safety Information Section, Executive Directorate, EASA. E-mail: <a href="mailto:ADs@easa.europa.eu">ADs@easa.europa.eu</a>.</li> <li>3. For any question concerning the technical content of the requirements in this PAD, please contact: <p>SOCATA, Direction des services, 65921 Tarbes Cedex 9, France. Tel. +33 (0) 5 62 41 73 00, Fax : + 33 (0) 5 62 41 76 54.</p> <p>or for the U.S.A</p> <p>SOCATA NORTH AMERICA, North Perry Airport, 7501 South Airport Road, Pembroke Pines, Florida 33023, The United States of America. Tel.: +1 (954) 893 1400 Fax: +1 (954) 964 4141.</p> </li> </ol>