


<b>EASA</b>	<b>NOTIFICATION OF A PROPOSAL TO ISSUE AN AIRWORTHINESS DIRECTIVE</b>	
	<b>PAD No.: 13-030</b>  <b>Date: 06 February 2013</b>  Note: This Proposed Airworthiness Directive (PAD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.	
	In accordance with the EASA Continuing Airworthiness Procedures, the Executive Director is proposing the issuance of an EASA Airworthiness Directive (AD), applicable to the aeronautical product(s) identified below. All interested persons may send their comments, referencing the PAD Number above, to the e-mail address specified in the 'Remarks' section, prior to the consultation closing date indicated.	
<b>Design Approval Holder's Name:</b> AIRBUS		<b>Type/Model designation(s):</b> A319, A320 and A321 aeroplanes
TCDS Number: EASA.A.064		
Foreign AD: Not Applicable		
Supersedure: None		
<b>ATA 53</b>		
<b>Fuselage – Tack and Rivet Holes in Fitting Brackets at Frame 50 to Frame 63 – Inspection / Repair</b>		
Manufacturer(s): Airbus (formerly Airbus Industrie)		
Applicability: Airbus A319-111, A319-112, A319-113, A319-114, A319-115, A319-131, A319-132, A319-133, A320-111, A320-211, A320-212, A320-214, A320-215, A320-216, A320-231, A320-232, A320-233, A321-111, A321-112, A321-131, A321-211, A321-212, A321-213, A321-231 and A321-232 aeroplanes, all manufacturer serial numbers, except those on which Airbus modification (mod) 153065 and mod 153563 have been incorporated in production, or Airbus Service Bulletin (SB) A320-53-1261 has been accomplished in-service.		
Reason: During a full scale fatigue test, several broken frames in the cargo compartment area between Frame (FR) 50 and FR 63, have been found, especially on the cargo floor support fittings and open tack holes on left hand side.  This condition, if not detected and corrected, could affect the structural integrity of the aeroplane.  For the reason described above, this AD requires repetitive inspections of the frames in the cargo compartment area and of the cargo floor support fittings and open tack holes on the left hand (LH) side and, depending on findings, the accomplishment of applicable corrective action(s).  This AD also includes an optional modification, which constitutes terminating action for the repetitive inspections required by this AD.		
Effective Date: [TBD: 14 days after final AD issue date]		

<p>Required Action(s) and Compliance Time(s):</p>	<p>Required as indicated, unless accomplished previously:</p> <p>(1) Within the compliance time(s) defined in Table 1 of this AD, as applicable, and, thereafter, at intervals not to exceed 5 000 flight cycles (FC) or 10 000 flight hours (FH), whichever occurs first, accomplish a rototest inspection of open tack holes and rivet holes at the cargo floor support fittings between frame (FR) 50 and FR 63 (LH side only) for A320/A321 aeroplanes, and between FR 53 and FR 63 (LH side only) for A319 aeroplanes, as applicable, in accordance with the instructions of Airbus SB A320-53-1257.</p> <p style="text-align: center;">Table 1 - Initial inspection</p> <table border="1" data-bbox="563 495 1406 1023"> <thead> <tr> <th data-bbox="563 495 1007 633">Time accumulated on the effective date of this AD (whichever occurs first since aeroplane first flight)</th><th data-bbox="1007 495 1406 633">Compliance Time (whichever occurs first)</th></tr> </thead> <tbody> <tr> <td data-bbox="563 633 1007 741">Equal to or more than 45 000 FC or 90 000 FH</td><td data-bbox="1007 633 1406 741">Within 1 000 FC or 2 000 FH after the effective date of this AD</td></tr> <tr> <td data-bbox="563 741 1007 913">Equal to or more than 36 200 FC or 72 400 FH, but less than 45 000 FC or 90 000 FH</td><td data-bbox="1007 741 1406 913">Within 2 000 FC or 4000 FH after the effective date of this AD, but not exceeding 46 000 FC or 92 000 FH since aeroplane first flight</td></tr> <tr> <td data-bbox="563 913 1007 1023">Less than 36 200 FC or 72 400 FH</td><td data-bbox="1007 913 1406 1023">Before exceeding 38 200 FC or 76 400 FH since aeroplane first flight</td></tr> </tbody> </table> <p>(2) If, during any inspection as required by paragraph (1) of this AD, a crack is detected, before next flight, contact Airbus for approved repair instructions and accomplish those instructions accordingly.</p> <p>(3) Modification of an aeroplane in accordance with the instructions of Airbus SB A320-53-1261 constitutes terminating action for the repetitive inspections required by paragraph (1) of this AD.</p>	Time accumulated on the effective date of this AD (whichever occurs first since aeroplane first flight)	Compliance Time (whichever occurs first)	Equal to or more than 45 000 FC or 90 000 FH	Within 1 000 FC or 2 000 FH after the effective date of this AD	Equal to or more than 36 200 FC or 72 400 FH, but less than 45 000 FC or 90 000 FH	Within 2 000 FC or 4000 FH after the effective date of this AD, but not exceeding 46 000 FC or 92 000 FH since aeroplane first flight	Less than 36 200 FC or 72 400 FH	Before exceeding 38 200 FC or 76 400 FH since aeroplane first flight
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<p>Ref. Publications:</p>	<p>Airbus SB A320-53-1257 Original Issue dated 21 December 2012.</p> <p>Airbus SB A320-53-1261 Original Issue dated 21 December 2012.</p> <p>The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.</p>								
<p>Remarks:</p>	<ol style="list-style-type: none"> <li>1. This Proposed AD will be closed for consultation on 06 March 2013.</li> <li>2. Enquiries regarding this PAD should be referred to the Safety Information Section, Executive Directorate, EASA. E-mail: <a href="mailto:ADs@easa.europa.eu">ADs@easa.europa.eu</a>.</li> <li>3. For any question concerning the technical content of the requirements in this PAD, please contact: AIRBUS - Airworthiness Office - EIAS. Fax +33 5 61 93 44 51. E-mail: <a href="mailto:account.airworth-eas@airbus.com">account.airworth-eas@airbus.com</a>.</li> </ol>								