


EASA	AIRWORTHINESS DIRECTIVE
	<p>AD No.: 2013-0232</p> <p>Date: 24 September 2013</p> <p>Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.</p>
<p>This AD is issued in accordance with EU 748/2012, Part 21.A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].</p>	
<p>Design Approval Holder's Name:</p> <p>AIRBUS</p>	<p>Type/Model designation(s):</p> <p>A300-600 and A300-600ST aeroplanes</p>
<p>TCDS Numbers: France No.145 and EASA.A.014</p>	
<p>Foreign AD: Not applicable</p>	
<p>Supersedure: This AD supersedes DGAC France AD 97-374-238(B) dated 03 December 1997 and DGAC France AD 1999-008-020(B) dated 13 January 1999.</p>	
ATA 57	Wings – Top Skin at Front Spar between Ribs 1 and 7 – Inspection / Modification
Manufacturer(s):	Airbus (formerly Airbus Industrie)
Applicability:	Airbus A300-600 and A300-600ST aeroplanes, all certified models, all manufacturer serial numbers.
Reason:	<p>During full-scale fatigue testing conducted in the early 1990's, cracks were found on the top skin of the wing between Ribs 1 and 7, starting at the front spar fastener holes.</p> <p>This condition, if not detected and corrected, could adversely affect the structural integrity of the wing.</p> <p>Consequently, Airbus issued Service Bulletin (SB) A300-57-6045 and DGAC France issued AD 97-374-238 for A300-600 aeroplanes and AD 1999-008-020 for A300-600ST aeroplanes to require repetitive detailed inspections of the wing top skin and, in case of findings, an Eddy Current (EC) inspection, and, depending on the size of the cracks, repair.</p> <p>After those ADs were issued, further cracks to the wing top skin were reported by operators, within an area not covered by the existing ADs. To address this potential unsafe condition, Airbus revised SB A300-57-6045 to extend the area to be inspected.</p> <p>In addition, a fleet survey and updated Fatigue and Damage Tolerance analyses were performed in order to substantiate the second A300-600 Extended Service Goal (ESG2) exercise. The results of these analyses have determined that the inspection thresholds and intervals must be reduced to allow timely detection of these cracks and the accomplishment of an applicable corrective action(s).</p>

	<p>As the ESG2 exercise is only applicable to A300-600 aeroplanes, A300-600ST aeroplanes are now addressed through new Airbus SB A300-57-9026.</p> <p>For the reasons described above, this AD retains the requirements of DGAC France AD 97-374-238(B) and AD 1999-008-020(B), which are superseded, but requires those actions, for A300-600 aeroplanes only, within reduced thresholds and intervals.</p>																																																							
Effective Date:	08 October 2013																																																							
Required Action(s) and Compliance Time(s):	<p>Required as indicated, unless accomplished previously:</p> <p>(1) Initially, within the compliance time defined in Table 1 of this AD, and, thereafter, at intervals not to exceed those defined in Table 2 of this AD, as applicable to aeroplane configuration, accomplish a detailed inspection of the wing top skin between Ribs 1 to 7 in accordance with the instructions of Airbus SB A300-57-6045 Revision 09 or SB A300-57-9026 original issue, as applicable to aeroplane type.</p> <p>Table 1: Inspection threshold (whichever occurs first since aeroplane first flight)</p> <table><tr><th rowspan="2">Aircraft</th><th rowspan="2">Operation</th><th colspan="2">Compliance Time</th></tr><tr><th>FC</th><th>FH</th></tr><tr><td>A300B4-600 A300B4-600R A300C4-600</td><td>Normal Range</td><td>17 100</td><td>38 400</td></tr><tr><td></td><td>Short Range</td><td>17 100</td><td>38 400</td></tr><tr><td>A300F4-600 A300F4-600R A300-600ST</td><td>Normal Range</td><td>22 000</td><td>49 500</td></tr><tr><td></td><td>Short Range</td><td>22 000</td><td>49 500</td></tr></table> <p>Table 2: Inspection interval (whichever occurs first since last inspection)</p> <table><tr><th rowspan="2">Aircraft</th><th rowspan="2">Operation</th><th colspan="2">Compliance Time</th></tr><tr><th>FC</th><th>FH</th></tr><tr><td>A300B4-600 A300B4-600R A300C4-600</td><td>Normal Range</td><td>5 100</td><td>11 000</td></tr><tr><td></td><td>Short Range</td><td>5 500</td><td>8 300</td></tr><tr><td>A300F4-600 A300F4-600R A300-600ST</td><td>Normal Range</td><td>6 500</td><td>14 100</td></tr><tr><td></td><td>Short Range</td><td>7 000</td><td>10 600</td></tr></table> <p>Table 3: Grace Period (whichever occurs later after the effective date of this AD)</p> <table><tr><th rowspan="2">Aircraft</th><th colspan="2">Grace Period</th></tr><tr><th>FC</th><th>FH</th></tr><tr><td>A300B4-600, A300B4-600R A300C4-600</td><td>1 000</td><td>2 200</td></tr><tr><td>A300F4-600R A300-600ST</td><td>1 300</td><td>2 800</td></tr></table> <p>Note: The grace period specified in Table 3 may be applied once for the initial or the next repetitive inspection after the effective date of this AD as required by paragraph (1) of this AD.</p>	Aircraft	Operation	Compliance Time		FC	FH	A300B4-600 A300B4-600R A300C4-600	Normal Range	17 100	38 400		Short Range	17 100	38 400	A300F4-600 A300F4-600R A300-600ST	Normal Range	22 000	49 500		Short Range	22 000	49 500	Aircraft	Operation	Compliance Time		FC	FH	A300B4-600 A300B4-600R A300C4-600	Normal Range	5 100	11 000		Short Range	5 500	8 300	A300F4-600 A300F4-600R A300-600ST	Normal Range	6 500	14 100		Short Range	7 000	10 600	Aircraft	Grace Period		FC	FH	A300B4-600, A300B4-600R A300C4-600	1 000	2 200	A300F4-600R A300-600ST	1 300	2 800
Aircraft	Operation			Compliance Time																																																				
		FC	FH																																																					
A300B4-600 A300B4-600R A300C4-600	Normal Range	17 100	38 400																																																					
	Short Range	17 100	38 400																																																					
A300F4-600 A300F4-600R A300-600ST	Normal Range	22 000	49 500																																																					
	Short Range	22 000	49 500																																																					
Aircraft	Operation	Compliance Time																																																						
		FC	FH																																																					
A300B4-600 A300B4-600R A300C4-600	Normal Range	5 100	11 000																																																					
	Short Range	5 500	8 300																																																					
A300F4-600 A300F4-600R A300-600ST	Normal Range	6 500	14 100																																																					
	Short Range	7 000	10 600																																																					
Aircraft	Grace Period																																																							
	FC	FH																																																						
A300B4-600, A300B4-600R A300C4-600	1 000	2 200																																																						
A300F4-600R A300-600ST	1 300	2 800																																																						

- (2) If, during any inspection as required by paragraph (1) of this AD, a crack is found in the top skin in the area forward of the front spar attachment, before next flight, contact Airbus for approved repair instructions and, within the compliance time defined in those instructions, accomplish the repair accordingly.
- (3) If, during any inspection as required by paragraph (1) of this AD, any crack is found or suspected in the top skin at or aft of the spar attachment, before next flight, accomplish an EC inspection of the affected or suspected area to confirm and measure the length of the crack, in accordance with the instructions of Airbus SB A300-57-6045 Revision 09 or SB A300-57-9026, as applicable to aeroplane type.
- (4) If, during any EC inspection as required by paragraph (3) of this AD, a crack is confirmed, but less than 6 mm in length, within 50 FC or 110 FH, whichever occurs first after crack confirmation, and, thereafter, at intervals not to exceed 50 FC or 110 FH, whichever occurs first, accomplish a detailed inspection of the crack in accordance with the instructions of Airbus SB A300-57-6045 Revision 09 or SB A300-57-9026, as applicable to aeroplane type.
- (5) If, during any inspection as required by paragraph (3) or (4) of this AD, as applicable, a crack is confirmed and is equal to or more than 6 mm but less than 75 mm in length, accomplish the following actions:
- (5.1) Before next flight after crack confirmation, accomplish a temporary repair in accordance with the instructions of Airbus SB A300-57-6045 Revision 09 or SB A300-57-9026, as applicable to aeroplane type.
- (5.2) Within 50 FC or 110 FH, whichever occurs first, after the temporary repair as required by paragraph (5.1) of this AD, and, thereafter, at intervals not to exceed 50 FC or 110 FH, whichever occurs first, accomplish a detailed inspection in accordance with Airbus SB A300-57-6045 Revision 09 or SB A300-57-9026, as applicable to aeroplane type.
- (5.3) Within 250 FC or 550 FH, whichever occurs first, after embodiment of the temporary repair as required by paragraph (5.1) of this AD, contact Airbus for approved permanent repair procedures and accomplish these procedures accordingly.
- (5.4) Permanent repair as required by paragraph (5.3) of this AD constitutes terminating action for the repetitive inspections required by paragraph (5.2) of this AD.
- (6) If, during an inspection as required by paragraph (3) or (4) of this AD, a crack is confirmed and is equal to or more than 75 mm in length, before next flight, contact Airbus for approved repair instructions and accomplish those instructions accordingly.
- (7) Accomplishment of inspections and/or repair as required by paragraph (2), (3), (4), (5) or (6) of this AD, as applicable, does not constitute terminating action for the repetitive inspections required by paragraph (1) of this AD.
- (8) Inspections and/or repair, accomplished before the effective date of this AD in accordance with the instructions of Airbus SB A300-57-6045 at original issue up to Revision 08, are acceptable to comply with the initial requirements of this AD. From the effective date of this AD, the actions required by this AD must be accomplished in accordance with Airbus SB A300-57-6045 Revision 09 or SB A300-57-9026 original issue, as applicable to aeroplane type.

Ref. Publications:	<p>Airbus SB A300-57-6045 Revision 09 dated 21 May 2013.</p> <p>Airbus SB A300-57-9026 original issue dated 06 June 2013.</p> <p>The use of later approved revisions of these documents is acceptable for compliance with the requirements of this AD.</p>
Remarks:	<ol style="list-style-type: none"> 1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD. 2. This AD was posted on 16 August 2013 as PAD 13-120 for consultation until 13 September 2013. No comments were received during the consultation period. 3. Enquiries regarding this AD should be referred to the Safety Information Section, Executive Directorate, EASA. E-mail: ADs@easa.europa.eu. 4. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS SAS – EIAW (Airworthiness Office) E-mail: continued.airworthiness-wb.external@airbus.com.