


<b>EASA</b>	<b>NOTIFICATION OF A PROPOSAL TO ISSUE AN AIRWORTHINESS DIRECTIVE</b>
	<p><b>PAD No.: 13-025</b></p> <p><b>Date: 04 February 2013</b></p> <p>Note: This Proposed Airworthiness Directive (PAD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.</p>
<p>In accordance with the EASA Continuing Airworthiness Procedures, the Executive Director is proposing the issuance of an EASA Airworthiness Directive (AD), applicable to the aeronautical product(s) identified below. All interested persons may send their comments, referencing the PAD Number above, to the e-mail address specified in the 'Remarks' section, prior to the consultation closing date indicated.</p>	
<p><b>Design Approval Holder's Name:</b></p> <p>AIRBUS</p>	<p><b>Type/Model designation(s):</b></p> <p>A318, A319, A320 and A321 aeroplanes</p>
TCDS Number:	EASA.A.064
Foreign AD:	Not Applicable
Supersedure:	None
<b>ATA 53</b>	<b>Fuselage – Side Box Beam Flange in Frame 43 Area – Inspection / Repair</b>
Manufacturer(s):	Airbus (Formerly Airbus Industries)
Applicability:	Airbus A318-111, A318-112, A318-121, A318-122, A319-111, A319-112, A319-113, A319-114, A319-115, A319-131, A319-132, A319-133, A320-111, A320-211, A320-212, A320-214, A320-215, A320-216, A320-231, A320-232, A320-233, A321-111, A321-112, A321-131, A321-211, A321-212, A321-213, A321-231 and A321-232 aeroplanes, all manufacturer serial numbers, except those on which Airbus modification (mod.) 21202 and mod.152569 have been embodied in production.
Reason:	<p>During the full scale fatigue test campaign of the A320 family type design, a crack was reported in the fuselage side box beam flange at frame (FR) 43 level, both sides.</p> <p>This condition, if not detected and corrected, could affect the structural integrity of the aeroplane.</p> <p>For the reason describe above, this AD requires repetitive inspections of the fuselage side box beam flange at FR43, and, depending on findings, corrective action(s). This AD also includes an optional modification, through Airbus Service Bulletin (SB) A320-53-1251 (mod. 153835), accomplishment of which constitutes terminating action for the repetitive inspections required by this AD.</p>
Effective Date:	[TBD: 14 days after final AD issue date]

<p>Required Action(s) and Compliance Time(s):</p>	<p>Required as indicated, unless accomplished previously:</p> <p>(1) Within the compliance time defined in Table 1 of this AD, as applicable, and thereafter at intervals not to exceed the values defined in Airbus SB A320-53-1258, accomplish a rototest inspection of Stiffener 15 side box beam flange on left hand and right hand sides in FR43 area in accordance with the instructions of Airbus SB A320-53-1258.</p> <p style="text-align: center;">Table 1 - Initial inspection</p> <table border="1" data-bbox="563 432 1386 642"> <tr> <th></th><th><b>Compliance time (whichever occurs later, A or B)</b></th></tr> <tr> <td><b>A</b></td><td>Within the thresholds defined in Airbus SB A320-53-1258</td></tr> <tr> <td><b>B</b></td><td>Within 3 000 flight cycles or 6 000 flight hours, whichever occurs first after the effective date of this AD</td></tr> </table> <p>(2) If, during any inspection as required by paragraph (1) of this AD, a crack is detected, before next flight, contact Airbus for approved repair instructions and accomplish those instructions accordingly.</p> <p>(3) Modification of an aeroplane in accordance with the instructions of Airbus SB A320-53-1251 constitutes terminating action for the repetitive inspections required by paragraph (1) of this AD for that aeroplane.</p>		<b>Compliance time (whichever occurs later, A or B)</b>	<b>A</b>	Within the thresholds defined in Airbus SB A320-53-1258	<b>B</b>	Within 3 000 flight cycles or 6 000 flight hours, whichever occurs first after the effective date of this AD
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<b>B</b>	Within 3 000 flight cycles or 6 000 flight hours, whichever occurs first after the effective date of this AD						
<p>Ref. Publications:</p>	<p>Airbus SB A320-53-1258 Original Issue dated 18 October 2012.</p> <p>Airbus SB A320-53-1251 Original Issue dated 16 November 2012.</p> <p>The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.</p>						
<p>Remarks:</p>	<ol style="list-style-type: none"> <li>1. This Proposed AD will be closed for consultation on 04 March 2013.</li> <li>2. Enquiries regarding this PAD should be referred to the Safety Information Section, Executive Directorate, EASA. E-mail: <a href="mailto:ADs@easa.europa.eu">ADs@easa.europa.eu</a>.</li> <li>3. For any question concerning the technical content of the requirements in this PAD, please contact: AIRBUS - Airworthiness Office - EIAS. Fax +33 5 61 93 44 51. E-mail: <a href="mailto:account.airworth-eas@airbus.com">account.airworth-eas@airbus.com</a>.</li> </ol>						